

Robust Control Toolbox™

Getting Started Guide

R2012b

*Gary Balas
Richard Chiang
Andy Packard
Michael Safonov*

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508-647-7001 (Fax)



The MathWorks, Inc.
3 Apple Hill Drive
Natick, MA 01760-2098

For contact information about worldwide offices, see the MathWorks Web site.

Robust Control Toolbox™ Getting Started Guide

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Product Description

Design robust controllers for uncertain plants

Robust Control Toolbox™ provides tools for analyzing and automatically tuning control systems for performance and robustness. You can create uncertain models by combining nominal dynamics with uncertain elements, such as an uncertain parameter or unmodeled dynamics. You can analyze the impact of plant model uncertainty on control system performance and identify worst-case combinations of uncertain elements. Using H-infinity or mu-synthesis techniques, you can design controllers that maximize robust stability and performance. The toolbox can automatically tune both SISO and MIMO robust controllers, including decentralized control architectures modeled in Simulink®. You can validate your design by calculating worst-case gain and phase margins and worst-case sensitivity to disturbances.

Key Features

- Modeling of systems with uncertain parameters or neglected dynamics
- Worst-case stability and performance analysis of uncertain systems
- Automatic tuning of centralized and decentralized control systems
- Robustness analysis and controller tuning in Simulink
- H-infinity and mu-synthesis algorithms
- General-purpose LMI solvers for feasibility, minimization of linear objectives, and generalized eigenvalue minimization
- Model reduction algorithms based on Hankel singular values

Product Requirements

Robust Control Toolbox software requires that you have installed Control System Toolbox™ software.

Modeling Uncertainty

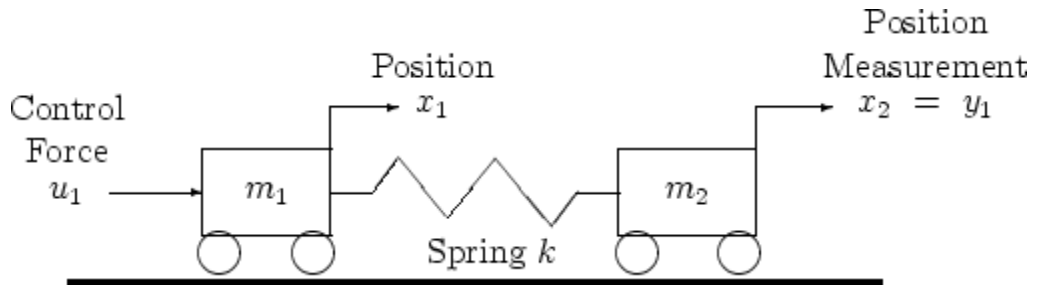
At the heart of robust control is the concept of an uncertain LTI system. Model uncertainty arises when system gains or other parameters are not precisely known, or can vary over a given range. Examples of real parameter uncertainties include uncertain pole and zero locations and uncertain gains. You can also have unstructured uncertainties, by which is meant complex parameter variations satisfying given magnitude bounds.

With Robust Control Toolbox software you can create uncertain LTI models as MATLAB® objects specifically designed for robust control applications. You can build models of complex systems by combining models of subsystems using addition, multiplication, and division, as well as with Control System Toolbox commands like `feedback` and `lft`.

For information about LTI model types, see “Linear System Representation”.

System with Uncertain Parameters

For instance, consider the two-cart "ACC Benchmark" system [13] consisting of two frictionless carts connected by a spring shown as follows.



ACC Benchmark Problem

The system has the block diagram model shown below, where the individual carts have the respective transfer functions.

$$G_1(s) = \frac{1}{m_1 s^2}$$

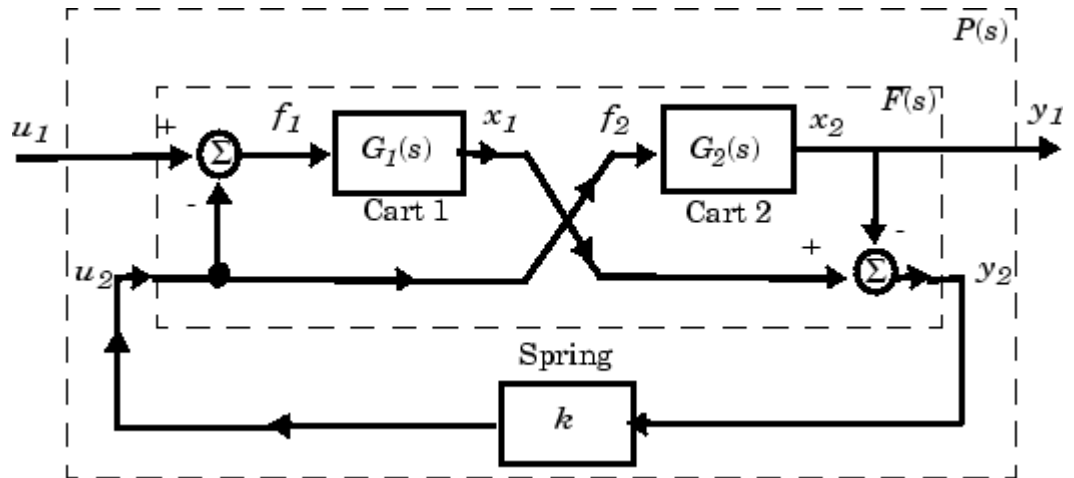
$$G_2(s) = \frac{1}{m_2 s^2}.$$

The parameters m_1 , m_2 , and k are uncertain, equal to one plus or minus 20%:

$$m_1 = 1 \pm 0.2$$

$$m_2 = 1 \pm 0.2$$

$$k = 1 \pm 0.2$$



"ACC Benchmark" Two-Cart System Block Diagram $y_1 = P(s) u_1$

The upper dashed-line block has transfer function matrix $F(s)$:

$$F(s) = \begin{bmatrix} 0 \\ G_1(s) \end{bmatrix} [1 \quad -1] + \begin{bmatrix} 1 \\ -1 \end{bmatrix} [0 \quad G_2(s)].$$

This code builds the uncertain system model P shown above:

```
% Create the uncertain real parameters m1, m2, & k
m1 = ureal('m1',1,'percent',20);
m2 = ureal('m2',1,'percent',20);
k = ureal('k',1,'percent',20);

s = zpk('s'); % create the Laplace variable s
G1 = ss(1/s^2)/m1; % Cart 1
G2 = ss(1/s^2)/m2; % Cart 2

% Now build F and P
F = [0;G1]*[1 -1]+[1;-1]*[0,G2];
P = lft(F,k) % close the loop with the spring k
```

The variable P is a SISO uncertain state-space (USS) object with four states and three uncertain parameters, m_1 , m_2 , and k . You can recover the nominal plant with the command

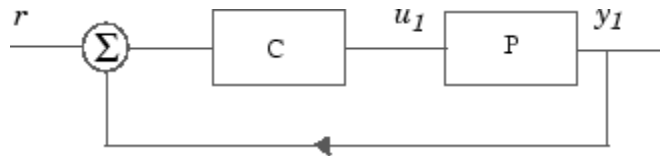
```
zpk(P.nominal)
```

which returns

```
Zero/pole/gain:
      1
-----
s^2 (s^2 + 2)
```

If the uncertain model $P(s)$ has LTI negative feedback controller

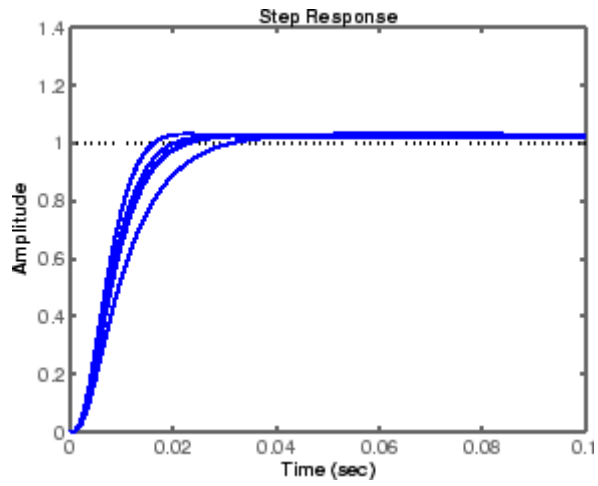
$$C(s) = \frac{100(s+1)^3}{(0.001s+1)^3}$$



then you can form the controller and the closed-loop system $y_1 = T(s) u_1$ and view the closed-loop system's step response on the time interval from $t=0$ to $t=0.1$ for a Monte Carlo random sample of five combinations of the three uncertain parameters k , m_1 , and m_2 using this code:

```
C=100*ss((s+1)/(.001*s+1))^3 % LTI controller
T=feedback(P*C,1); % closed-loop uncertain system
step(usample(T,5),.1);
```

The resulting plot is shown below.



Monte Carlo Sampling of Uncertain System's Step Response

Worst-Case Performance

To be robust, your control system should meet your stability and performance requirements for all possible values of uncertain parameters. Monte Carlo parameter sampling via `usample` can be used for this purpose as shown in Monte Carlo Sampling of Uncertain System's Step Response on page 1-8, but Monte Carlo methods are inherently hit or miss. With Monte Carlo methods, you might need to take an impossibly large number of samples before you hit upon or near a worst-case parameter combination.

Robust Control Toolbox software gives you a powerful assortment of *robustness analysis* commands that let you directly calculate upper and lower bounds on worst-case performance without random sampling.

Worst-Case Robustness Analysis Commands	
<code>loopmargin</code>	Comprehensive analysis of feedback loop
<code>loopsens</code>	Sensitivity functions of feedback loop
<code>ncfmargin</code>	Normalized coprime stability margin of feedback loop
<code>robustperf</code>	Robust performance of uncertain systems
<code>robuststab</code>	Stability margins of uncertain systems
<code>wcgain</code>	Worst-case gain of an uncertain system
<code>wcmargin</code>	Worst-case gain/phase margins for feedback loop
<code>wcsens</code>	Worst-case sensitivity functions of feedback loop

Worst-Case Performance of Uncertain System

Returning to the “System with Uncertain Parameters” on page 1-5, the closed loop system is:

```
T=feedback(P*C,1); % Closed-loop uncertain system
```

This uncertain state-space model T has three uncertain parameters, k , $m1$, and $m2$, each equal to $1\pm 20\%$ uncertain variation. To analyze whether the closed-loop system T is robustly stable for all combinations of values for these three parameters, you can execute the commands:

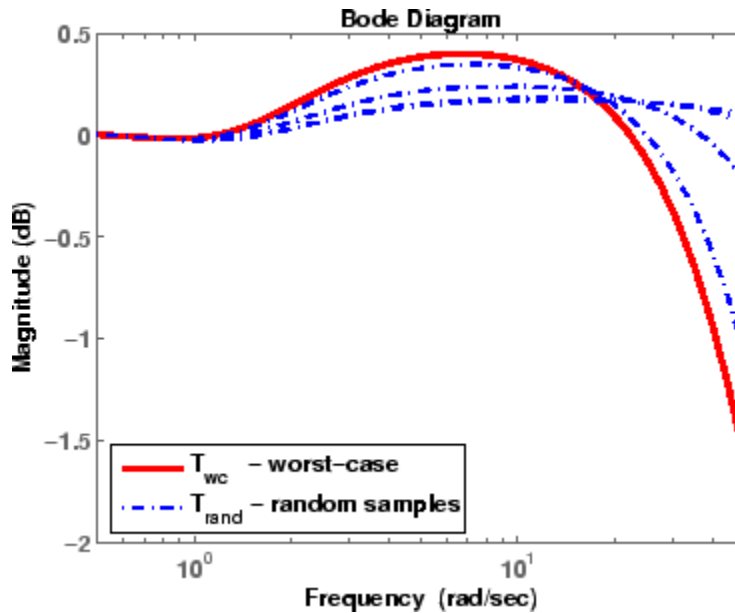
```
[StabilityMargin,Udestab,REPORT] = robuststab(T);  
REPORT
```

This displays the REPORT:

```
Uncertain System is robustly stable to modeled uncertainty.  
-- It can tolerate up to 311% of modeled uncertainty.  
-- A destabilizing combination of 500% the modeled uncertainty exists,  
    causing an instability at 44.3 rad/s.
```

The report tells you that the control system is robust for all parameter variations in the $\pm 20\%$ range, and that the smallest destabilizing combination of real variations in the values k , $m1$, and $m2$ has sizes somewhere between 311% and 500% greater than $\pm 20\%$, i.e., between $\pm 62.2\%$ and $\pm 100\%$. The value $Udestab$ returns an estimate of the 500% destabilizing parameter variation combination:

```
Udestab =  
    k: 1.2174e-005  
    m1: 1.2174e-005  
    m2: 2.0000.
```



Uncertain System Closed-Loop Bode Plots

You have a comfortable safety margin of between 311% to 500% larger than the anticipated $\pm 20\%$ parameter variations before the closed loop goes unstable. But how much can closed-loop performance deteriorate for parameter variations constrained to lie strictly within the anticipated $\pm 20\%$ range? The following code computes worst-case peak gain of T , and estimates the frequency and parameter values at which the peak gain occurs:

```
[PeakGain,Uwc] = wcgain(T);
Twc=usubs(T,Uwc);
% Worst case closed-loop system T
Trand=usample(T,4);
% 4 random samples of uncertain system T
bodemag(Twc,'r',Trand,'b-.',{.5,50}); % Do bode plot
legend('T_{wc} - worst-case',...
'T_{rand} - random samples',3);
```

The resulting plot is shown in Uncertain System Closed-Loop Bode Plots on page 1-11.

Synthesis of Robust MIMO Controllers

You can design controllers for multiinput-multioutput (MIMO) LTI models with your Robust Control Toolbox software using the following command.

Robust Control Synthesis Commands	
h2hinfosyn	Mixed H_2/H_∞ controller synthesis
h2syn	H_2 controller synthesis
hinfosyn	H_∞ controller synthesis
loopsyn	H_∞ loop-shaping controller synthesis
ltrsyn	Loop-transfer recovery controller synthesis
mixsyn	H_∞ mixed-sensitivity controller synthesis
ncfsyn	H_∞ normalized coprime factor controller synthesis
sdhinfosyn	Sampled-data H_∞ controller synthesis

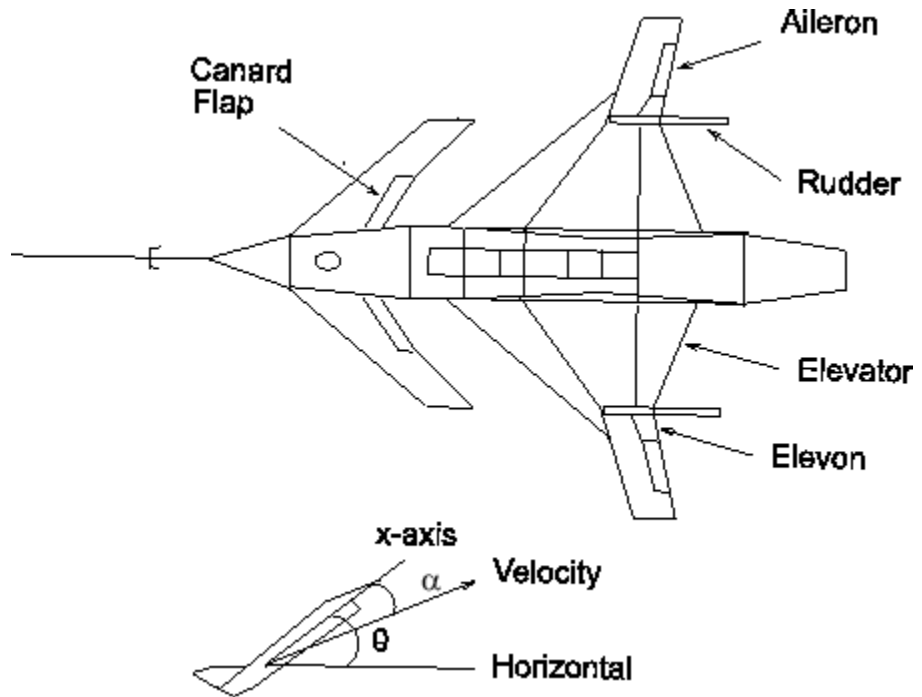
Loop-Shaping Controller Design

One of the most powerful yet simple controller synthesis tools is `loopsyn`. Given an LTI plant, you specify the shape of the open-loop systems frequency response plot that you want, then `loopsyn` computes a stabilizing controller that best approximates your specified loop shape.

For example, consider the 2-by-2 NASA HiMAT aircraft model (Safonov, Laub, and Hartmann [8]) depicted in the following figure. The control variables are elevon and canard actuators (δ_e and δ_c). The output variables are angle of attack (α) and attitude angle (θ). The model has six states:

$$x = \begin{bmatrix} x_1 \\ x_2 \\ x_3 \\ x_4 \\ x_5 \\ x_6 \end{bmatrix} = \begin{bmatrix} \dot{\alpha} \\ \alpha \\ \dot{\theta} \\ \theta \\ x_e \\ x_\delta \end{bmatrix}$$

where x_e and x_δ are elevon and canard actuator states.



Aircraft Configuration and Vertical Plane Geometry

You can enter the state-space matrices for this model with the following code:

```
% NASA HiMAT model G(s)
ag = [ -2.2567e-02  -3.6617e+01  -1.8897e+01  -3.2090e+01  3.2509e+00  -7.6257e-01;
        9.2572e-05  -1.8997e+00  9.8312e-01  -7.2562e-04  -1.7080e-01  -4.9652e-03;
        1.2338e-02  1.1720e+01  -2.6316e+00  8.7582e-04  -3.1604e+01  2.2396e+01;
        0           0           1.0000e+00      0           0           0;
        0           0           0           0          -3.0000e+01  0;
        0           0           0           0           0          -3.0000e+01];
bg = [ 0   0;
       0   0;
       0   0;
       0   0;
       30  0;
       0  30];
cg = [ 0   1   0   0   0   0];
```

```

        0   0   0   1   0   0];
dg = [ 0   0;
      0   0];
G=ss(ag,bg,cg,dg);

```

To design a controller to shape the frequency response (`sigma`) plot so that the system has approximately a bandwidth of 10 rad/s, you can set as your target desired loop shape $G_d(s)=10/s$, then use `loopsyn(G,Gd)` to find a loop-shaping controller for `G` that optimally matches the desired loop shape `Gd` by typing:

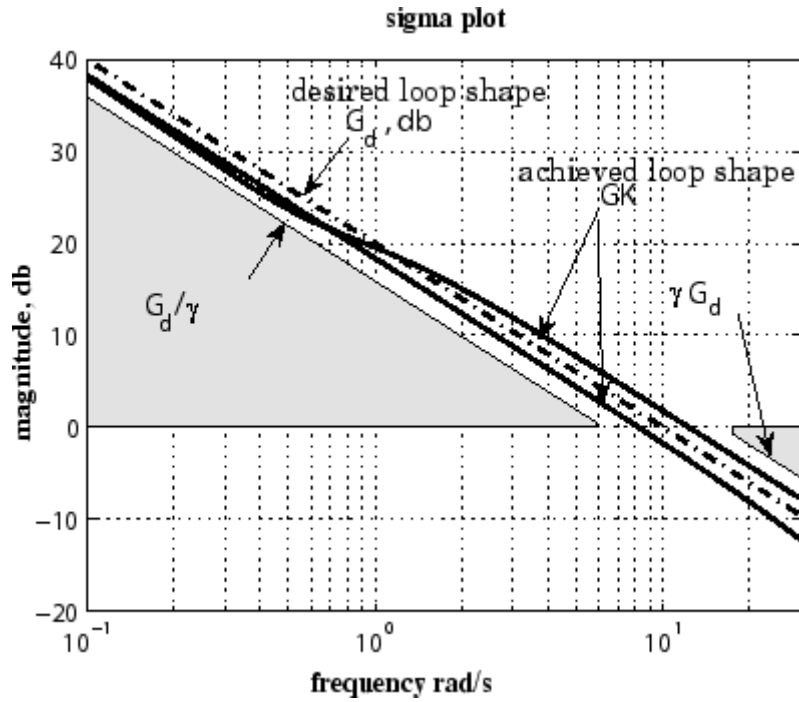
```

s=zpk('s'); w0=10; Gd=w0/(s+.001);
[K,CL,GAM]=loopsyn(G,Gd); % Design a loop-shaping controller K

% Plot the results
sigma(G*K,'r',Gd,'k-.',Gd/GAM,'k:',Gd*GAM,'k:',{.1,30})
figure ;T=feedback(G*K,eye(2));
sigma(T,ss(GAM),'k:',{.1,30});grid

```

The value of $\gamma = \text{GAM}$ returned is an indicator of the accuracy to which the optimal loop shape matches your desired loop shape and is an upper bound on the resonant peak magnitude of the closed-loop transfer function `T=feedback(G*K,eye(2))`. In this case, $\gamma = 1.6024 = 4 \text{ dB}$ — see the next figure.



MIMO Robust Loop Shaping with `loopsyn(G,Gd)`

The achieved loop shape matches the desired target G_d to within about γ dB.

Model Reduction and Approximation

Complex models are not always required for good control. Unfortunately, however, optimization methods (including methods based on H_∞ , H_2 , and μ -synthesis optimal control theory) generally tend to produce controllers with at least as many states as the plant model. For this reason, Robust Control Toolbox software offers you an assortment of model-order reduction commands that help you to find less complex low-order approximations to plant and controller models.

Model Reduction Commands	
<code>reduce</code>	Main interface to model approximation algorithms
<code>balancmr</code>	Balanced truncation model reduction
<code>bstmr</code>	Balanced stochastic truncation model reduction
<code>hankelmr</code>	Optimal Hankel norm model approximations
<code>modreal</code>	State-space modal truncation/realization
<code>ncfmr</code>	Balanced normalized coprime factor model reduction
<code>schurmr</code>	Schur balanced truncation model reduction
<code>slowfast</code>	State-space slow-fast decomposition
<code>stabsep</code>	State-space stable/antistable decomposition
<code>imp2ss</code>	Impulse response to state-space approximation

Among the most important types of model reduction methods are minimize bounds methods on additive, multiplicative, and normalized coprime factor (NCF) model error. You can access all three of these methods using the command `reduce`.

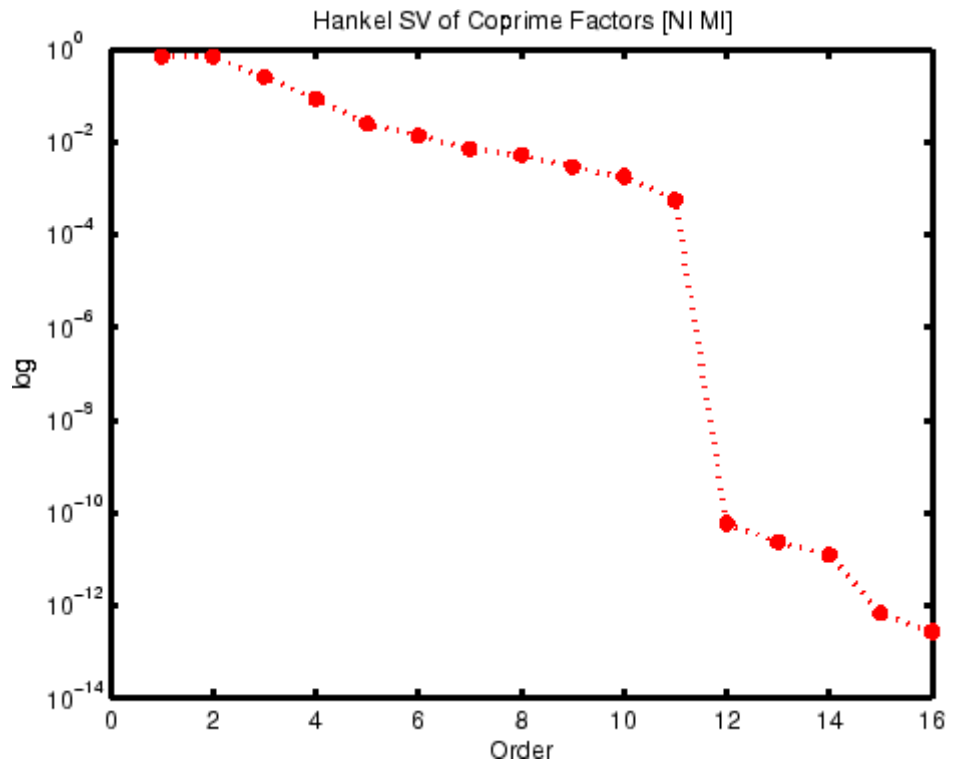
Reduce Order of Synthesized Controller

For instance, the NASA HiMAT model considered in “Loop-Shaping Controller Design” on page 1-13 has eight states, and the optimal loop-shaping controller turns out to have 16 states. Using model reduction, you can remove at least some of the states without appreciably affecting stability or closed-loop performance. For controller order reduction, the NCF model reduction is particularly useful, and it works equally well with controllers that have poles anywhere in the complex plane.

For the NASA HiMAT design in the last section, you can type

```
hanke1sv(K, 'ncf', 'log');
```

which displays a logarithmic plot of the NCF Hankel singular values — see the following figure.

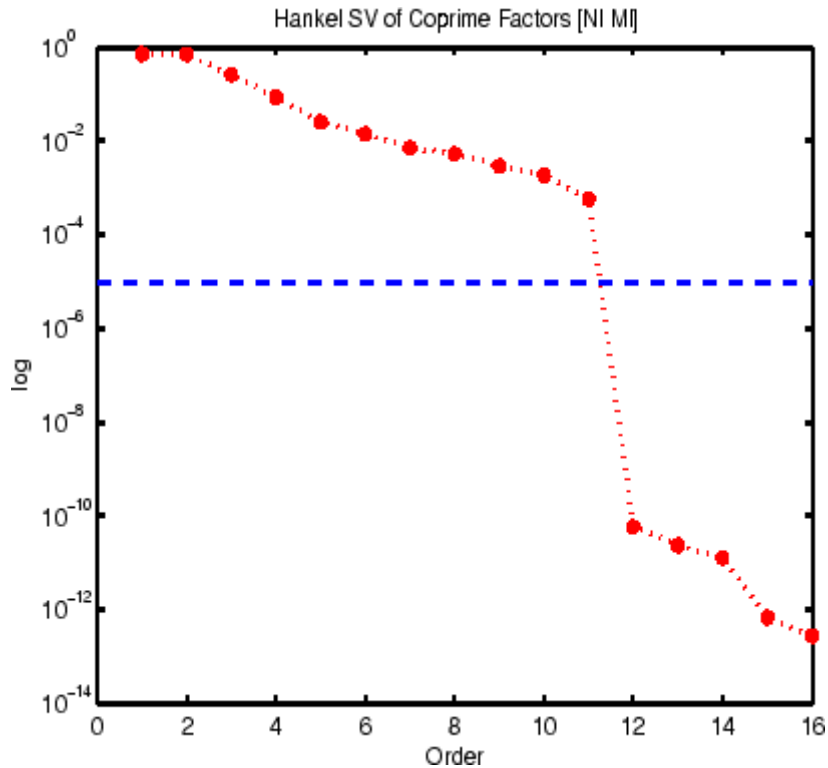


Hankel Singular Values of Coprime Factorization of K

Theory says that, without danger of inducing instability, you can confidently discard at least those controller states that have NCF Hankel singular values that are much smaller than `ncfmargin(G,K)`.

Compute `ncfmargin(G,K)` and add it to your Hankel singular values plot.

```
hankelsv(K,'ncf','log');v=axis;
hold on; plot(v(1:2), ncfmargin(G,K)*[1 1], '--'); hold off
```



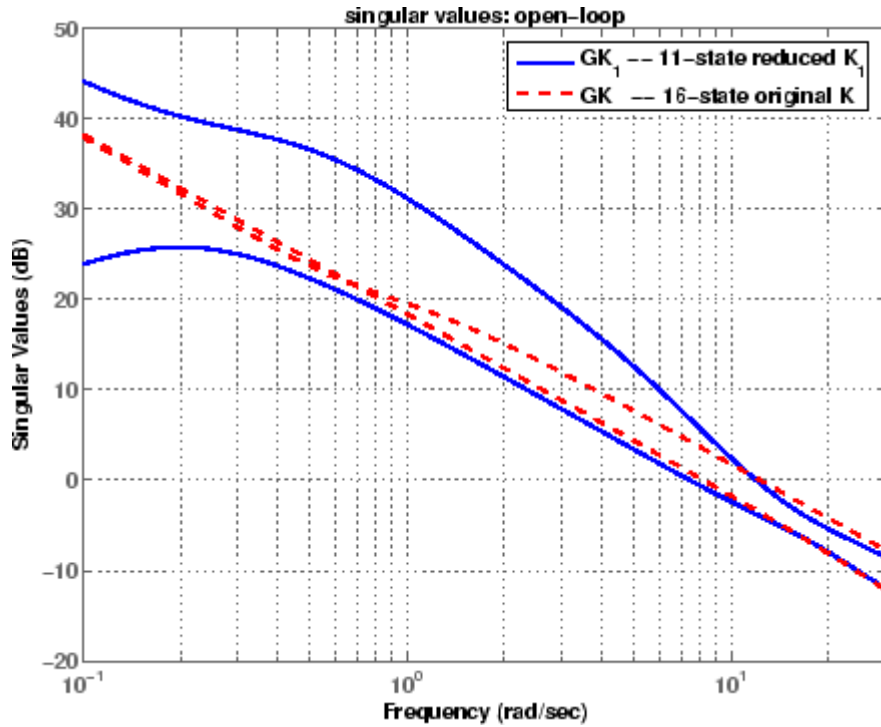
Five of the 16 NCF Hankel Singular Values of HiMAT Controller K Are Small Compared to $\text{ncfmargin}(G,K)$

In this case, you can safely discard 5 of the 16 states of K and compute an 11-state reduced controller by typing:

```
K1=reduce(K,11,'errortype','ncf');
```

The result is plotted in the following figure.

```
sigma(G*K1,'b',G*K,'r--',{.1,30});
```

HiMAT with 11-State Controller K₁ vs. Original 16-State Controller K

The picture above shows that low-frequency gain is decreased considerably for inputs in one vector direction. Although this does not affect stability, it affects performance. If you wanted to better preserve low-frequency performance, you would discard fewer than five of the 16 states of K.

LMI Solvers

At the core of many emergent robust control analysis and synthesis routines are powerful general-purpose functions for solving a class of convex nonlinear programming problems known as linear matrix inequalities. The LMI capabilities are invoked by Robust Control Toolbox software functions that evaluate worst-case performance, as well as functions like `hinfsyn` and `h2hinfyn`. Some of the main functions that help you access the LMI capabilities of the toolbox are shown in the following table.

Specification of LMIs	
<code>lmiedit</code>	GUI for LMI specification
<code>setlmis</code>	Initialize the LMI description
<code>lmivar</code>	Define a new matrix variable
<code>lmiterm</code>	Specify the term content of an LMI
<code>newlmi</code>	Attach an identifying tag to new LMIs
<code>getlmis</code>	Get the internal description of the LMI system

LMI Solvers	
<code>feasp</code>	Test feasibility of a system of LMIs
<code>gevp</code>	Minimize generalized eigenvalue with LMI constraints
<code>mincx</code>	Minimize a linear objective with LMI constraints
<code>dec2mat</code>	Convert output of the solvers to values of matrix variables

Evaluation of LMIs/Validation of Results	
<code>evallmi</code>	Evaluate for given values of the decision variables
<code>showlmi</code>	Return the left and right sides of an evaluated LMI

Extends Control System Toolbox Capabilities

Robust Control Toolbox software is designed to work with Control System Toolbox software. Robust Control Toolbox software extends the capabilities of Control System Toolbox software and leverages the LTI and plotting capabilities of Control System Toolbox software. The major analysis and synthesis commands in Robust Control Toolbox software accept LTI object inputs, e.g., LTI state-space systems produced by commands such as:

```
G=tf(1,[1 2 3])  
G=ss([-1 0; 0 -1], [1;1],[1 1],3)
```

The uncertain system (USS) objects in Robust Control Toolbox software generalize the Control System Toolbox LTI SS objects and help ease the task of analyzing and plotting uncertain systems. You can do many of the same algebraic operations on uncertain systems that are possible for LTI objects (multiply, add, invert), and Robust Control Toolbox software provides USS uncertain system extensions of Control System Toolbox software interconnection and plotting functions like `feedback`, `lft`, and `bode`.

Acknowledgments

Professor **Andy Packard** is with the Faculty of Mechanical Engineering at the University of California, Berkeley. His research interests include robustness issues in control analysis and design, linear algebra and numerical algorithms in control problems, applications of system theory to aerospace problems, flight control, and control of fluid flow.

Professor **Gary Balas** is with the Faculty of Aerospace Engineering & Mechanics at the University of Minnesota and is president of MUSYN Inc. His research interests include aerospace control systems, both experimental and theoretical.

Dr. **Michael Safonov** is with the Faculty of Electrical Engineering at the University of Southern California. His research interests include control and decision theory.

Dr. **Richard Chiang** is employed by Boeing Satellite Systems, El Segundo, CA. He is a Boeing Technical Fellow and has been working in the aerospace industry over 25 years. In his career, Richard has designed 3 flight control laws, 12 spacecraft attitude control laws, and 3 large space structure vibration controllers, using modern robust control theory and the tools he built in this toolbox. His research interests include robust control theory, model reduction, and in-flight system identification. Working in industry instead of academia, Richard serves a unique role in our team, bridging the gap between theory and reality.

The linear matrix inequality (LMI) portion of Robust Control Toolbox software was developed by these two authors:

Dr. **Pascal Gahinet** is employed by MathWorks. His research interests include robust control theory, linear matrix inequalities, numerical linear algebra, and numerical software for control.

Professor **Arkadi Nemirovski** is with the Faculty of Industrial Engineering and Management at Technion, Haifa, Israel. His research interests include convex optimization, complexity theory, and nonparametric statistics.

The structured H_∞ synthesis (`hinfstruct`) portion of Robust Control Toolbox software was developed by the following author in collaboration with Pascal Gahinet:

Professor **Pierre Apkarian** is with ONERA (The French Aerospace Lab) and the Institut de Mathématiques at Paul Sabatier University, Toulouse, France. His research interests include robust control, LMIs, mathematical programming, and nonsmooth optimization techniques for control.

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Multivariable Loop Shaping

- “Tradeoff Between Performance and Robustness” on page 2-2
- “Norms and Singular Values” on page 2-4
- “Typical Loop Shapes, S and T Design” on page 2-6
- “Using LOOPSYN to Do H-Infinity Loop Shaping” on page 2-15
- “Loop-Shaping Control Design of Aircraft Model” on page 2-16
- “Fine-Tuning the LOOPSYN Target Loop Shape G_d to Meet Design Goals” on page 2-21
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- “Mixed-Sensitivity Loop-Shaping Controller Design” on page 2-24
- “Loop-Shaping Commands” on page 2-26

Tradeoff Between Performance and Robustness

When the plant modeling uncertainty is not too big, you can design high-gain, high-performance feedback controllers. High loop gains significantly larger than 1 in magnitude can attenuate the effects of plant model uncertainty and reduce the overall sensitivity of the system to plant noise. But if your plant model uncertainty is so large that you do not even know the sign of your plant gain, then you cannot use large feedback gains without the risk that the system will become unstable. Thus, plant model uncertainty can be a fundamental limiting factor in determining what can be achieved with feedback.

Multiplicative Uncertainty: Given an approximate model of the plant G_0 of a plant G , the *multiplicative uncertainty* Δ_M of the model G_0 is defined

$$\text{as } \Delta_M = G_0^{-1}(G - G_0)$$

or, equivalently,

$$G = (I + \Delta_M)G_0.$$

Plant model uncertainty arises from many sources. There might be small unmodeled time delays or stray electrical capacitance. Imprecisely understood actuator time constants or, in mechanical systems, high-frequency torsional bending modes and similar effects can be responsible for plant model uncertainty. These types of uncertainty are relatively small at lower frequencies and typically increase at higher frequencies.

In the case of single-input/single-output (SISO) plants, the frequency at which there are uncertain variations in your plant of size $|\Delta_M|=2$ marks a critical threshold beyond which there is insufficient information about the plant to reliably design a feedback controller. With such a 200% model uncertainty, the model provides no indication of the phase angle of the true plant, which means that the only way you can reliably stabilize your plant is to ensure that the loop gain is less than 1. Allowing for an additional factor of 2 margin for error, your control system bandwidth is essentially limited

to the frequency range over which your multiplicative plant uncertainty Δ_M has gain magnitude $|\Delta_M| < 1$.

Norms and Singular Values

For MIMO systems the transfer functions are matrices, and relevant measures of gain are determined by singular values, H_∞ , and H_2 norms, which are defined as follows:

H_2 and H_∞ Norms The H_2 -norm is the energy of the impulse response of plant G . The H_∞ -norm is the peak gain of G across all frequencies and all input directions.

Another important concept is the notion of singular values.

Singular Values: The *singular values* of a rank r matrix $A \in C^{m \times n}$, denoted σ_i , are the nonnegative square roots of the eigenvalues of A^*A ordered such that $\sigma_1 \geq \sigma_2 \geq \dots \geq \sigma_p > 0$, $p \leq \min\{m, n\}$.

If $r < p$ then there are $p - r$ zero singular values, i.e., $\sigma_{r+1} = \sigma_{r+2} = \dots = \sigma_p = 0$.

The greatest singular value σ_1 is sometimes denoted

$$\bar{\sigma}(A) = \sigma_1.$$

When A is a square n -by- n matrix, then the n th singular value (i.e., the least singular value) is denoted

$$\bar{\sigma}(A) \square \sigma_n.$$

Properties of Singular Values

Some useful properties of singular values are:

$$\bar{\sigma}(A) = \max_{x \in C^h} \frac{\|Ax\|}{\|x\|}$$

$$\underline{\sigma}(A) = \min_{x \in C^h} \frac{\|Ax\|}{\|x\|}$$

These properties are especially important because they establish that the greatest and least singular values of a matrix A are the maximal and minimal "gains" of the matrix as the input vector x varies over all possible directions.

For stable continuous-time LTI systems $G(s)$, the H_2 -norm and the H_∞ -norms are defined terms of the frequency-dependent singular values of $G(j\omega)$:

H_2 -norm:

$$\|G\|_2 \square \left[\frac{1}{2\pi} \right] \int_{-\infty}^{\infty} \sum_{i=1}^p (\sigma_i(G(j\omega)))^2 d\omega$$

H_∞ -norm:

$$\|G\|_2 \square \sup_{\omega} \bar{\sigma}(G(j\omega))$$

where sup denotes the least upper bound.

Typical Loop Shapes, S and T Design

Consider the multivariable feedback control system shown in the following figure. In order to quantify the multivariable stability margins and performance of such systems, you can use the singular values of the closed-loop transfer function matrices from r to each of the three outputs e , u , and y , *viz.*

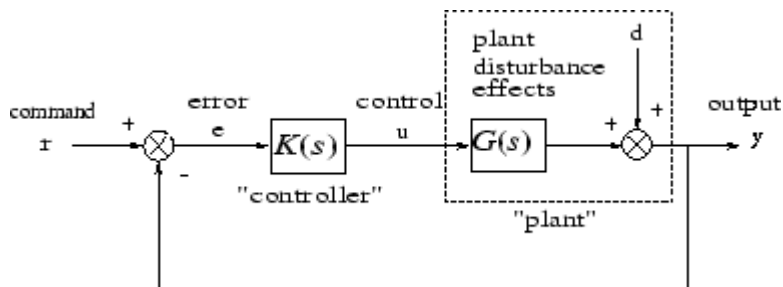
$$S(s) \stackrel{\text{def}}{=} (I + L(s))^{-1}$$

$$R(s) \stackrel{\text{def}}{=} K(s)(I + L(s))^{-1}$$

$$T(s) \stackrel{\text{def}}{=} L(s)(I + L(s))^{-1} = I - S(s)$$

where the $L(s)$ is the loop transfer function matrix

$$L(s) = G(s)K(s). \quad (2-1)$$



Block Diagram of the Multivariable Feedback Control System

The two matrices $S(s)$ and $T(s)$ are known as the *sensitivity function* and *complementary sensitivity function*, respectively. The matrix $R(s)$ has no common name. The singular value Bode plots of each of the three transfer function matrices $S(s)$, $R(s)$, and $T(s)$ play an important role in robust multivariable control system design. The singular values of the loop transfer function matrix $L(s)$ are important because $L(s)$ determines the matrices $S(s)$ and $T(s)$.

Robustness in Terms of Singular Values

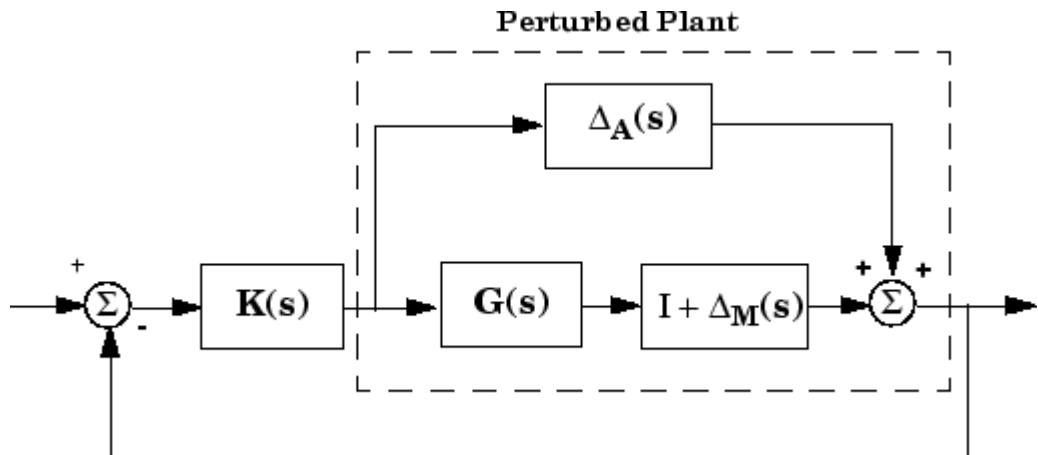
The singular values of $S(j\omega)$ determine the disturbance attenuation, because $S(s)$ is in fact the closed-loop transfer function from disturbance d to plant output y — see Block Diagram of the Multivariable Feedback Control System on page 2-6. Thus a disturbance attenuation performance specification can be written as

$$\bar{\sigma}(S(j\omega)) \leq |W_1^{-1}(j\omega)| \quad (2-2)$$

where $|W_1^{-1}(j\omega)|$ is the desired disturbance attenuation factor. Allowing $|W_1(j\omega)|$ to depend on frequency ω enables you to specify a different attenuation factor for each frequency ω .

The singular value Bode plots of $R(s)$ and of $T(s)$ are used to measure the stability margins of multivariable feedback designs in the face of additive plant perturbations Δ_A and multiplicative plant perturbations Δ_M , respectively. See the following figure.

Consider how the singular value Bode plot of complementary sensitivity $T(s)$ determines the stability margin for multiplicative perturbations Δ_M . The multiplicative stability margin is, by definition, the "size" of the smallest stable $\Delta_M(s)$ that destabilizes the system in the figure below when $\Delta_A = 0$.



Additive/Multiplicative Uncertainty

Taking $\bar{\sigma}(\Delta_M(j\omega))$ to be the definition of the "size" of $\Delta_M(j\omega)$, you have the following useful characterization of "multiplicative" stability robustness:

Multiplicative Robustness: The size of the smallest destabilizing multiplicative uncertainty $\Delta_M(s)$ is:

$$\bar{\sigma}(\Delta_M(j\omega)) = \frac{1}{\bar{\sigma}(T(j\omega))}.$$

The smaller is $\bar{\sigma}(T(j\omega))$, the greater will be the size of the smallest destabilizing multiplicative perturbation, and hence the greater will be the stability margin.

A similar result is available for relating the stability margin in the face of additive plant perturbations $\Delta_A(s)$ to $R(s)$ if you take $\bar{\sigma}(\Delta_A(j\omega))$ to be the definition of the "size" of $\Delta_A(j\omega)$ at frequency ω .

Additive Robustness: The size of the smallest destabilizing additive uncertainty Δ_A is:

$$\bar{\sigma}(\Delta_A(j\omega)) = \frac{1}{\bar{\sigma}(R(j\omega))}.$$

As a consequence of robustness theorems 1 and 2, it is common to specify the stability margins of control systems via singular value inequalities such as

$$\bar{\sigma}(R\{j\omega\}) \leq |W_2^{-1}(j\omega)| \quad (2-3)$$

$$\bar{\sigma}(T\{j\omega\}) \leq |W_3^{-1}(j\omega)| \quad (2-4)$$

where $|W_2(j\omega)|$ and $|W_3(j\omega)|$ are the respective sizes of the largest anticipated additive and multiplicative plant perturbations.

It is common practice to lump the effects of all plant uncertainty into a single fictitious multiplicative perturbation Δ_M , so that the control design requirements can be written

$$\frac{1}{\sigma_i(S(j\omega))} \geq |W_1(j\omega)|; \quad \bar{\sigma}_i(T[j\omega]) \leq |W_3^{-1}(j\omega)|$$

as shown in Singular Value Specifications on L, S, and T on page 2-12.

It is interesting to note that in the upper half of the figure (above the 0 dB line),

$$\underline{\sigma}(L(j\omega)) \approx \frac{1}{\bar{\sigma}(S(j\omega))}$$

while in the lower half of Singular Value Specifications on L, S, and T on page 2-12 (below the 0 dB line),

$$\underline{\sigma}(L(j\omega)) \approx \bar{\sigma}(T(j\omega)).$$

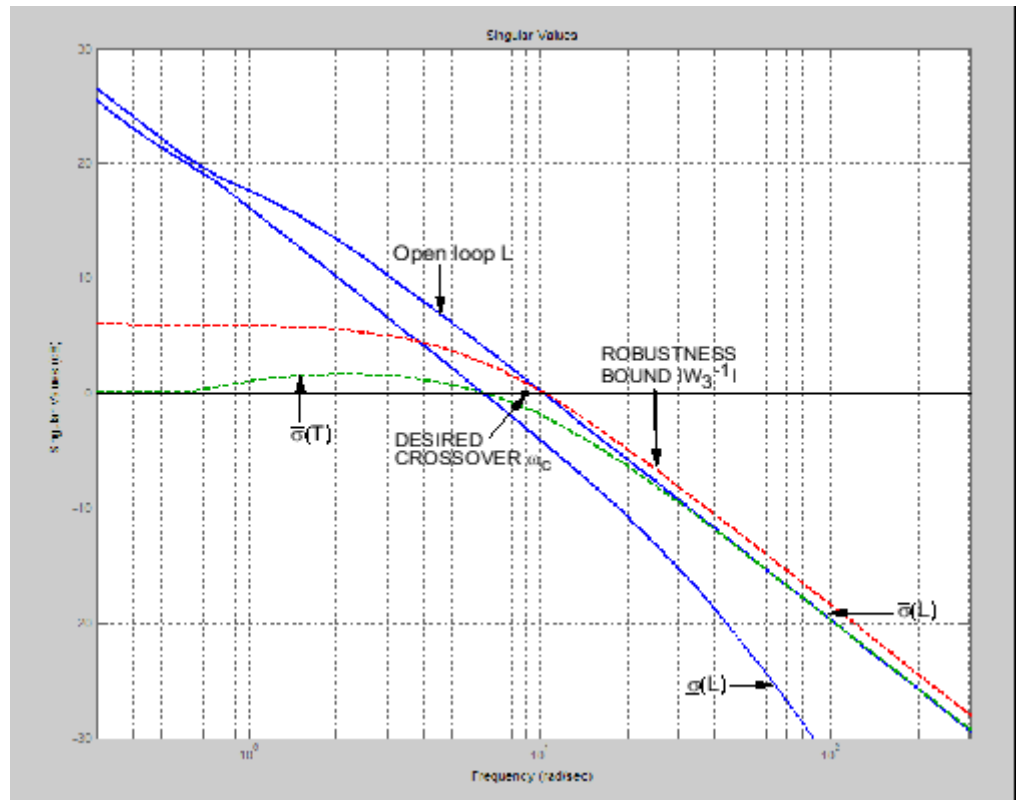
This results from the fact that

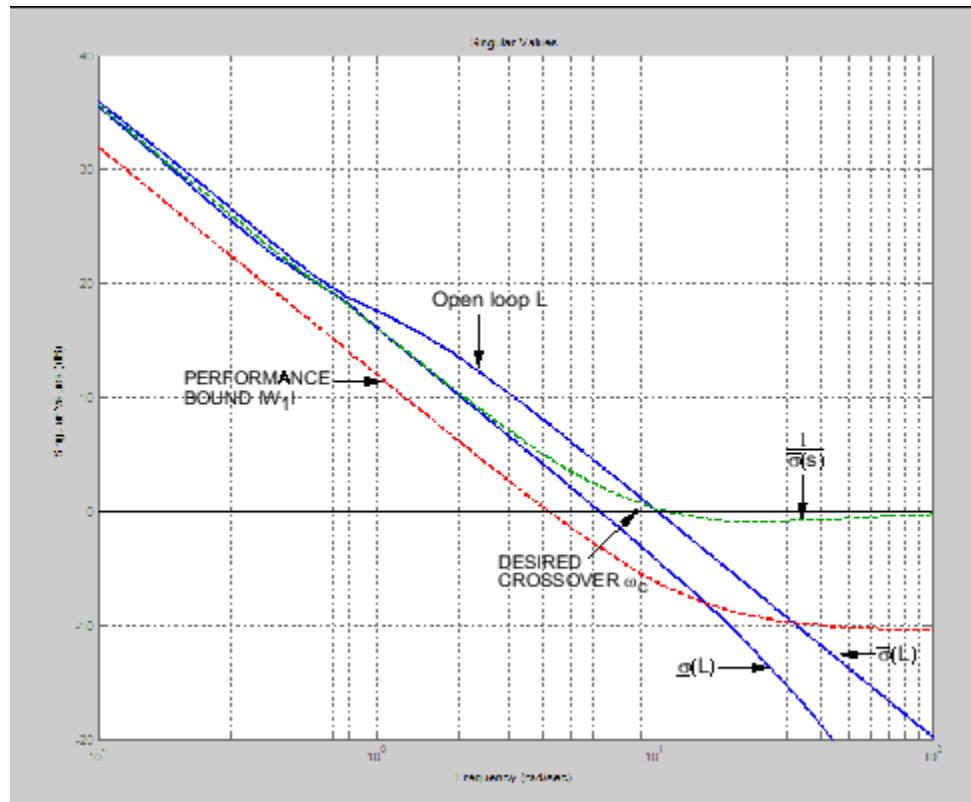
$$S(s) \stackrel{\text{def}}{=} (I + L(s))^{-1} \approx L(s)^{-1}$$

if $\underline{\sigma}(L(s)) \gg 1$, and

$$T(s) \stackrel{\text{def}}{=} L(s)(I + L(s))^{-1} \approx L(s)$$

if $\bar{\sigma}(L(s)) \gg 1$.





Singular Value Specifications on L, S, and T

Thus, it is not uncommon to see specifications on disturbance attenuation and multiplicative stability margin expressed directly in terms of forbidden regions for the Bode plots of $\sigma_i(L(j\omega))$ as "singular value loop shaping" requirements, either as specified upper/lower bounds or as a target desired loop shape — see the preceding figure.

Guaranteed Gain/Phase Margins in MIMO Systems

For those who are more comfortable with classical single-loop concepts, there are the important connections between the multiplicative stability margins predicted by $\bar{\sigma}(T)$ and those predicted by classical M -circles, as found on the Nichols chart. Indeed in the single-input/single-output case,

$$\bar{\sigma}(T(j\omega)) = \left| \frac{L(j\omega)}{1+L(j\omega)} \right|$$

which is precisely the quantity you obtain from Nichols chart M -circles. Thus,

$\|T\|_\infty$ is a multiloop generalization of the closed-loop resonant peak magnitude which, as classical control experts will recognize, is closely related to the damping ratio of the dominant closed-loop poles. Also, it turns out that you

can relate $\|T\|_\infty$, $\|S\|_\infty$ to the classical gain margin G_M and phase margin θ_M in each feedback loop of the multivariable feedback system of Block Diagram of the Multivariable Feedback Control System on page 2-6 via the formulas:

$$G_M \geq 1 + \frac{1}{\|T\|_\infty}$$

$$G_M \geq 1 + \frac{1}{1 - \frac{1}{\|S\|_\infty}}$$

$$\theta_M \geq 2 \sin^{-1} \left(\frac{1}{2\|T\|_\infty} \right)$$

$$\theta_M \geq 2 \sin^{-1} \left(\frac{1}{2\|T\|_\infty} \right).$$

(See [6].) These formulas are valid provided $\|S\|_\infty$ and $\|T\|_\infty$ are larger than 1, as is normally the case. The margins apply even when the gain perturbations or phase perturbations occur simultaneously in several feedback channels.

The infinity norms of S and T also yield gain reduction tolerances. The *gain reduction tolerance* g_m is defined to be the minimal amount by which the gains in each loop would have to be *decreased* in order to destabilize the system.

Upper bounds on g_m are as follows:

$$g_M \leq 1 - \frac{1}{\|T\|_\infty}$$
$$g_M \leq \frac{1}{1 + \frac{1}{\|S\|_\infty}}.$$

Using LOOPSYN to Do H-Infinity Loop Shaping

The command `loopsyn` lets you design a stabilizing feedback controller to optimally shape the open loop frequency response of a MIMO feedback control system to match as closely as possible a desired loop shape G_d — see the preceding figure. The basic syntax of the `loopsyn` loop-shaping controller synthesis command is:

```
K = loopsyn(G,Gd)
```

Here G is the LTI transfer function matrix of a MIMO plant model, G_d is the target desired loop shape for the loop transfer function $L=G*K$, and K is the optimal loop-shaping controller. The LTI controller K has the property that it shapes the loop $L=G*K$ so that it matches the frequency response of G_d as closely as possible, subject to the constraint that the compensator must stabilize the plant model G .

Loop-Shaping Control Design of Aircraft Model

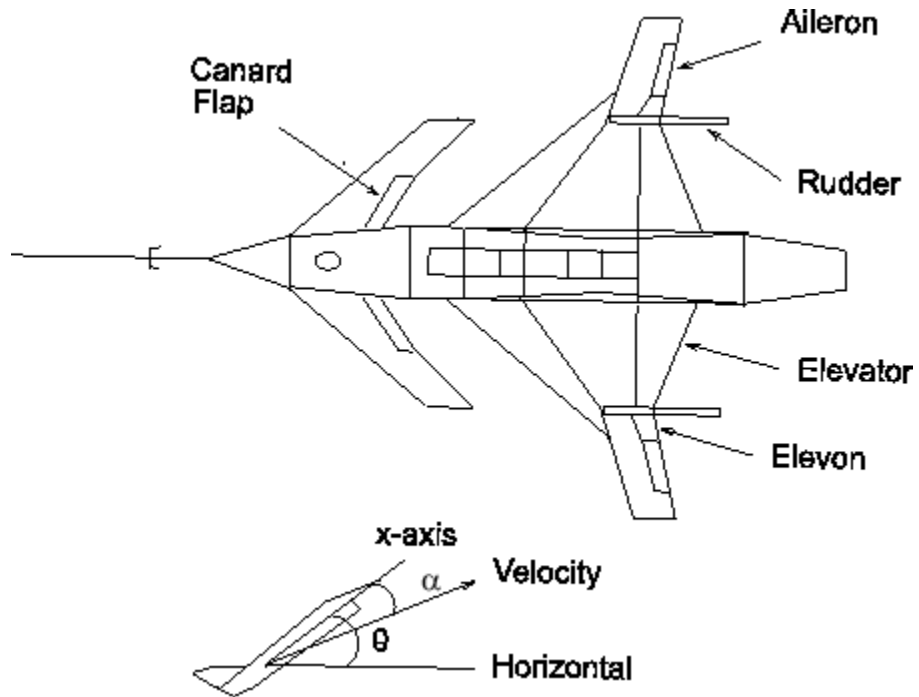
To see how the `loopsyn` command works in practice to address robustness and performance tradeoffs, consider again the NASA HiMAT aircraft model taken from the paper of Safonov, Laub, and Hartmann [8]. The longitudinal dynamics of the HiMAT aircraft trimmed at 25000 ft and 0.9 Mach are unstable and have two right-half-plane phugoid modes. The linear model has state-space realization $G(s) = C(Is - A)^{-1}B$ with six states, with the first four states representing angle of attack (α) and attitude angle (θ) and their rates of change, and the last two representing elevon and canard control actuator dynamics — see Aircraft Configuration and Vertical Plane Geometry on page 2-17.

```

ag = [
-2.2567e-02  -3.6617e+01  -1.8897e+01  -3.2090e+01  3.2509e+00  -7.6257e-01;
9.2572e-05  -1.8997e+00  9.8312e-01  -7.2562e-04  -1.7080e-01  -4.9652e-03;
1.2338e-02  1.1720e+01  -2.6316e+00  8.7582e-04  -3.1604e+01  2.2396e+01;
0           0           1.0000e+00           0           0           0;
0           0           0           0           -3.0000e+01  0;
0           0           0           0           0           -3.0000e+01];
bg = [0  0;
      0  0;
      0  0;
      0  0;
      30 0;
      0 30];
cg = [0  1  0  0  0  0;
      0  0  0  1  0  0];
dg = [0  0;
      0  0];
G=ss(ag,bg,cg,dg);

```

The control variables are elevon and canard actuators (δ_e and δ_c). The output variables are angle of attack (α) and attitude angle (θ).



Aircraft Configuration and Vertical Plane Geometry

This model is good at frequencies below 100 rad/s with less than 30% variation between the true aircraft and the model in this frequency range. However as noted in [8], it does not reliably capture very high-frequency behaviors, because it was derived by treating the aircraft as a rigid body and neglecting lightly damped fuselage bending modes that occur at somewhere between 100 and 300 rad/s. These unmodeled bending modes might cause as much as 20 dB deviation (i.e., 1000%) between the frequency response of the model and the actual aircraft for frequency $\omega > 100$ rad/s. Other effects like control actuator time delays and fuel sloshing also contribute to model inaccuracy at even higher frequencies, but the dominant unmodeled effects are the fuselage bending modes. You can think of these unmodeled bending modes as multiplicative uncertainty of size 20 dB, and design your controller using loopsyn, by making sure that the loop has gain less than -20 dB at, and beyond, the frequency $\omega > 100$ rad/s.

Design Specifications

The singular value design specifications are

- **Robustness Spec.:** -20 dB/decade roll-off slope and -20 dB loop gain at 100 rad/s
- **Performance Spec.:** Minimize the sensitivity function as much as possible.

Both specs can be accommodated by taking as the desired loop shape

$$G_d(s)=8/s$$

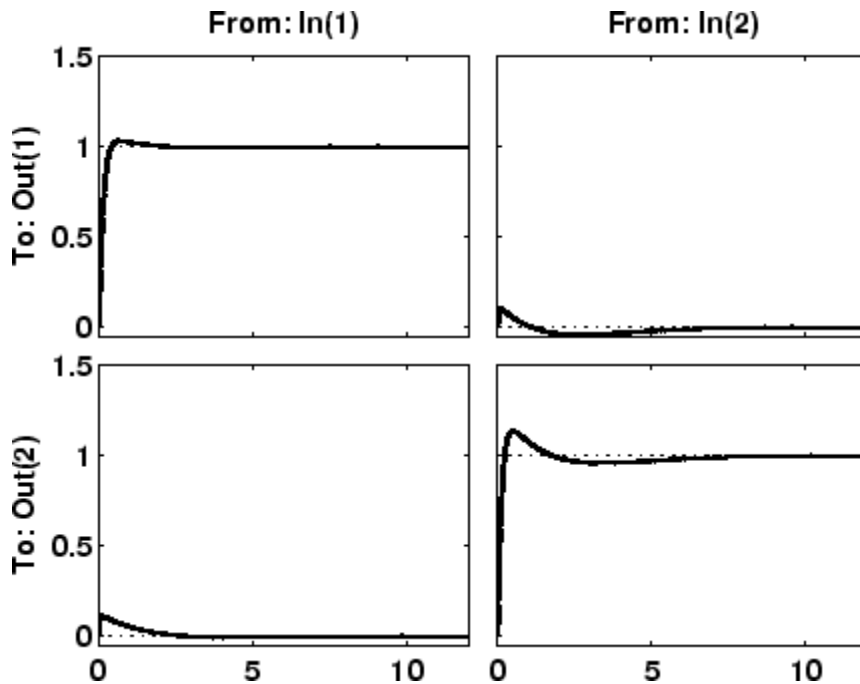
MATLAB Commands for a LOOPSYN Design

```
%% Enter the desired loop shape Gd
s=zpk('s'); % Laplace variable s
Gd=8/s; % desired loop shape
%% Compute the optimal loop shaping controller K
[K,CL,GAM]=loopsyn(G,Gd);
%% Compute the loop L, sensitivity S and
%% complementary sensitivity T:
L=G*K;
I=eye(size(L));
S=feedback(I,L); % S=inv(I+L);
T=I-S;
%% Plot the results:
% step response plots
step(T);title('\alpha and \theta command step responses');
% frequency response plots
figure;
sigma(I+L,'--',T,':',L,'r--',Gd,'k-',Gd/GAM,'k:',...
    Gd*GAM,'k',{.1,100});grid
legend('1/\sigma(S) performance',...
    '\sigma(T) robustness',...
    '\sigma(L) loopshape',...
    '\sigma(Gd) desired loop',...
    '\sigma(Gd) \pm GAM, dB');
```

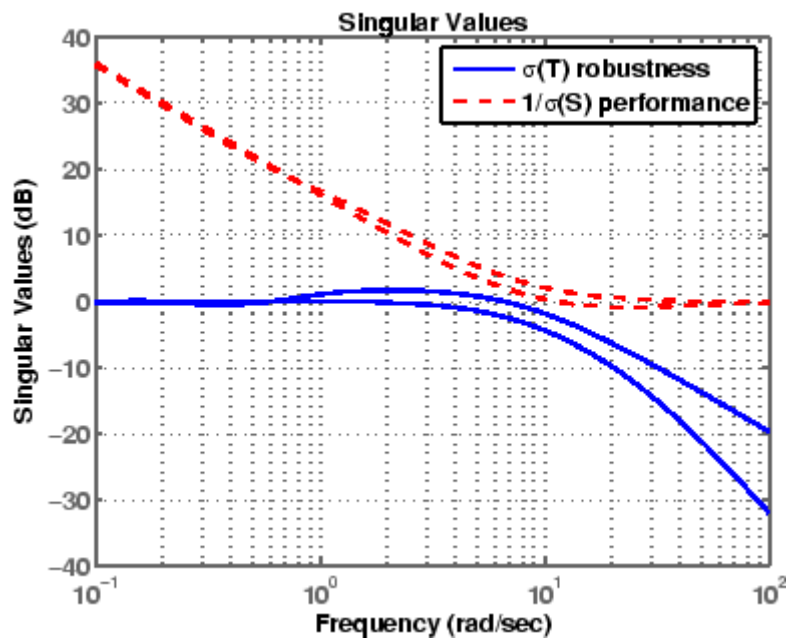
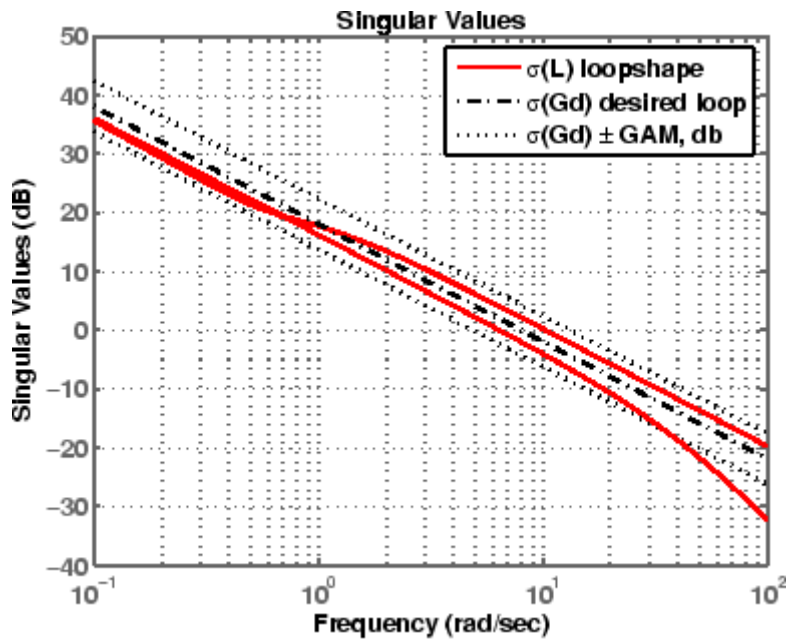
The plots of the resulting step and frequency response for the NASA HiMAT loopsyn loop-shaping controller design are shown in the following figure. The number $\pm \text{GAM}$, dB (i.e., $20\log_{10}(\text{GAM})$) tells you the accuracy with which your loopsyn control design matches the target desired loop:

$$\bar{\sigma}(GK), \text{db} \geq |G_d|, \text{db} - \text{GAM}, \text{db} \quad (\omega < \omega_c)$$

$$\bar{\sigma}(GK), \text{db} \geq |G_d|, \text{db} + \text{GAM}, \text{db} \quad (\omega > \omega_c).$$



HiMAT Closed Loop Step Responses



LOOPSYN Design Results for NASA HiMAT

Fine-Tuning the LOOPSYN Target Loop Shape G_d to Meet Design Goals

If your first attempt at loopsyn design does not achieve everything you wanted, you will need to readjust your target desired loop shape G_d . Here are some basic design tradeoffs to consider:

- **Stability Robustness.** Your target loop G_d should have low gain (as small as possible) at high frequencies where typically your plant model is so poor that its phase angle is completely inaccurate, with errors approaching $\pm 180^\circ$ or more.
- **Performance.** Your G_d loop should have high gain (as great as possible) at frequencies where your model is good, in order to ensure good control accuracy and good disturbance attenuation.
- **Crossover and Roll-Off.** Your desired loop shape G_d should have its 0 dB crossover frequency (denoted ω_c) between the above two frequency ranges, and below the crossover frequency ω_c it should roll off with a negative slope of between -20 and -40 dB/decade, which helps to keep phase lag to less than -180° inside the control loop bandwidth ($0 < \omega < \omega_c$).

Other considerations that might affect your choice of G_d are the right-half-plane poles and zeros of the plant G , which impose fundamental limits on your 0 dB crossover frequency ω_c [12]. For instance, your 0 dB crossover ω_c must be greater than the magnitude of any plant right-half-plane poles and less than the magnitude of any right-half-plane zeros.

$$\max_{\operatorname{Re}(p_i) > 0} |p_i| < \omega_c < \min_{\operatorname{Re}(z_i) > 0} |z_i|.$$

If you do not take care to choose a target loop shape G_d that conforms to these fundamental constraints, then loopsyn will still compute the optimal loop-shaping controller K for your G_d , but you should expect that the optimal loop $L = G * K$ will have a poor fit to the target loop shape G_d , and consequently it might be impossible to meet your performance goals.

Mixed-Sensitivity Loop Shaping

A popular alternative approach to `loopsyn` loop shaping is H_∞ *mixed-sensitivity* loop shaping, which is implemented by the Robust Control Toolbox software command:

```
K=mixsyn(G,W1,[],W3)
```

With `mixsyn` controller synthesis, your performance and stability robustness specifications equations (2-2) and (2-4) are combined into a single infinity norm specification of the form

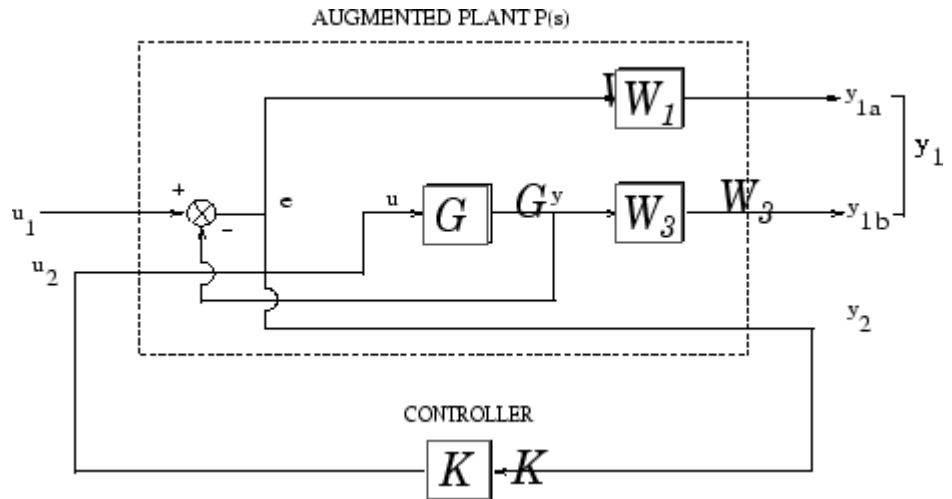
$$\|T_{y_1 u_1}\|_\infty \leq 1$$

where (see MIXSYN H_∞ Mixed-Sensitivity Loop Shaping $T_{y_1 u_1}$ on page 2-23):

$$T_{y_1 u_1} \stackrel{\text{def}}{=} \begin{bmatrix} W_1 & S \\ W_3 & T \end{bmatrix}.$$

The term $\|T_{y_1 u_1}\|_\infty$ is called a *mixed-sensitivity cost function* because it penalizes both sensitivity $S(s)$ and complementary sensitivity $T(s)$. Loop shaping is achieved when you choose W_1 to have the target loop shape for frequencies $\omega < \omega_c$, and you choose $1/W_3$ to be the target for $\omega > \omega_c$. In choosing design specifications W_1 and W_3 for a `mixsyn` controller design, you need to ensure that your 0 dB crossover frequency for the Bode plot of W_1 is below the 0 dB crossover frequency of $1/W_3$, as shown in Singular Value Specifications on L, S, and T on page 2-12, so that there is a gap for the desired loop shape Gd to pass between the performance bound W_1 and your robustness bound

W_3^{-1} . Otherwise, your performance and robustness requirements will not be achievable.



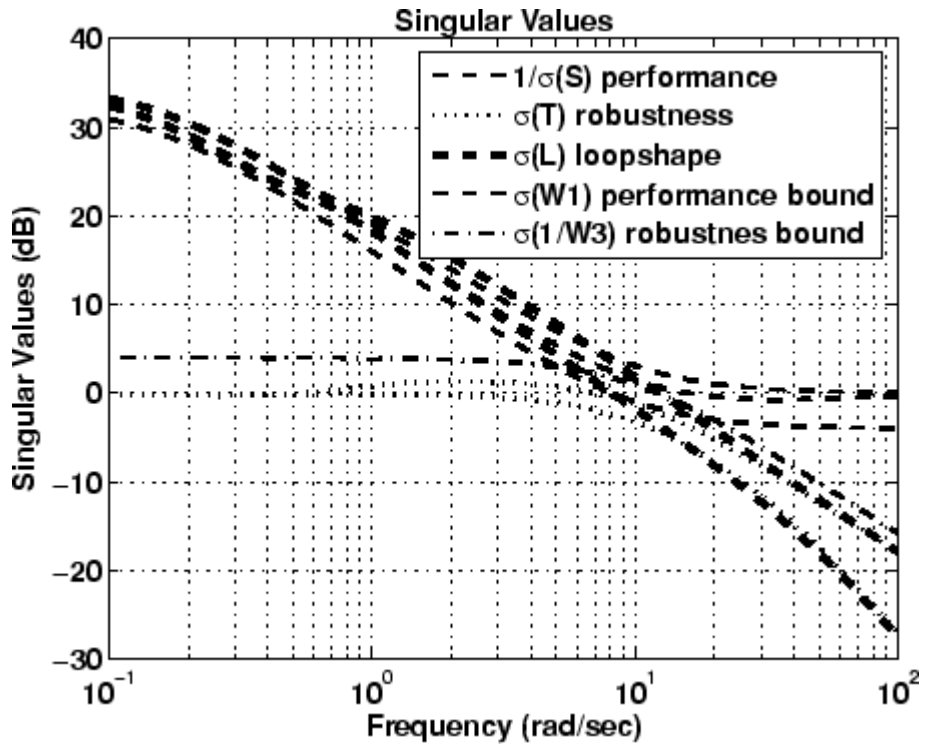
MIXSYN H. Mixed-Sensitivity Loop Shaping $T_{y_1 u_1}$

Mixed-Sensitivity Loop-Shaping Controller Design

To do a H_∞ mixed-sensitivity synthesis design on the HiMAT model, start with the plant model G created in “Mixed-Sensitivity Loop-Shaping Controller Design” on page 2-24 and type the following commands:

```
% Set up the performance and robustness bounds W1 & W3
s=zpk('s'); % Laplace variable s
MS=2;AS=.03;WS=5;
W1=(s/MS+WS)/(s+AS*WS);
MT=2;AT=.05;WT=20;
W3=(s+WT/MT)/(AT*s+WT);
% Compute the H-infinity mixed-sensitivity optimal controller K1
[K1,CL1,GAM1]=mixsyn(G,W1,[],W3);
% Next compute and plot the closed-loop system.
% Compute the loop L1, sensitivity S1, and comp sensitivity T1:
L1=G*K1;
I=eye(size(L1));
S1=feedback(I,L1); % S=inv(I+L1);
T1=I-S1;
% Plot the results:
% step response plots
step(T1,1.5);
title('\alpha and \theta command step responses');
% frequency response plots
figure;
sigma(I+L1,'- -',T1,': ',L1,'r- -',... W1/GAM1,'k- -',GAM1/W3,'k- .',{.1,100});grid
legend('1/\sigma(S) performance',...
'\sigma(T) robustness',...
'\sigma(L) loopshape',...
'\sigma(W1) performance bound',...
'\sigma(1/W3) robustness bound');
```

The resulting mixsyn singular value plots for the NASA HiMAT model are shown below.

**MIXSYN Design Results for NASA HiMAT**

Loop-Shaping Commands

Robust Control Toolbox software gives you several choices for shaping the frequency response properties of multiinput/multioutput (MIMO) feedback control loops. Some of the main commands that you are likely to use for loop-shaping design, and associated utility functions, are listed below:

MIMO Loop-Shaping Commands	
loopsyn	H_∞ loop-shaping controller synthesis
ltrsyn	LQG loop-transfer recovery
mixsyn	H_∞ mixed-sensitivity controller synthesis
ncfsyn	Glover-McFarlane H_∞ normalized coprime factor loop-shaping controller synthesis
MIMO Loop-Shaping Utility Functions	
augw	Augmented plant for weighted H_2 and H_∞ mixed-sensitivity control synthesis
makeweight	Weights for H_∞ mixed sensitivity (mixsyn, augw)
sigma	Singular value plots of LTI feedback loops

Model Reduction for Robust Control

- “Why Reduce Model Order?” on page 3-2
- “Hankel Singular Values” on page 3-3
- “Model Reduction Techniques” on page 3-5
- “Approximate Plant Model by Additive Error Methods” on page 3-7
- “Approximate Plant Model by Multiplicative Error Method” on page 3-9
- “Using Modal Algorithms” on page 3-11
- “Reducing Large-Scale Models” on page 3-14
- “Normalized Coprime Factor Reduction” on page 3-15
- “Bibliography” on page 3-16

Why Reduce Model Order?

In the design of robust controllers for complicated systems, model reduction fits several goals:

- 1** To simplify the best available model in light of the purpose for which the model is to be used—namely, to design a control system to meet certain specifications.
- 2** To speed up the simulation process in the design validation stage, using a smaller size model with most of the important system dynamics preserved.
- 3** Finally, if a modern control method such as LQG or H_∞ is used for which the complexity of the control law is not explicitly constrained, the order of the resultant controller is likely to be considerably greater than is truly needed. A good model reduction algorithm applied to the control law can sometimes significantly reduce control law complexity with little change in control system performance.

Model reduction routines in this toolbox can be put into two categories:

- **Additive error method** — The reduced-order model has an additive error bounded by an error criterion.
- **Multiplicative error method** — The reduced-order model has a multiplicative or relative error bounded by an error criterion.

The error is measured in terms of peak gain across frequency (H_∞ norm), and the error bounds are a function of the neglected Hankel singular values.

Hankel Singular Values

In control theory, eigenvalues define a system stability, whereas *Hankel singular values* define the “energy” of each state in the system. Keeping larger energy states of a system preserves most of its characteristics in terms of stability, frequency, and time responses. Model reduction techniques presented here are all based on the Hankel singular values of a system. They can achieve a reduced-order model that preserves the majority of the system characteristics.

Mathematically, given a *stable* state-space system (A,B,C,D) , its Hankel singular values are defined as [1]

$$\sigma_H = \sqrt{\lambda_i(PQ)}$$

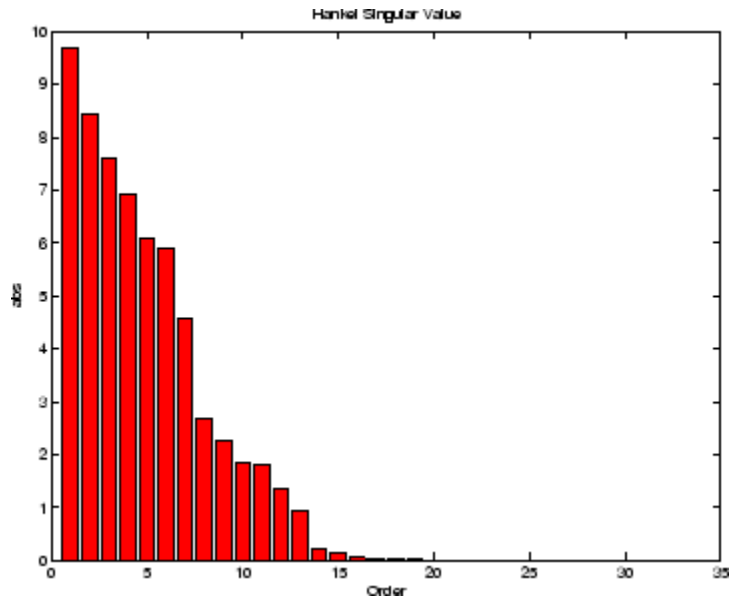
where P and Q are *controllability* and *observability grammians* satisfying

$$\begin{aligned}AP + PA^T &= -BB^T \\ A^T Q + QA &= -C^T C.\end{aligned}$$

For example,

```
rand('state',1234); randn('state',5678);  
G = rss(30,4,3);  
hankelsv(G)
```

returns a Hankel singular value plot as follows:



which shows that system G has most of its “energy” stored in states 1 through 15 or so. Later, you will see how to use model reduction routines to keep a 15-state reduced model that preserves most of its dynamic response.

Model Reduction Techniques

Robust Control Toolbox software offers several algorithms for model approximation and order reduction. These algorithms let you control the absolute or relative approximation error, and are all based on the Hankel singular values of the system.

Robust control theory quantifies a system uncertainty as either *additive* or *multiplicative* types. These model reduction routines are also categorized into two groups: *additive error* and *multiplicative error* types. In other words, some model reduction routines produce a reduced-order model G_{red}

of the original model G with a bound on the error $\|G - G_{red}\|_{\infty}$, the peak gain across frequency. Others produce a reduced-order model with a bound on

the relative error $\|G^{-1}(G - G_{red})\|_{\infty}$.

These theoretical bounds are based on the “tails” of the Hankel singular values of the model, i.e.,

Additive Error Bound

$$\|G - G_{red}\|_{\infty} \leq 2 \sum_{k=1}^n \sigma_k \quad (3-1)$$

where σ_i are denoted the i th Hankel singular value of the original system G .

Multiplicative (Relative) Error Bound

$$\|G^{-1}(G - G_{red})\|_{\infty} \leq \prod_{k=1}^n \left(1 + 2\sigma_k \left(\sqrt{1 + \sigma_k^2} + \sigma_k \right) \right) - 1 \quad (3-2)$$

where σ_i are denoted the i th Hankel singular value of the phase matrix of the model G (see the `bstmr` reference page).

Top-Level Model Reduction Command

Method	Description
reduce	Main interface to model approximation algorithms

Normalized Coprime Balanced Model Reduction Command

Method	Description
ncfmr	Normalized coprime balanced truncation

Additive Error Model Reduction Commands

Method	Description
balancmr	Square-root balanced model truncation
schurmr	Schur balanced model truncation
hankelmr	Hankel minimum degree approximation

Multiplicative Error Model Reduction Command

Method	Description
bstmr	Balanced stochastic truncation

Additional Model Reduction Tools

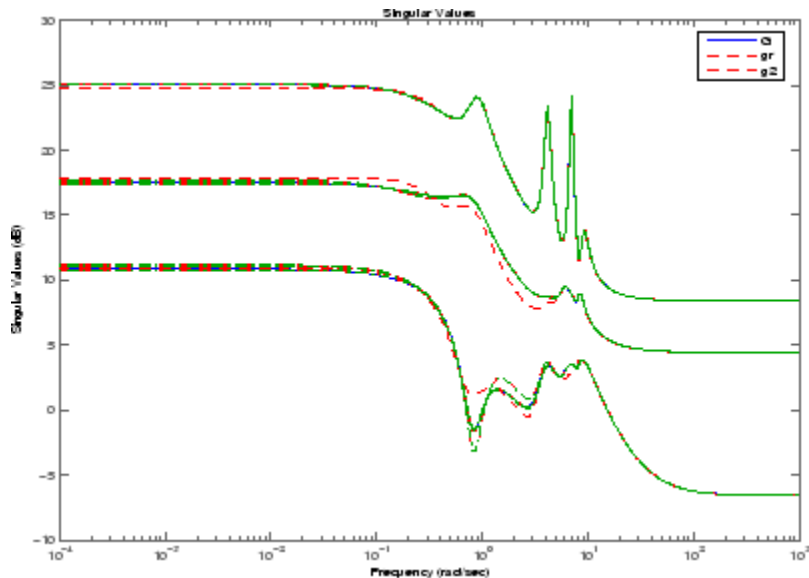
Method	Description
modreal	Modal realization and truncation
slowfast	Slow and fast state decomposition
stabsep	Stable and antistable state projection

Approximate Plant Model by Additive Error Methods

Given a system in LTI form, the following commands reduce the system to any desired order you specify. The judgment call is based on its Hankel singular values as shown in the previous paragraph.

```
rand('state',1234); randn('state',5678);
G = rss(30,4,3);
% balanced truncation to models with sizes 12:16
[g1,info1] = balancmr(G,12:16); % or use reduce
% Schur balanced truncation by specifying `MaxError`
[g2,info2] = schurmr(G,'MaxError',[1,0.8,0.5,0.2]);
sigma(G,'b-',g1,'r--',g2,'g-.')
```

shows a comparison plot of the original model G and reduced models g1 and g2.

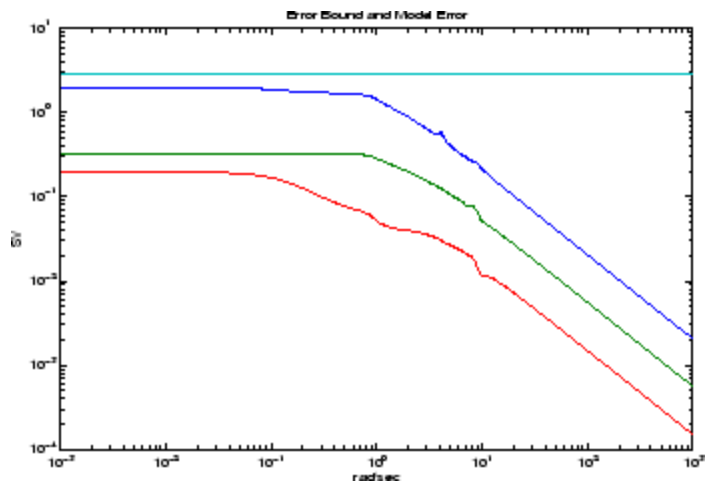


To determine whether the theoretical error bound is satisfied,

```
norm(G-g1(:, :, 1), 'inf') % 2.0123
info1.ErrorBound(1) % 2.8529
```

or plot the model error vs. error bound via the following commands:

```
[sv,w] = sigma(G-g1(:,:,1));  
loglog(w,sv,w,info1.ErrorBound(1)*ones(size(w)))  
xlabel('rad/sec');ylabel('SV');  
title('Error Bound and Model Error')
```

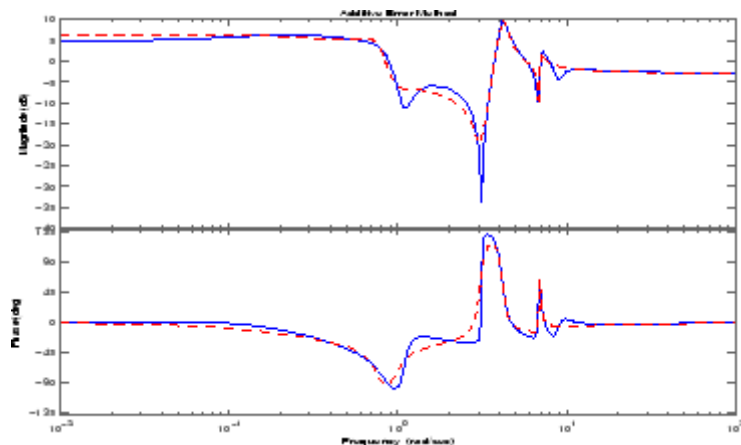


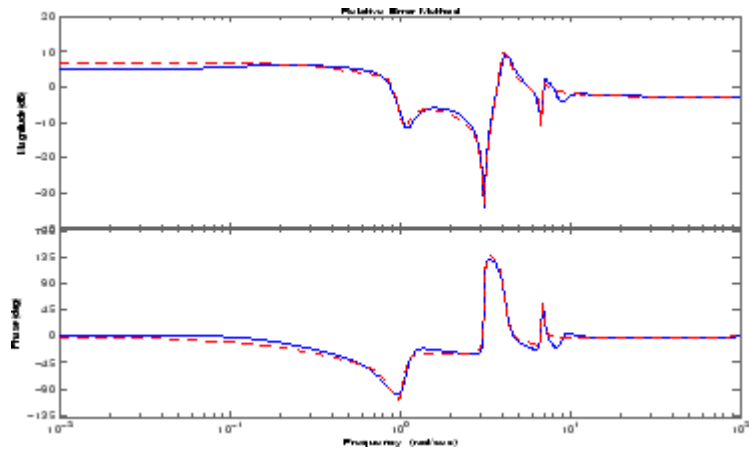
Approximate Plant Model by Multiplicative Error Method

In most cases, *multiplicative error* model reduction method `bstmr` tends to bound the relative error between the original and reduced-order models across the frequency range of interest, hence producing a more accurate reduced-order model than the *additive error* methods. This characteristic is obvious in system models with low damped poles.

The following commands illustrate the significance of a multiplicative error model reduction method as compared to any additive error type. Clearly, the phase-matching algorithm using `bstmr` provides a better fit in the Bode plot.

```
rand('state',1234); randn('state',5678); G = rss(30,1,1);
[gr,infor] = reduce(G,'algo','balance','order',7);
[gs,infos] = reduce(G,'algo','bst','order',7);
figure(1);bode(G,'b-',gr,'r--');
title('Additive Error Method')
figure(2);bode(G,'b-',gs,'r--');
title('Relative Error Method')
```





Therefore, for some systems with low damped poles/zeros, the balanced stochastic method (`bstmr`) produces a better reduced-order model fit in those frequency ranges to make multiplicative error small. Whereas additive error methods such as `balancmr`, `schurmr`, or `hanke1mr` only care about minimizing the overall “absolute” peak error, they can produce a reduced-order model missing those low damped poles/zeros frequency regions.

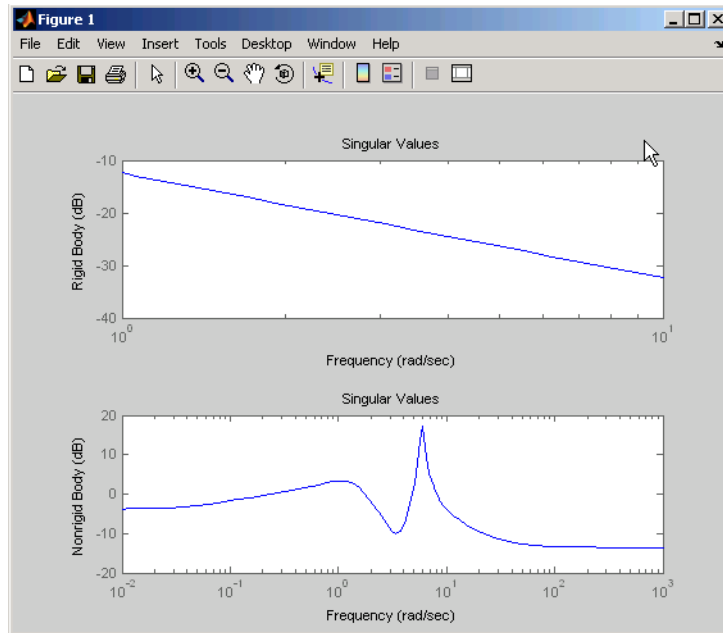
Using Modal Algorithms

Rigid Body Dynamics

In many cases, a model's $j\omega$ -axis poles are important to keep after model reduction, e.g., rigid body dynamics of a flexible structure plant or integrators of a controller. A unique routine, `modreal`, serves the purpose nicely.

`modreal` puts a system into its *modal form*, with eigenvalues appearing on the diagonal of its A-matrix. Real eigenvalues appear in 1-by-1 blocks, and complex eigenvalues appear in 2-by-2 real blocks. All the blocks are ordered in ascending order, based on their eigenvalue magnitudes, by default, or descending order, based on their real parts. Therefore, specifying the number of $j\omega$ -axis poles splits the model into two systems with one containing only $j\omega$ -axis dynamics, the other containing the non- $j\omega$ axis dynamics.

```
rand('state',5678); randn('state',1234); G = rss(30,1,1);
[Gjw,G2] = modreal(G,1); % only one rigid body dynamics
G2.d = Gjw.d; Gjw.d = 0; % put DC gain of G into G2
subplot(211);sigma(Gjw);ylabel('Rigid Body')
subplot(212);sigma(G2);ylabel('Nonrigid Body')
```



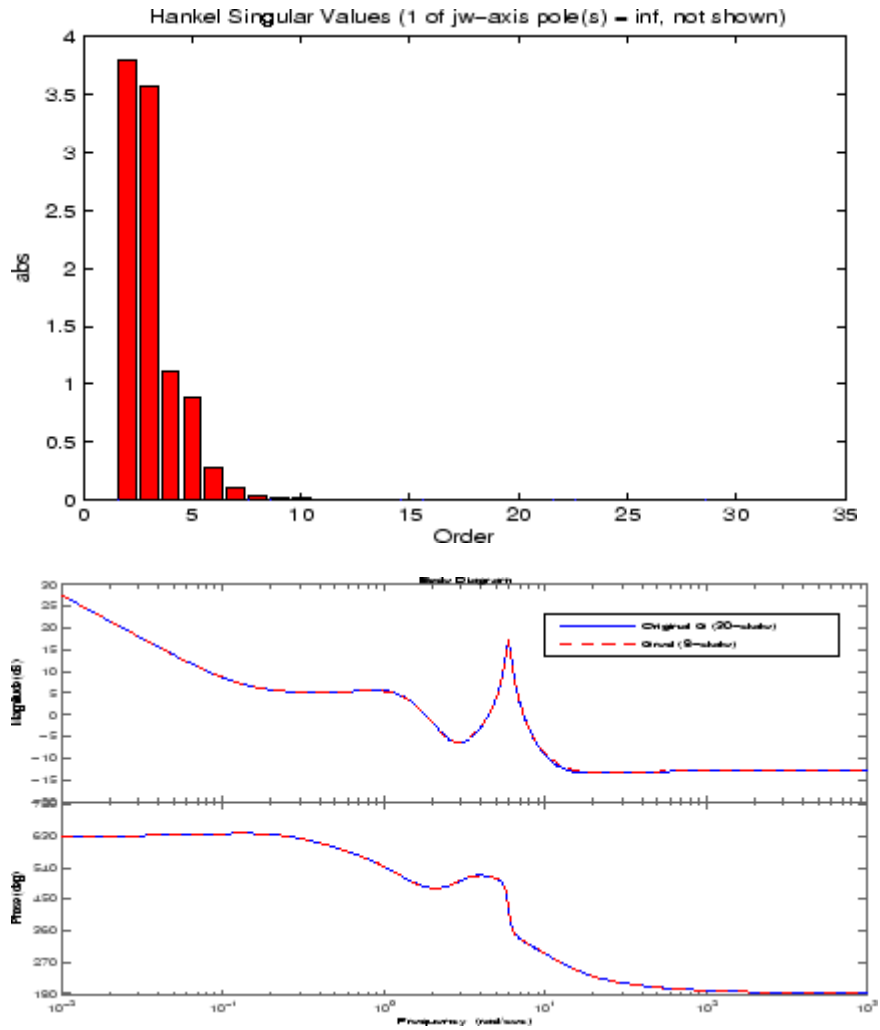
Further model reduction can be done on G2 without any numerical difficulty. After G2 is further reduced to Gred, the final approximation of the model is simply G_{jw}+Gred.

This process of splitting $j\omega$ -axis poles has been built in and automated in all the model reduction routines (balancmr, schurmr, hankelmr, bstmr, hankelsv) so that users need not worry about splitting the model.

The following single command creates a size 8 reduced-order model from its original 30-state model:

```
rand('state',5678); randn('state',1234); G = rss(30,1,1);
[gr,info] = reduce(G); % choose a size of 8 at prompt
bode(G,'b-',gr,'r--')
```

Without specifying the size of the reduced-order model, a Hankel singular value plot is shown below.



The default algorithm `balancmr` of `reduce` has done a great job of approximating a 30-state model with just eight states. Again, the rigid body dynamics are preserved for further controller design.

Reducing Large-Scale Models

For some really large size problems (states > 200), `modreal` turns out to be the only way to start the model reduction process. Because of the size and numerical properties associated with those large size, and low damped dynamics, most Hankel based routines can fail to produce a good reduced-order model.

`modreal` puts the large size dynamics into the modal form, then truncates the dynamic model to an intermediate stage model with a comfortable size of 50 or so states. From this point on, those more sophisticated Hankel singular value based routines can further reduce this intermediate stage model, in a much more accurate fashion, to a smaller size for final controller design.

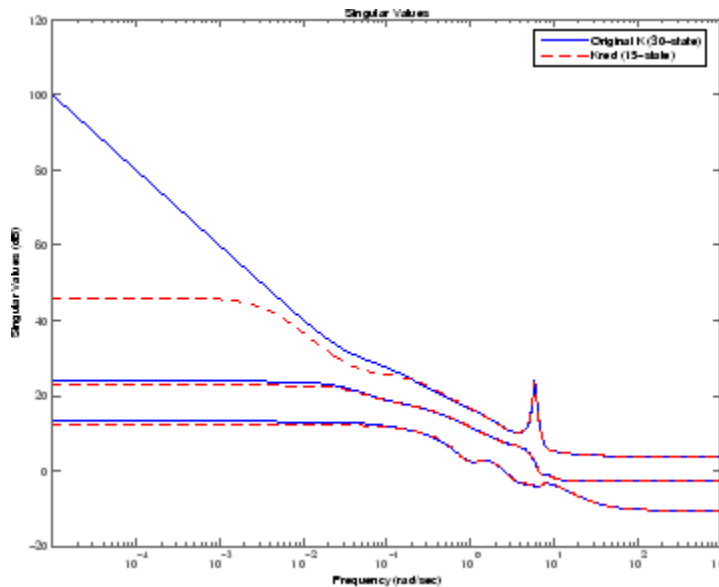
For a typical 240-state flexible spacecraft model in the spacecraft industry, applying `modreal` and `bstmr` (or any other additive routines) in sequence can reduce the original 240-state plant dynamics to a seven-state three-axis model including rigid body dynamics. Any modern robust control design technique mentioned in this toolbox can then be easily applied to this smaller size plant for a controller design.

Normalized Coprime Factor Reduction

A special model reduction routine `ncfmr` produces a reduced-order model by truncating a balanced coprime set of a given model. It can directly simplify a modern controller with integrators to a smaller size by balanced truncation of the normalized coprime factors. It does not need `modreal` for pre-/postprocessing as the other routines do. However, the integrators will not be preserved.

```
rand('state',5678); randn('state',1234);
K= rss(30,4,3); % The same model G used in the 1st example
[Kred,info2] = ncfmr(K);
sigma(K,Kred)
```

Again, without specifying the size of the reduced-order model, any model reduction routine presented here will plot a Hankel singular value bar chart and prompt you for a reduced model size. In this case, enter 15.



If integral control is important, previously mentioned methods (except `ncfmr`) can nicely preserve the original integrator(s) in the model.

Bibliography

- [1] Glover, K., "All Optimal Hankel Norm Approximation of Linear Multivariable Systems, and Their L^∞ - Error Bounds," *Int. J. Control*, Vol. 39, No. 6, 1984, pp. 1145-1193.
- [2] Zhou, K., Doyle, J.C., and Glover, K., *Robust and Optimal Control*, Englewood Cliffs, NJ, Prentice Hall, 1996.
- [3] Safonov, M.G., and Chiang, R.Y., "A Schur Method for Balanced Model Reduction," *IEEE Trans. on Automat. Contr.*, Vol. 34, No. 7, July 1989, pp. 729-733.
- [4] Safonov, M.G., Chiang, R.Y., and Limebeer, D.J.N., "Optimal Hankel Model Reduction for Nonminimal Systems," *IEEE Trans. on Automat. Contr.*, Vol. 35, No. 4, April 1990, pp. 496-502.
- [5] Safonov, M.G., and Chiang, R.Y., "Model Reduction for Robust Control: A Schur Relative Error Method," *International J. of Adaptive Control and Signal Processing*, Vol. 2, 1988, pp. 259-272.
- [6] Obinata, G., and Anderson, B.D.O., *Model Reduction for Control System Design*, London, Springer-Verlag, 2001.

Robustness Analysis

- “Create Models of Uncertain Systems” on page 4-2
- “Robust Controller Design” on page 4-9
- “MIMO Robustness Analysis” on page 4-14
- “Summary of Robustness Analysis Tools” on page 4-25

Create Models of Uncertain Systems

Dealing with and understanding the effects of uncertainty are important tasks for the control engineer. Reducing the effects of some forms of uncertainty (initial conditions, low-frequency disturbances) without catastrophically increasing the effects of other dominant forms (sensor noise, model uncertainty) is the primary job of the feedback control system.

Closed-loop stability is the way to deal with the (always present) uncertainty in initial conditions or arbitrarily small disturbances.

High-gain feedback in low-frequency ranges is a way to deal with the effects of unknown biases and disturbances acting on the process output. In this case, you are forced to use roll-off filters in high-frequency ranges to deal with high-frequency sensor noise in a feedback system.

Finally, notions such as gain and phase margins (and their generalizations) help quantify the sensitivity of stability and performance in the face of *model uncertainty*, which is the imprecise knowledge of how the control input directly affects the feedback variables.

Robust Control Toolbox software has built-in features allowing you to specify model uncertainty simply and naturally. The primary building blocks, called *uncertain elements* (or uncertain Control Design Blocks) are uncertain real parameters and uncertain linear, time-invariant objects. These can be used to create coarse and simple or detailed and complex descriptions of the model uncertainty present within your process models.

Once formulated, high-level system robustness tools can help you analyze the potential degradation of stability and performance of the closed-loop system brought on by the system model uncertainty.

Creating Uncertain Models of Dynamic Systems

The two dominant forms of model uncertainty are as follows:

- Uncertainty in parameters of the underlying differential equation models

- Frequency-domain uncertainty, which often quantifies model uncertainty by describing absolute or relative uncertainty in the process's frequency response

Using these two basic building blocks, along with conventional system creation commands (such as `ss` and `tf`), you can easily create uncertain system models.

Creating Uncertain Parameters

An uncertain parameter has a name (used to identify it within an uncertain system with many uncertain parameters) and a nominal value. Being uncertain, it also has variability, described in one of the following ways:

- An additive deviation from the nominal
- A range about the nominal
- A percentage deviation from the nominal

Create a real parameter, with name 'bw', nominal value 5, and a percentage uncertainty of 10%.

```
bw = ureal('bw',5,'Percentage',10)
```

This creates a `ureal` object. View its properties using the `get` command.

```
Uncertain Real Parameter: Name bw, NominalValue 5, variability = [-10 10]%
get(bw)
```

```
    Name: 'bw'
NominalValue: 5
    Mode: 'Percentage'
    Range: [4.5000 5.5000]
 PlusMinus: [-0.5000 0.5000]
  Percentage: [-10 10]
AutoSimplify: 'basic'
```

Note that the range of variation (`Range` property) and the additive deviation from nominal (the `PlusMinus` property) are consistent with the `Percentage` property value.

You can create state-space and transfer function models with uncertain real coefficients using `ureal` objects. The result is an *uncertain state-space object*, or `uss`. As an example, use the uncertain real parameter `bw` to model a first-order system whose bandwidth is between 4.5 and 5.5 rad/s.

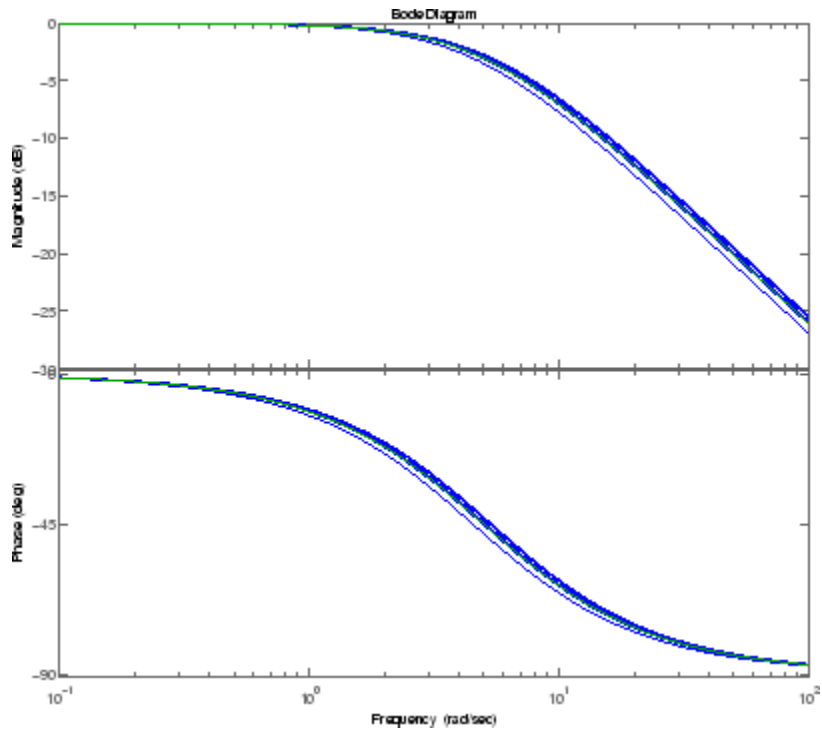
```
H = tf(1,[1/bw 1])
USS: 1 State, 1 Output, 1 Input, Continuous System
    bw: real, nominal = 5, variability = [-10 10]%, 1 occurrence
```

Note that the result `H` is an uncertain system, called a `uss` object. The nominal value of `H` is a state-space object. Verify that the pole is at -5 .

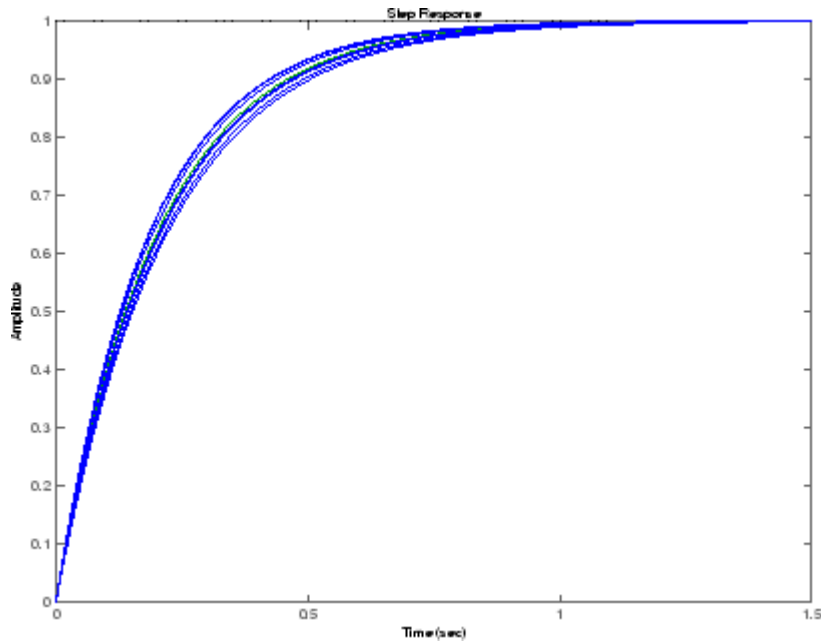
```
pole(H.NominalValue)
ans =
    -5
```

Next, use `bode` and `step` to examine the behavior of `H`.

```
bode(H,{1e-1 1e2});
```



step(H)



While there are variations in the bandwidth and time constant of H , the high-frequency rolls off at 20 dB/decade regardless of the value of bw . You can capture the more complicated uncertain behavior that typically occurs at high frequencies using the `ultidyn` uncertain element, which is described next.

Quantifying Unmodeled Dynamics

An informal way to describe the difference between the model of a process and the actual process behavior is in terms of bandwidth. It is common to hear “The model is good out to 8 radians/second.” The precise meaning is not clear, but it is reasonable to believe that for frequencies lower than, say, 5 rad/s, the model is accurate, and for frequencies beyond, say, 30 rad/s, the model is not necessarily representative of the process behavior. In the frequency range between 5 and 30, the guaranteed accuracy of the model degrades.

The uncertain linear, time-invariant dynamics object `ultidyn` can be used to model this type of knowledge. An `ultidyn` object represents an unknown linear system whose only known attribute is a uniform magnitude bound on its frequency response. When coupled with a nominal model and a

frequency-shaping filter, `ultidyn` objects can be used to capture uncertainty associated with the model dynamics.

Suppose that the behavior of the system modeled by H significantly deviates from its first-order behavior beyond 9 rad/s, for example, about 5% potential relative error at low frequency, increasing to 1000% at high frequency where H rolls off. In order to model frequency domain uncertainty as described above using `ultidyn` objects, follow these steps:

- 1** Create the nominal system G_{nom} , using `tf`, `ss`, or `zpk`. G_{nom} itself might already have parameter uncertainty. In this case G_{nom} is H , the first-order system with an uncertain time constant.
- 2** Create a filter W , called the “weight,” whose magnitude represents the relative uncertainty at each frequency. The utility `makeweight` is useful for creating first-order weights with specific low- and high-frequency gains, and specified gain crossover frequency.
- 3** Create an `ultidyn` object Δ with magnitude bound equal to 1.

The uncertain model G is formed by $G = G_{nom} * (1 + W * \Delta)$.

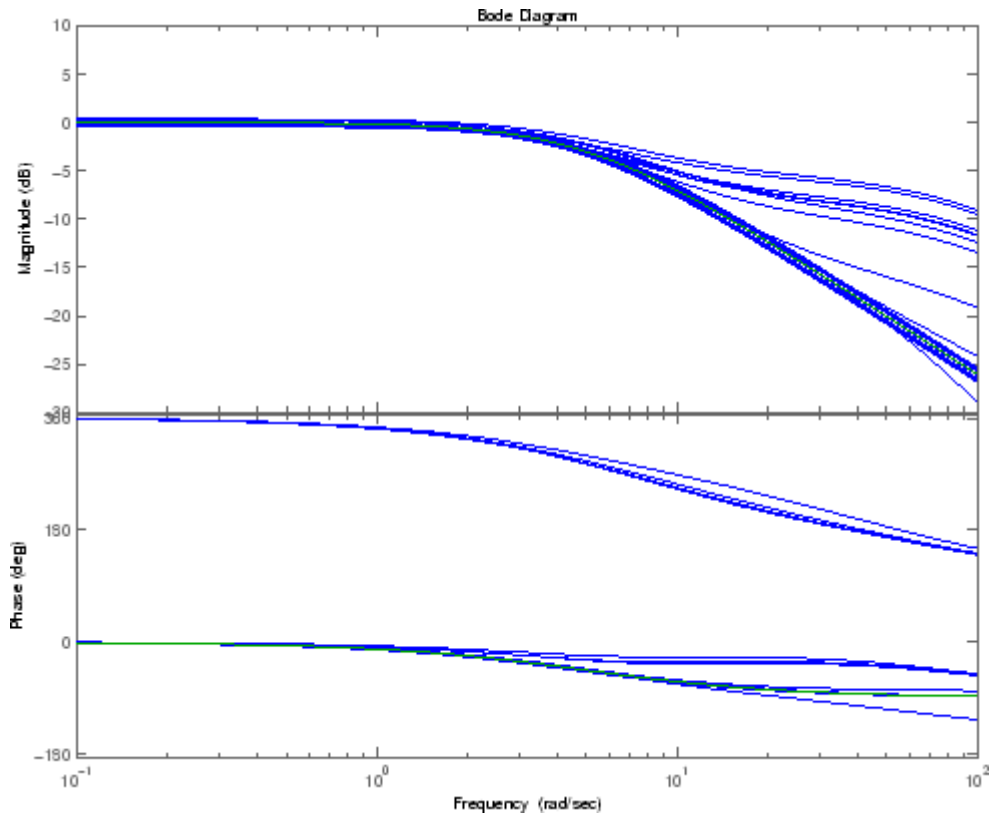
If the magnitude of W represents an absolute (rather than relative) uncertainty, use the formula $G = G_{nom} + W * \Delta$ instead.

The following commands carry out these steps:

```
Gnom = H;
W = makeweight(.05,9,10);
Delta = ultidyn('Delta',[1 1]);
G = Gnom*(1+W*Delta)
USS: 2 States, 1 Output, 1 Input, Continuous System
   Delta: 1x1 LTI, max. gain = 1, 1 occurrence
         bw: real, nominal = 5, variability = [-10 10]%, 1 occurrence
```

Note that the result G is also an uncertain system, with dependence on both Δ and bw . You can use `bode` to make a Bode plot of 20 random samples of G 's behavior over the frequency range [0.1 100] rad/s.

```
bode(G,{1e-1 1e2},25)
```



In the next section, you design and compare two feedback controllers for G .

Robust Controller Design

In this tutorial, design a feedback controller for G , the uncertain model created in “Create Models of Uncertain Systems” on page 4-2. The goals of this design are the usual ones: good steady-state tracking and disturbance rejection properties. Because the plant model is nominally a first-order lag, choose a PI control architecture. Given the desired closed-loop damping ratio ξ and natural frequency ω_n , the design equations for K_I and K_P (based on the nominal open-loop time constant of 0.2) are

$$K_I = \frac{\omega_n^2}{5}, K_P = \frac{2\xi\omega_n}{5} - 1.$$

Follow these steps to design the controller:

- 1 In order to study how the uncertain behavior of G affects the achievable closed-loop bandwidth, design two controllers, both achieving $\xi=0.707$, with different ω_n : 3 and 7.5 respectively.

```
xi = 0.707;
wn = 3;
K1 = tf([(2*xi*wn/5-1) wn*wn/5],[1 0]);
wn = 7.5;
K2 = tf([(2*xi*wn/5-1) wn*wn/5],[1 0]);
```

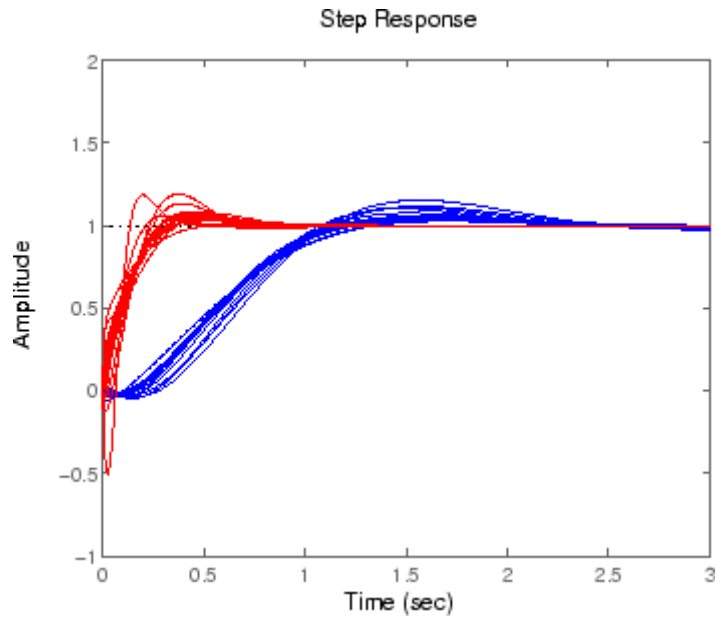
Note that the nominal closed-loop bandwidth achieved by K2 is in a region where G has significant model uncertainty. It will not be surprising if the model variations lead to significant degradations in the closed-loop performance.

- 2 Form the closed-loop systems using feedback.

```
T1 = feedback(G*K1,1);
T2 = feedback(G*K2,1);
```

- 3 Plot the step responses of 20 samples of each closed-loop system.

```
tfinal = 3;
step(T1,'b',T2,'r',tfinal,20)
```



The step responses for T2 exhibit a faster rise time because K2 sets a higher closed loop bandwidth. However, the model variations have a greater effect.

You can use `robuststab` to check the robustness of stability to the model variations.

```
[stabmarg1,destabu1,report1] = robuststab(T1);
stabmarg1
stabmarg1 =
    ubound: 4.0241
    lbound: 4.0241
    destabfreq: 3.4959
[stabmarg2,destabu2,report2] = robuststab(T2);
stabmarg2
stabmarg2 =
    ubound: 1.2545
    lbound: 1.2544
    destabfreq: 10.5249
```

The `stabmarg` variable gives lower and upper bounds on the stability margin. A stability margin greater than 1 means the system is stable for all values of the modeled uncertainty. A stability margin less than 1 means there are allowable values of the uncertain elements that make the system unstable. The `report` variable briefly summarizes the analysis.

```
report1
report1 =
Uncertain System is robustly stable to modeled uncertainty.
-- It can tolerate up to 402% of modeled uncertainty.
-- A destabilizing combination of 402% the modeled uncertainty exists, causing an instability at
3.5 rad/s.
report2
report2 =
Uncertain System is robustly stable to modeled uncertainty.
-- It can tolerate up to 125% of modeled uncertainty.
-- A destabilizing combination of 125% the modeled uncertainty exists, causing an instability at
10.5 rad/s.
```

While both systems are stable for all variations, their performance is clearly affected to different degrees. To determine how the uncertainty affects closed-loop performance, you can use `wcgain` to compute the *worst-case* effect of the uncertainty on the peak magnitude of the closed-loop sensitivity ($S=1/(1+GK)$) function. This peak gain is typically correlated with the amount of overshoot in a step response.

To do this, form the closed-loop sensitivity functions and call `wcgain`.

```
S1 = feedback(1,G*K1);
S2 = feedback(1,G*K2);
[maxgain1,wcu1] = wcgain(S1);
maxgain1
maxgain1 =
    lbound: 1.8684
    ubound: 1.9025
    critfreq: 3.5152
[maxgain2,wcu2] = wcgain(S2);
maxgain2
maxgain2 =
    lbound: 4.6031
```

```

ubound: 4.6671
critfreq: 11.0231

```

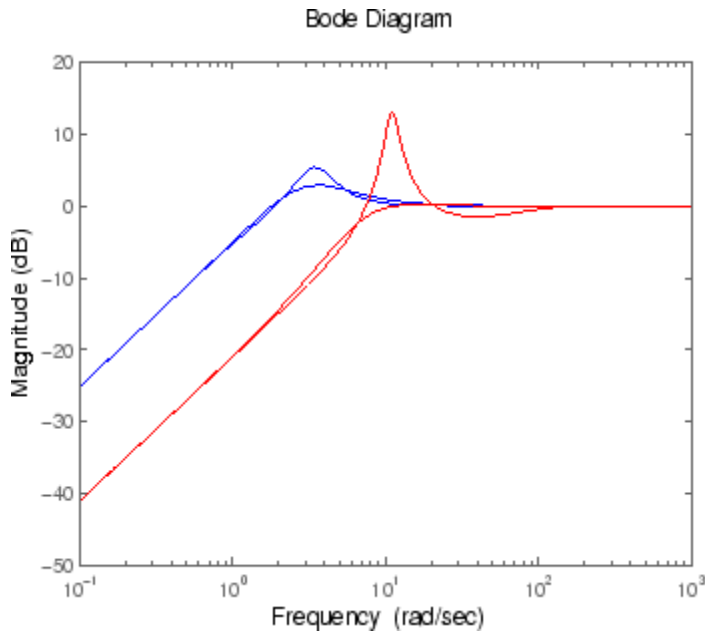
The maxgain variable gives lower and upper bounds on the worst-case peak gain of the sensitivity transfer function, as well as the specific frequency where the maximum gain occurs. The wcu variable contains specific values of the uncertain elements that achieve this worst-case behavior.

You can use usubs to substitute these worst-case values for uncertain elements, and compare the nominal and worst-case behavior. Use bodemag and step to make the comparison.

```

bodemag(S1.NominalValue, 'b', usubs(S1, wcu1), 'b');
hold on
bodemag(S2.NominalValue, 'r', usubs(S2, wcu2), 'r');
hold off

```



Clearly, while K2 achieves better nominal sensitivity than K1, the nominal closed-loop bandwidth extends too far into the frequency range where the

process uncertainty is very large. Hence the worst-case performance of K_2 is inferior to K_1 for this particular uncertain model.

The next section explores these robustness analysis tools further on a multiinput, multioutput system.

MIMO Robustness Analysis

The previous sections focused on simple uncertainty models of single-input and single-output systems, predominantly from a transfer function perspective. You can also create uncertain state-space models made up of uncertain state-space matrices. Moreover, all the analysis tools covered thus far can be applied to these systems as well.

Consider, for example, a two-input, two-output, two-state system whose model has parametric uncertainty in the state-space matrices. First create an uncertain parameter p . Using the parameter, make uncertain A and C matrices. The B matrix happens to be not-uncertain, although you will add frequency domain input uncertainty to the model in “Adding Independent Input Uncertainty to Each Channel” on page 4-15.

```
p = ureal('p',10,'Percentage',10);
A = [0 p;-p 0];
B = eye(2);
C = [1 p;-p 1];
H = ss(A,B,C,[0 0;0 0]);
```

You can view the properties of the uncertain system H using the `get` command.

```
get(H)
      a: [2x2 umat]
      b: [2x2 double]
      c: [2x2 umat]
      d: [2x2 double]
      e: []
      StateName: {2x1 cell}
      StateUnit: {2x1 cell}
      NominalValue: [2x2 ss]
      Uncertainty: [1x1 struct]
      InternalDelay: [0x1 double]
      InputDelay: [2x1 double]
      OutputDelay: [2x1 double]
      Ts: 0
      TimeUnit: 'seconds'
      InputName: {2x1 cell}
      InputUnit: {2x1 cell}
```



```

    InputGroup: [1x1 struct]
    OutputName: {2x1 cell}
    OutputUnit: {2x1 cell}
    OutputGroup: [1x1 struct]
        Name: ''
        Notes: {}
    UserData: []

```

The properties `a`, `b`, `c`, `d`, and `StateName` behave in exactly the same manner as `ss` objects. The properties `InputName`, `OutputName`, `InputGroup`, and `OutputGroup` behave in exactly the same manner as all the system objects (`ss`, `zpk`, `tf`, and `frd`). The `NominalValue` is an `ss` object.

Adding Independent Input Uncertainty to Each Channel

The model for `H` does not include actuator dynamics. Said differently, the actuator models are unity-gain for all frequencies.

Nevertheless, the behavior of the actuator for channel 1 is modestly uncertain (say 10%) at low frequencies, and the high-frequency behavior beyond 20 rad/s is not accurately modeled. Similar statements hold for the actuator in channel 2, with larger modest uncertainty at low frequency (say 20%) but accuracy out to 45 rad/s.

Use `ultidyn` objects `Delta1` and `Delta2` along with shaping filters `W1` and `W2` to add this form of frequency domain uncertainty to the model.

```

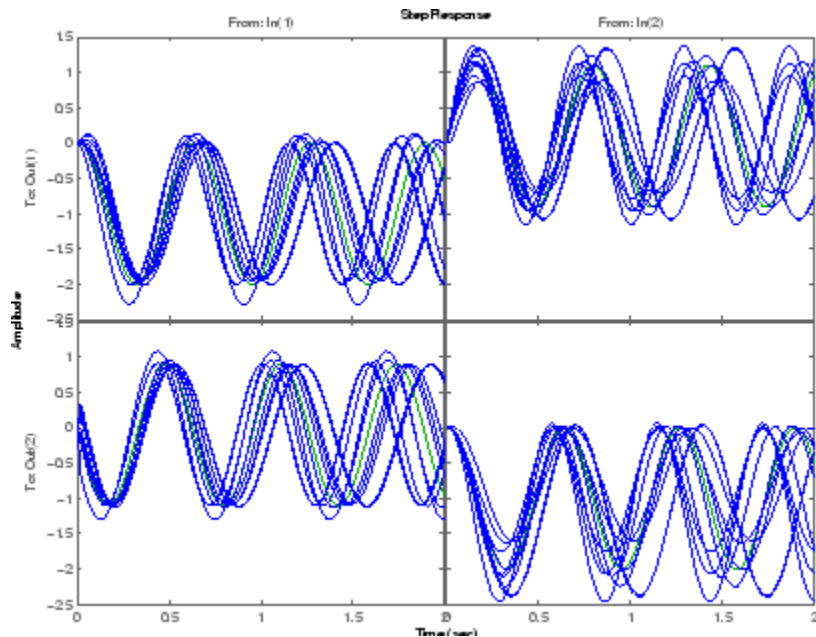
W1 = makeweight(.1,20,50);
W2 = makeweight(.2,45,50);
Delta1 = ultidyn('Delta1',[1 1]);
Delta2 = ultidyn('Delta2',[1 1]);
G = H*blkdiag(1+W1*Delta1,1+W2*Delta2)
USS: 4 States, 2 Outputs, 2 Inputs, Continuous System
    Delta1: 1x1 LTI, max. gain = 1, 1 occurrence
    Delta2: 1x1 LTI, max. gain = 1, 1 occurrence
           p: real, nominal = 10, variability = [-10 10]%, 2 occurrences

```

Note that G is a two-input, two-output uncertain system, with dependence on three uncertain elements, Δ_1 , Δ_2 , and p . It has four states, two from H and one each from the shaping filters W_1 and W_2 , which are embedded in G .

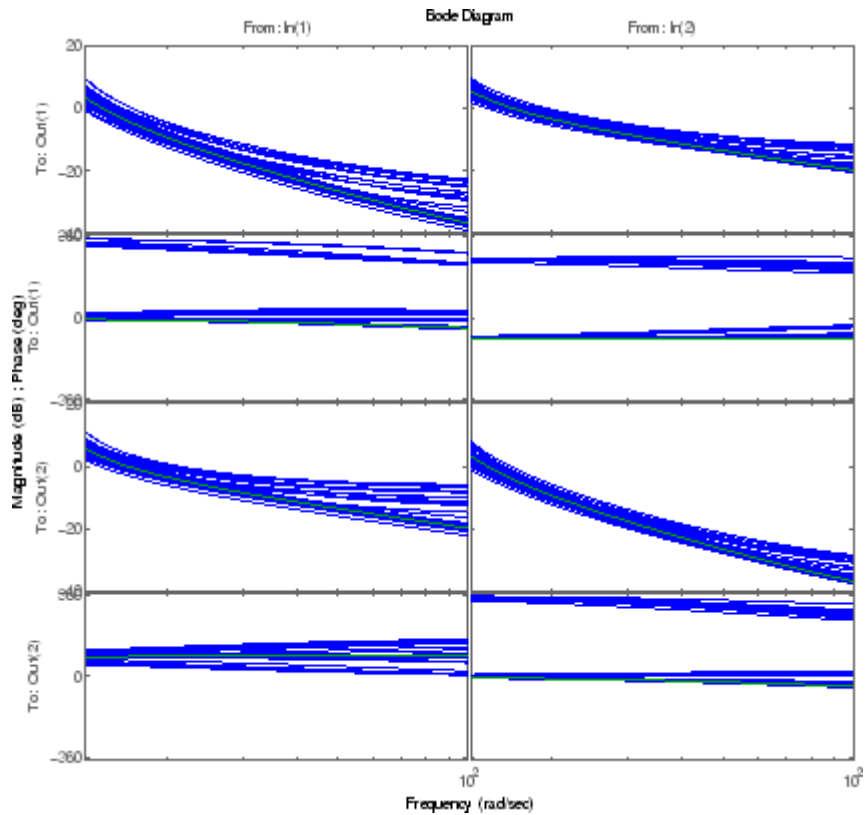
You can plot a 2-second step response of several samples of G . The 10% uncertainty in the natural frequency is obvious.

`step(G,2)`



You can create a Bode plot of 50 samples of G . The high-frequency uncertainty in the model is also obvious. For clarity, start the Bode plot beyond the resonance.

`bode(G, {13 100}, 50)`



Closed-Loop Robustness Analysis

You need to load the controller and verify that it is two-input and two-output.

```
load mimoKexample
size(K)
State-space model with 2 outputs, 2 inputs, and 9 states.
```

You can use the command `loopsens` to form all the standard plant/controller feedback configurations, including sensitivity and complementary sensitivity at both the input and output. Because G is uncertain, all the closed-loop systems are uncertain as well.

```
F = loopsens(G,K)
```

```
F =
  Poles: [13x1 double]
  Stable: 1
  Si: [2x2 uss]
  Ti: [2x2 uss]
  Li: [2x2 uss]
  So: [2x2 uss]
  To: [2x2 uss]
  Lo: [2x2 uss]
  PSi: [2x2 uss]
  CSo: [2x2 uss]
```

F is a structure with many fields. The poles of the nominal closed-loop system are in F.Poles, and F.Stable is 1 if the nominal closed-loop system is stable. In the remaining 10 fields, S stands for sensitivity, T for complementary sensitivity, and L for open-loop gain. The suffixes i and o refer to the input and output of the plant (G). Finally, P and C refer to the “plant” and “controller.”

Hence Ti is mathematically the same as

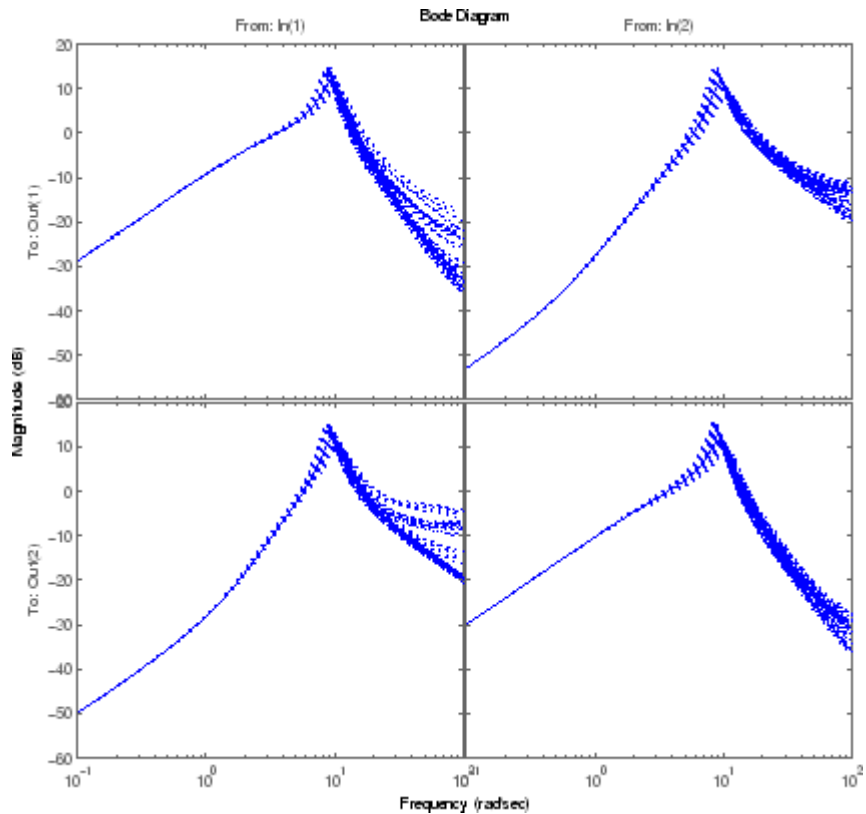
$$K(I + GK)^{-1}G$$

while Lo is $G*K$, and CSo is mathematically the same as

$$K(I + GK)^{-1}$$

You can examine the transmission of disturbances at the plant input to the plant output using `bodemag` on F.PSi. Graph 50 samples along with the nominal.

```
bodemag(F.PSi, ':-', {1e-1 100}, 50)
```



Nominal Stability Margins

You can use `loopmargin` to investigate loop-at-a-time gain and phase margins, loop-at-a-time disk margins, and simultaneous multivariable margins. They are computed for the nominal system and do not reflect the uncertainty models within G .

Explore the simultaneous margins individually at the plant input, individually at the plant output, and simultaneously at both input and output.

```
[I,DI,SimI,O,DO,SimO,Sim] = loopmargin(G,K);
```

The third output argument is the *simultaneous* gain and phase variations allowed in all input channels to the plant.

```
SimI
SimI =
    GainMargin: [0.1180 8.4769]
    PhaseMargin: [-76.5441 76.5441]
    Frequency: 6.2287
```

This information implies that the gain at the plant input can vary in both channels independently by factors between (approximately) 1/8 and 8, *as well as* phase variations up to 76 degrees.

The sixth output argument is the simultaneous gain and phase variations allowed in all output channels to the plant.

```
SimO
SimO =
    GainMargin: [0.1193 8.3836]
    PhaseMargin: [-76.3957 76.3957]
    Frequency: 18.3522
```

Note that the simultaneous margins at the plant output are similar to those at the input. This is not always the case in multiloop feedback systems.

The last output argument is the simultaneous gain and phase variations allowed in all input *and* output channels to the plant. As expected, when you consider all such variations simultaneously, the margins are somewhat smaller than those at the input or output alone.

```
Sim
Sim =
    GainMargin: [0.5671 1.7635]
    PhaseMargin: [-30.8882 30.8882]
    Frequency: 18.3522
```

Nevertheless, these numbers indicate a generally robust closed-loop system, able to tolerate significant gain (more than +/-50% in each channel) and 30 degree phase variations simultaneously in all input and output channels of the plant.

Robustness of Stability Model Uncertainty

With `loopmargin`, you determined various margins of the nominal, multiloop system. These margins are computed only for the nominal system, and do not reflect the uncertainty explicitly modeled by the `ureal` and `ultidyn` objects. When you work with detailed, complex uncertain system models, the conventional margins computed by `loopmargin` might not always be indicative of the actual stability margins associated with the uncertain elements. You can use `robuststab` to check the stability margin of the system to these specific modeled variations.

In this example, use `robuststab` to compute the stability margin of the closed-loop system represented by `Delta1`, `Delta2`, and `p`.

Use any of the closed-loop systems within `F = loopsens(G,K)`. All of them, `F.Si`, `F.To`, etc., have the same internal dynamics, and hence the stability properties are the same.

```
[stabmarg,desgtabu,report] = robuststab(F.So);
stabmarg
stabmarg =
    ubound: 2.2175
    lbound: 2.2175
    destabfreq: 13.7576
report
report =
Uncertain System is robustly stable to modeled uncertainty.
-- It can tolerate up to 222% of modeled uncertainty.
-- A destabilizing combination of 222% the modeled uncertainty exists,causing an instability at
13.8 rad/s.
```

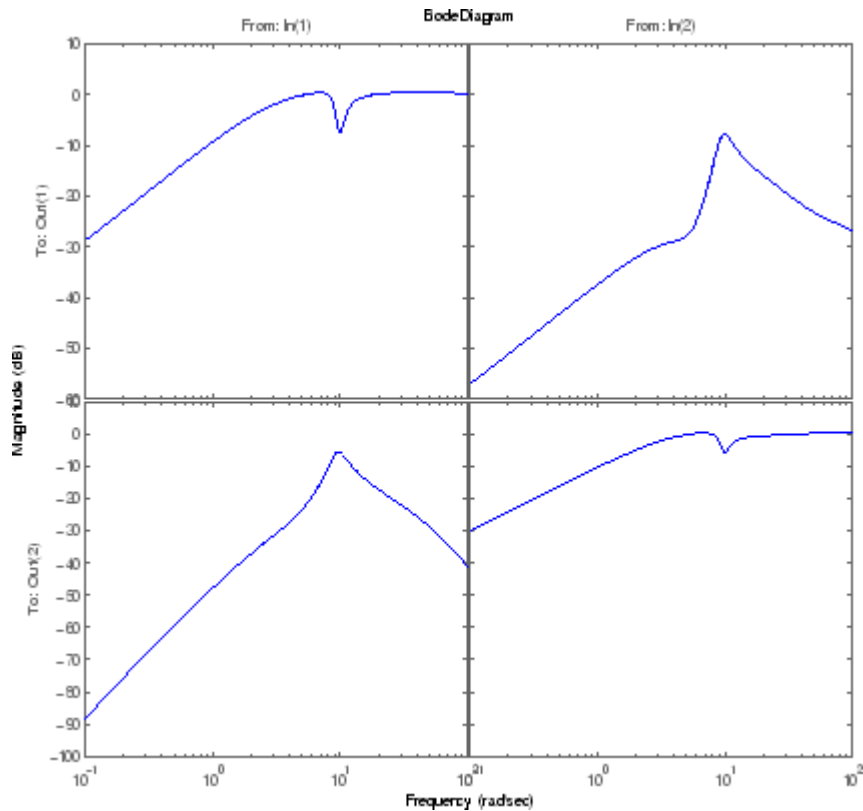
This analysis confirms what the `loopmargin` analysis suggested. The closed-loop system is quite robust, *in terms of stability*, to the variations modeled by the uncertain parameters `Delta1`, `Delta2`, and `p`. In fact, the system can tolerate more than twice the modeled uncertainty without losing closed-loop stability.

The next section studies the effects of these variations on the closed-loop output sensitivity function.

Worst-Case Gain Analysis

You can plot the Bode magnitude of the nominal output sensitivity function. It clearly shows decent disturbance rejection in all channels at low frequency.

```
bodemag(F.So.NominalValue, {1e-1 100})
```



You can compute the peak value of the maximum singular value of the frequency response matrix using `norm`.

```
[PeakNom, freq] = norm(F.So.NominalValue, 'inf')
PeakNom =
    1.1317
freq =
    7.0483
```


The peak is about 1.13, occurring at a frequency of 36 rad/s.

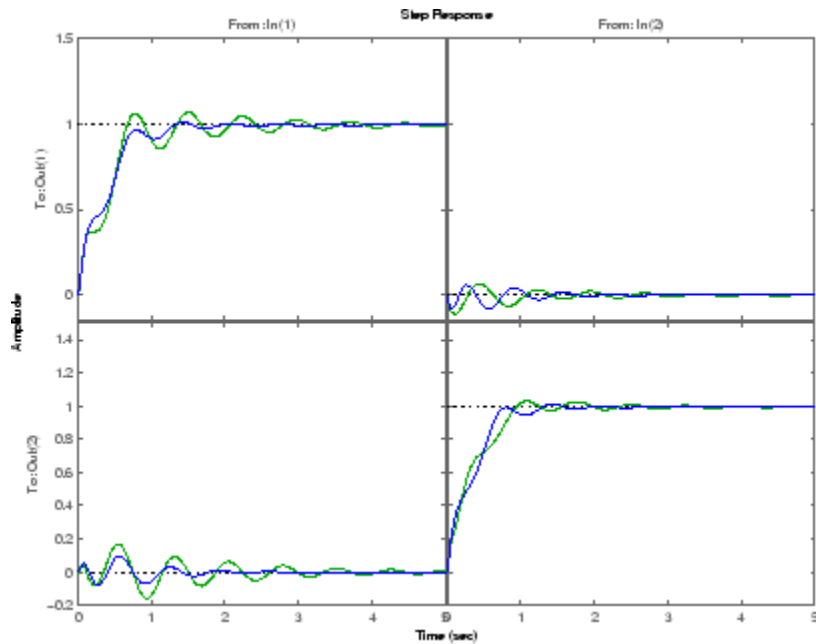
What is the maximum output sensitivity gain that is achieved when the uncertain elements Δ_1 , Δ_2 , and p vary over their ranges? You can use `wcgain` to answer this.

```
[maxgain,wcu] = wcgain(F.So);  
maxgain  
maxgain =  
    lbound: 2.1017  
    ubound: 2.1835  
    critfreq: 8.5546
```

The analysis indicates that the worst-case gain is somewhere between 2.1 and 2.2. The frequency where the peak is achieved is about 8.5.

You can replace the values for Δ_1 , Δ_2 , and p that achieve the gain of 2.1, using `usubs`. Make the substitution in the output complementary sensitivity, and do a step response.

```
step(F.To.NominalValue,usubs(F.To,wcu),5)
```



The perturbed response, which is the worst combination of uncertain values in terms of output sensitivity amplification, does not show significant degradation of the command response. The settling time is increased by about 50%, from 2 to 4, and the off-diagonal coupling is increased by about a factor of about 2, but is still quite small.

Summary of Robustness Analysis Tools

Function	Description
ureal	Create uncertain real parameter.
ultidyn	Create uncertain, linear, time-invariant dynamics.
uss	Create uncertain state-space object from uncertain state-space matrices.
ufrd	Create uncertain frequency response object.
loopsens	Compute all relevant open and closed-loop quantities for a MIMO feedback connection.
loopmargin	Compute loop-at-a-time as well as MIMO gain and phase margins for a multiloop system, including the simultaneous gain/phase margins.
robustperf	Robustness performance of uncertain systems.
robuststab	Compute the robust stability margin of a nominally stable uncertain system.
wcgain	Compute the worst-case gain of a nominally stable uncertain system.
wcmargin	Compute worst-case (over uncertainty) loop-at-a-time disk-based gain and phase margins.
wcsens	Compute worst-case (over uncertainty) sensitivity of plant-controller feedback loop.

H-Infinity and Mu Synthesis

- “Interpretation of H-Infinity Norm” on page 5-2
- “H-Infinity Performance” on page 5-9
- “Functions for Control Design” on page 5-17
- “H-Infinity and Mu Synthesis for Active Suspension Control” on page 5-19
- “Bibliography” on page 5-35

Interpretation of H-Infinity Norm

Norms of Signals and Systems

There are several ways of defining norms of a scalar signal $e(t)$ in the time domain. We will often use the 2-norm, (L_2 -norm), for mathematical convenience, which is defined as

$$\|e\|_2 := \left(\int_{-\infty}^{\infty} e(t)^2 dt \right)^{\frac{1}{2}}.$$

If this integral is finite, then the signal e is *square integrable*, denoted as $e \in L_2$. For vector-valued signals

$$e(t) = \begin{bmatrix} e_1(t) \\ e_2(t) \\ \vdots \\ e_n(t) \end{bmatrix},$$

the 2-norm is defined as

$$\begin{aligned} \|e\|_2 &:= \left(\int_{-\infty}^{\infty} \|e(t)\|_2^2 dt \right)^{\frac{1}{2}} \\ &= \left(\int_{-\infty}^{\infty} e^T(t) e(t) dt \right)^{\frac{1}{2}}. \end{aligned}$$

In μ -tools the dynamic systems we deal with are exclusively linear, with state-space model

$$\begin{bmatrix} \dot{x} \\ e \end{bmatrix} = \begin{bmatrix} A & B \\ C & D \end{bmatrix} \begin{bmatrix} x \\ d \end{bmatrix},$$

or, in the transfer function form,

$$e(s) = T(s)d(s), \quad T(s) := C(sI - A)^{-1}B + D$$

Two mathematically convenient measures of the transfer matrix $T(s)$ in the frequency domain are the matrix H_2 and H_∞ norms,

$$\|T\|_2 := \left[\frac{1}{2\pi} \int_{-\infty}^{\infty} \|T(j\omega)\|_F^2 d\omega \right]^{\frac{1}{2}}$$

$$\|T\|_\infty := \max_{\omega \in R} \bar{\sigma}[T(j\omega)],$$

where the Frobenius norm (see the MATLAB `norm` command) of a complex matrix M is

$$\|M\|_F := \sqrt{\text{Trace}(M^* M)}.$$

Both of these transfer function norms have input/output time-domain interpretations. If, starting from initial condition $x(0) = 0$, two signals d and e are related by

$$\begin{bmatrix} \dot{x} \\ e \end{bmatrix} = \begin{bmatrix} A & B \\ C & D \end{bmatrix} \begin{bmatrix} x \\ d \end{bmatrix},$$

then

- For d , a unit intensity, white noise process, the steady-state variance of e is $\|T\|_2$.
- The L_2 (or RMS) gain from $d \rightarrow e$,

$$\max_{d \neq 0} \frac{\|e\|_2}{\|d\|_2}$$

is equal to $\|T\|_\infty$. This is discussed in greater detail in the next section.

Using Weighted Norms to Characterize Performance

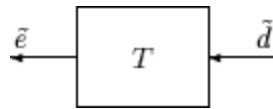
Any performance criterion must also account for

- Relative magnitude of outside influences
- Frequency dependence of signals
- Relative importance of the magnitudes of regulated variables

So, if the performance objective is in the form of a matrix norm, it should actually be a *weighted norm*

$$\|W_L T W_R\|$$

where the weighting function matrices W_L and W_R are frequency dependent, to account for bandwidth constraints and spectral content of exogenous signals. The most natural (mathematical) manner to characterize acceptable performance is in terms of the MIMO $\|\cdot\|_\infty (H_\infty)$ norm. For this reason, this section now discusses some interpretations of the H_∞ norm.

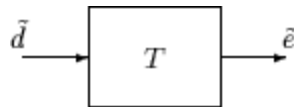


Unweighted MIMO System

Suppose T is a MIMO stable linear system, with transfer function matrix $T(s)$.

For a given driving signal $\tilde{d}(t)$, define \tilde{e} as the output, as shown below.

Note that it is more traditional to write the diagram in Unweighted MIMO System: Vectors from Left to Right on page 5-4 with the arrows going from left to right as in Weighted MIMO System on page 5-6.



Unweighted MIMO System: Vectors from Left to Right

The two diagrams shown above represent the exact same system. We prefer to write these block diagrams with the arrows going right to left to be consistent with matrix and operator composition.

Assume that the dimensions of T are $n_e \times n_d$. Let $\beta > 0$ be defined as

$$\beta := \|T\|_\infty := \max_{\omega \in R} \bar{\sigma}[T(j\omega)].$$

Now consider a response, starting from initial condition equal to 0. In that case, Parseval's theorem gives that

$$\frac{\|\tilde{e}\|_2}{\|\tilde{d}\|_2} = \frac{\left[\int_0^\infty \tilde{e}^T(t) \tilde{e}(t) dt \right]^{\frac{1}{2}}}{\left[\int_0^\infty \tilde{d}^T(t) \tilde{d}(t) dt \right]^{\frac{1}{2}}} \leq \beta.$$

Moreover, there are specific disturbances d that result in the ratio $\|\tilde{e}\|_2 / \|\tilde{d}\|_2$ arbitrarily close to β . Because of this, $\|T\|_\infty$ is referred to as the L_2 (or RMS) gain of the system.

As you would expect, a sinusoidal, steady-state interpretation of $\|T\|_\infty$ is also possible: For any frequency $\bar{\omega} \in R$, any vector of amplitudes $a \in R^{n_d}$, and any vector of phases $\phi \in R^{n_d}$, with $\|a\|_2 \leq 1$, define a time signal

$$\tilde{d}(t) = \begin{bmatrix} a_1 \sin(\bar{\omega}t + \phi_1) \\ \vdots \\ a_{n_d} \sin(\bar{\omega}t + \phi_{n_d}) \end{bmatrix}.$$

Applying this input to the system T results in a steady-state response \tilde{e}_{ss} of the form

$$\tilde{e}_{ss}(t) = \begin{bmatrix} b_1 \sin(\bar{\omega}t + \phi_1) \\ \vdots \\ b_{n_e} \sin(\bar{\omega}t + \phi_{n_e}) \end{bmatrix}.$$

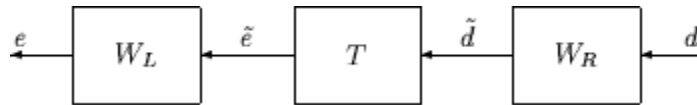
The vector $b \in \mathbb{R}^{n_e}$ will satisfy $\|b\|_2 \leq \beta$. Moreover, β , as defined in Weighted MIMO System on page 5-6, is the smallest number such that this is true for every $\|a\|_2 \leq 1$, $\bar{\omega}$, and φ .

Note that in this interpretation, the vectors of the sinusoidal magnitude responses are unweighted, and measured in Euclidean norm. If realistic multivariable performance objectives are to be represented by a single MIMO $\|\cdot\|_\infty$ objective on a closed-loop transfer function, additional scalings are necessary. Because many different objectives are being lumped into one matrix and the associated cost is the norm of the matrix, it is important to use frequency-dependent weighting functions, so that different requirements can be meaningfully combined into a single cost function. Diagonal weights are most easily interpreted.

Consider the diagram of Weighted MIMO System on page 5-6, along with Unweighted MIMO System: Vectors from Left to Right on page 5-4.

Assume that W_L and W_R are diagonal, stable transfer function matrices, with diagonal entries denoted L_i and R_i .

$$W_L = \begin{bmatrix} L_1 & 0 & \dots & 0 \\ 0 & L_2 & \dots & 0 \\ \vdots & \vdots & \ddots & \vdots \\ 0 & 0 & \dots & L_{n_e} \end{bmatrix}, \quad W_R = \begin{bmatrix} R_1 & 0 & \dots & 0 \\ 0 & R_2 & \dots & 0 \\ \vdots & \vdots & \ddots & \vdots \\ 0 & 0 & \dots & R_{n_d} \end{bmatrix}.$$



$$e = W_L \tilde{e} = W_L T \tilde{d} = W_L T W_R d$$

Weighted MIMO System

Bounds on the quantity $\|W_L T W_R\|_\infty$ will imply bounds about the sinusoidal

steady-state behavior of the signals \tilde{d} and $\tilde{e} (= T\tilde{d})$ in the diagram of Unweighted MIMO System: Vectors from Left to Right on page 5-4.

Specifically, for sinusoidal signal \tilde{d} , the steady-state relationship between

$\tilde{e}(=T\tilde{d})$, \tilde{d} and $\|W_LTW_R\|_\infty$ is as follows. The steady-state solution \tilde{e}_{ss} , denoted as

$$\tilde{e}_{ss}(t) = \begin{bmatrix} \tilde{e}_1 \sin(\bar{\omega}t + \phi_1) \\ \vdots \\ \tilde{e}_{n_e} \sin(\bar{\omega}t + \phi_{n_d}) \end{bmatrix} \quad (5-1)$$

satisfies

$$\sum_{i=1}^{n_e} |W_{L_i}(j\bar{\omega})\tilde{e}_i|^2 \leq 1$$

for all sinusoidal input signals \tilde{d} of the form

$$\tilde{d}(t) = \begin{bmatrix} \tilde{d}_1 \sin(\bar{\omega}t + \phi_1) \\ \vdots \\ \tilde{d}_{n_e} \sin(\bar{\omega}t + \phi_{n_d}) \end{bmatrix} \quad (5-2)$$

satisfying

$$\sum_{i=1}^{n_d} \frac{|\tilde{d}_i|^2}{|W_{R_i}(j\bar{\omega})|^2} \leq 1$$

if and only if $\|W_LTW_R\|_\infty \leq 1$.

This approximately (*very* approximately — the next statement is not actually correct) implies that $\|W_LTW_R\|_\infty \leq 1$ if and only if for every fixed frequency $\bar{\omega}$, and all sinusoidal disturbances \tilde{d} of the form Equation 5-2 satisfying

$$|\tilde{d}_i| \leq |W_{R_i}(j\bar{\omega})|$$

the steady-state error components will satisfy

$$|\tilde{e}_i| \leq \frac{1}{|W_{L_i}(j\bar{\omega})|}.$$

This shows how one could pick performance weights to reflect the desired frequency-dependent performance objective. Use W_R to represent the relative magnitude of sinusoids disturbances that might be present, and use $1/W_L$ to represent the desired upper bound on the subsequent errors that are produced.

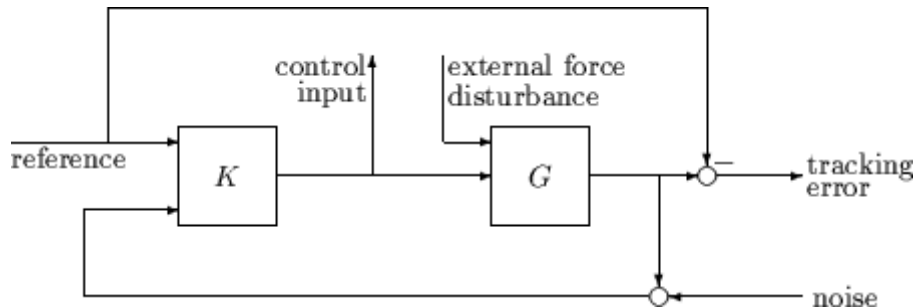
Remember, however, that the weighted H_∞ norm does *not* actually give element-by-element bounds on the components of \tilde{e} based on element-by-element bounds on the components of \tilde{d} . The precise bound it gives is in terms of Euclidean norms of the components of \tilde{e} and \tilde{d} (weighted appropriately by $W_L(j\bar{\omega})$ and $W_R(j\bar{\omega})$).

H-Infinity Performance

Performance as Generalized Disturbance Rejection

The modern approach to characterizing closed-loop performance objectives is to measure the size of certain closed-loop transfer function matrices using various matrix norms. Matrix norms provide a measure of how large output signals can get for certain classes of input signals. Optimizing these types of performance objectives over the set of stabilizing controllers is the main thrust of recent optimal control theory, such as L_1 , H_2 , H_∞ , and optimal control. Hence, it is important to understand how many types of control objectives can be posed as a minimization of closed-loop transfer functions.

Consider a tracking problem, with disturbance rejection, measurement noise, and control input signal limitations, as shown in Generalized and Weighted Performance Block Diagram on page 5-11. K is some controller to be designed and G is the system you want to control.



Typical Closed-Loop Performance Objective

A reasonable, though not precise, design objective would be to design K to keep tracking errors and control input signal small for all reasonable reference commands, sensor noises, and external force disturbances.

Hence, a natural performance objective is the closed-loop gain from exogenous influences (reference commands, sensor noise, and external force disturbances) to regulated variables (tracking errors and control input signal). Specifically, let T denote the closed-loop mapping from the outside influences to the regulated variables:

$$\underbrace{\begin{bmatrix} \text{tracking error} \\ \text{control input} \end{bmatrix}}_{\text{regulated variables}} = T \underbrace{\begin{bmatrix} \text{reference} \\ \text{external force} \\ \text{noise} \end{bmatrix}}_{\text{outside influences}}$$

You can assess performance by measuring the gain from *outside influences* to *regulated variables*. In other words, good performance is associated with T being small. Because the closed-loop system is a multiinput, multioutput (MIMO) dynamic system, there are two different aspects to the gain of T :

- Spatial (*vector* disturbances and *vector* errors)
- Temporal (dynamic relationship between input/output signals)

Hence the performance criterion must account for

- Relative magnitude of outside influences
- Frequency dependence of signals
- Relative importance of the magnitudes of regulated variables

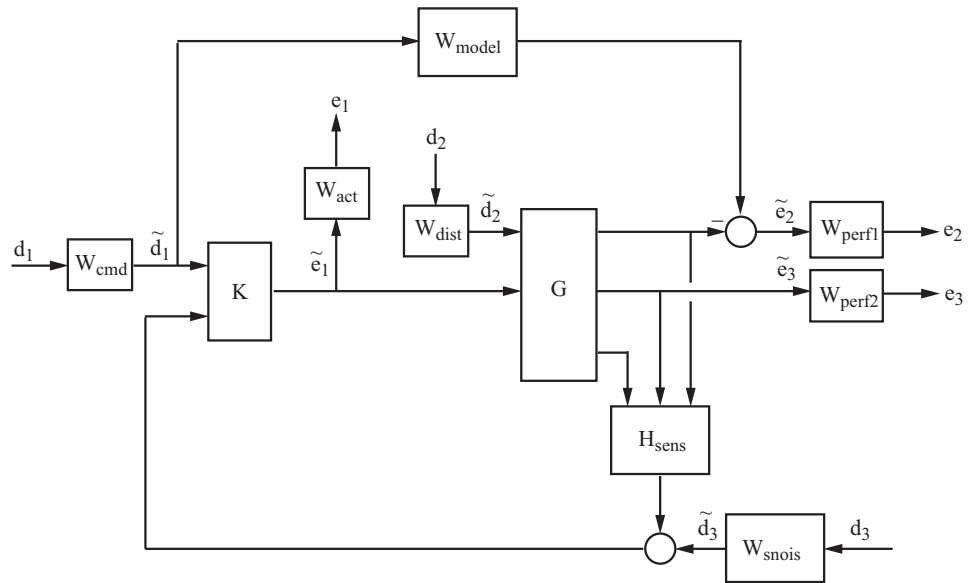
So if the performance objective is in the form of a matrix norm, it should actually be a *weighted norm*

$$\|W_L T W_R\|$$

where the weighting function matrices W_L and W_R are frequency dependent, to account for bandwidth constraints and spectral content of exogenous signals. A natural (mathematical) manner to characterize acceptable performance is in terms of the MIMO $\|\cdot\|_\infty$ (H_∞) norm. See “Interpretation of H-Infinity Norm” on page 5-2 for an interpretation of the H_∞ norm and signals.

Interconnection with Typical MIMO Performance Objectives

The closed-loop performance objectives are formulated as weighted closed-loop transfer functions that are to be made small through feedback. A generic example, which includes many relevant terms, is shown in block diagram form in Generalized and Weighted Performance Block Diagram on page 5-11. In the diagram, G denotes the plant model and K is the feedback controller.



Generalized and Weighted Performance Block Diagram

The blocks in this figure might be scalar (SISO) and/or multivariable (MIMO), depending on the specific example. The mathematical objective of H_∞ control is to make the closed-loop MIMO transfer function T_{ed} satisfy $\|T_{ed}\|_\infty < 1$. The weighting functions are used to scale the input/output transfer functions such that when $\|T_{ed}\|_\infty < 1$, the relationship between \tilde{d} and \tilde{e} is suitable.

Performance requirements on the closed-loop system are transformed into the H_∞ framework with the help of *weighting* or *scaling* functions. Weights are selected to account for the relative magnitude of signals, their frequency dependence, and their relative importance. This is captured in the figure above, where the weights or scalings $[W_{cmd}, W_{dist}, W_{snois}]$ are used to transform and scale the normalized input signals $[d_1, d_2, d_3]$ into physical units defined as $[d_1, d_2, d_3]$. Similarly weights or scalings $[W_{act}, W_{perf1}, W_{perf2}]$ transform and scale physical units into normalized output signals $[e_1, e_2, e_3]$. An interpretation of the signals, weighting functions, and models follows.

Signal	Meaning
d_1	Normalized reference command
\tilde{d}_1	Typical reference command in physical units
d_2	Normalized exogenous disturbances
\tilde{d}_2	Typical exogenous disturbances in physical units
d_3	Normalized sensor noise
\tilde{d}_3	Typical sensor noise in physical units
e_1	Weighted control signals
\tilde{e}_1	Actual control signals in physical units
e_2	Weighted tracking errors
\tilde{e}_2	Actual tracking errors in physical units
e_3	Weighted plant errors
\tilde{e}_3	Actual plant errors in physical units

W_{cmd}

W_{cmd} is included in H_∞ control problems that require tracking of a reference command. W_{cmd} shapes the normalized reference command signals (magnitude and frequency) into the actual (or typical) reference signals that you expect to occur. It describes the magnitude and the frequency dependence of the reference commands generated by the normalized reference signal. Normally W_{cmd} is flat at low frequency and rolls off at high frequency. For example, in a flight control problem, fighter pilots generate stick input reference commands up to a bandwidth of about 2 Hz. Suppose that the stick has a maximum travel of three inches. Pilot commands could be modeled as normalized signals passed through a first-order filter:

$$W_{cmd} = \frac{3}{\frac{1}{2 \cdot 2\pi} s + 1}.$$

W_{model}

W_{model} represents a desired ideal model for the closed-loop system and is often included in problem formulations with tracking requirements. Inclusion of an ideal model for tracking is often called a *model matching* problem, i.e., the objective of the closed-loop system is to match the defined model. For good command tracking response, you might want the closed-loop system to respond like a well-damped second-order system. The ideal model would then be

$$W_{model} = \frac{\omega^2}{s^2 + 2\zeta\omega + \omega^2}$$

for specific desired natural frequency ω and desired damping ratio ζ . Unit conversions might be necessary to ensure exact correlation between the ideal model and the closed-loop system. In the fighter pilot example, suppose that roll-rate is being commanded and 10°/second response is desired for each inch of stick motion. Then, in these units, the appropriate model is:

$$W_{model} = 10 \frac{\omega^2}{s^2 + 2\zeta\omega + \omega^2}.$$

 W_{dist}

W_{dist} shapes the frequency content and magnitude of the exogenous disturbances affecting the plant. For example, consider an electron microscope as the plant. The dominant performance objective is to mechanically isolate the microscope from outside mechanical disturbances, such as ground excitations, sound (pressure) waves, and air currents. You can capture the spectrum and relative magnitudes of these disturbances with the transfer function weighting matrix W_{dist} .

 W_{perf1}

W_{perf1} weights the difference between the response of the closed-loop system and the ideal model W_{model} . Often you might want accurate matching of the ideal model at low frequency and require less accurate matching at higher frequency, in which case W_{perf1} is flat at low frequency, rolls off at first or

second order, and flattens out at a small, nonzero value at high frequency. The inverse of the weight is related to the allowable size of tracking errors, when dealing with the reference commands and disturbances described by W_{cmd} and W_{dist} .

W_{perf2}

W_{perf2} penalizes variables internal to the process G , such as actuator states that are internal to G or other variables that are not part of the tracking objective.

W_{act}

W_{act} is used to shape the penalty on control signal use. W_{act} is a frequency varying weighting function used to penalize limits on the deflection/position, deflection rate/velocity, etc., response of the control signals, when dealing with the tracking and disturbance rejection objectives defined above. Each control signal is usually penalized independently.

W_{snois}

W_{snois} represents frequency domain models of sensor noise. Each sensor measurement feedback to the controller has some noise, which is often higher in one frequency range than another. The W_{snois} weight tries to capture this information, derived from laboratory experiments or based on manufacturer measurements, in the control problem. For example, medium grade accelerometers have substantial noise at low frequency and high frequency. Therefore the corresponding W_{snois} weight would be larger at low and high frequency and have a smaller magnitude in the mid-frequency range. Displacement or rotation measurement is often quite accurate at low frequency and in steady state, but responds poorly as frequency increases. The weighting function for this sensor would be small at low frequency, gradually increase in magnitude as a first- or second-order system, and level out at high frequency.

H_{sens}

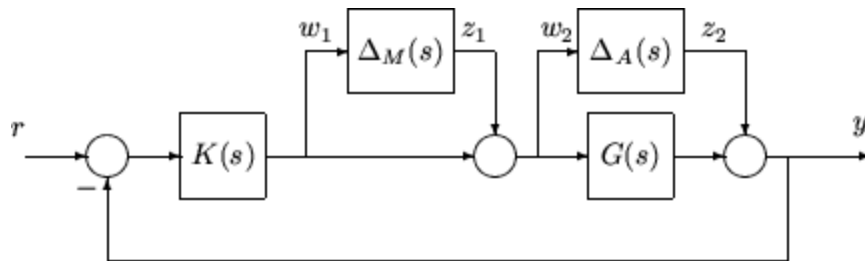
H_{sens} represents a model of the sensor dynamics or an external antialiasing filter. The transfer functions used to describe H_{sens} are based on physical

characteristics of the individual components. These models might also be lumped into the plant model G .

This generic block diagram has tremendous flexibility and many control performance objectives can be formulated in the H_∞ framework using this block diagram description.

Robustness in the H-Infinity Framework

Performance and robustness tradeoffs in control design were discussed in the context of multivariable loop shaping in “Tradeoff Between Performance and Robustness” on page 2-2. In the H_∞ control design framework, you can include robustness objectives as additional disturbance to error transfer functions — disturbances to be kept small. Consider the following figure of a closed-loop feedback system with additive and multiplicative uncertainty models.

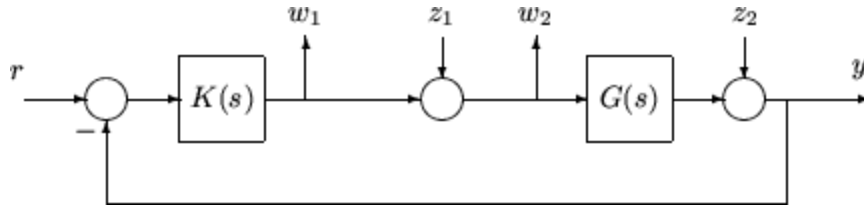


The transfer function matrices are defined as:

$$TF(s)_{z_1 \rightarrow w_1} = T_I(s) = KG(I + GK)^{-1}$$

$$TF(s)_{z_2 \rightarrow w_2} = KS_O(s) = K(I + GK)^{-1}$$

where $T_I(s)$ denotes the input complementary sensitivity function and $S_O(s)$ denotes the output sensitivity function. Bounds on the size of the transfer function matrices from z_1 to w_1 and z_2 to w_2 ensure that the closed-loop system is robust to multiplicative uncertainty, $\Delta_M(s)$, at the plant input, and additive uncertainty, $\Delta_A(s)$, around the plant $G(s)$. In the H_∞ control problem formulation, the robustness objectives enter the synthesis procedure as additional input/output signals to be kept small. The interconnection with the uncertainty blocks removed follows.



The H_∞ control robustness objective is now in the same format as the performance objectives, that is, to minimize the H_∞ norm of the transfer matrix from z , $[z_1, z_2]$, to w , $[w_1, w_2]$.

Weighting or scaling matrices are often introduced to shape the frequency and magnitude content of the sensitivity and complementary sensitivity transfer function matrices. Let W_M correspond to the multiplicative uncertainty and W_A correspond to the additive uncertainty model. $\Delta_M(s)$ and $\Delta_A(s)$ are assumed to be a norm bounded by 1, i.e., $|\Delta_M(s)| < 1$ and $|\Delta_A(s)| < 1$. Hence as a function of frequency, $|W_M(j\omega)|$ and $|W_A(j\omega)|$ are the respective sizes of the largest anticipated additive and multiplicative plant perturbations.

The multiplicative weighting or scaling W_M represents a percentage error in the model and is often small in magnitude at low frequency, between 0.05 and 0.20 (5% to 20% modeling error), and growing larger in magnitude at high frequency, 2 to 5 ((200% to 500% modeling error). The weight will transition by crossing a magnitude value of 1, which corresponds to 100% uncertainty in the model, at a frequency at least twice the bandwidth of the closed-loop system. A typical multiplicative weight is

$$W_M = 0.10 \frac{\frac{1}{5}s + 1}{\frac{1}{200}s + 1}.$$

By contrast, the additive weight or scaling W_A represents an absolute error that is often small at low frequency and large in magnitude at high frequency. The magnitude of this weight depends directly on the magnitude of the plant model, $G(s)$.

Functions for Control Design

The term control system design refers to the process of synthesizing a feedback control law that meets design specifications in a closed-loop control system. The design methods are iterative, combining parameter selection with analysis, simulation, and insight into the dynamics of the plant. Robust Control Toolbox software provides a set of commands that you can use for a broad range of multivariable control applications, including

- H_2 control design
- H_∞ standard and loop-shaping control design
- H_∞ tuning of controllers with fixed structure
- H_∞ normalized coprime factor control design
- Mixed H_2/H_∞ control design
- μ -synthesis via $D-K$ and $D-G-K$ iteration
- Sampled-data H_∞ control design

These functions cover both continuous and discrete-time problems. The following table summarizes the H_2 and H_∞ control design commands.

Function	Description
augw	Augments plant weights for mixed-sensitivity control design
h2hinfsyn	Mixed H_2/H_∞ controller synthesis
h2syn	H_2 controller synthesis
hinfsyn	H_∞ controller synthesis
hinfstruct	H_∞ tuning of fixed control structures
loopsyn	H_∞ loop-shaping controller synthesis
ltrsyn	Loop-transfer recovery controller synthesis
mixsyn	H_∞ mixed-sensitivity controller synthesis
ncfsyn	H_∞ normalized coprime factor controller synthesis
sdhinfsyn	Sample-data H_∞ controller synthesis

The following table summarizes μ -synthesis control design commands.

Function	Description
<code>dksyn</code>	Synthesis of a robust controller via μ -synthesis
<code>dkitopt</code>	Create a <code>dksyn</code> options object
<code>drawmag</code>	Interactive mouse-based sketching and fitting tool
<code>fitfrd</code>	Fit scaling frequency response data with LTI model
<code>fitmagfrd</code>	Fit scaling magnitude data with stable, minimum-phase model

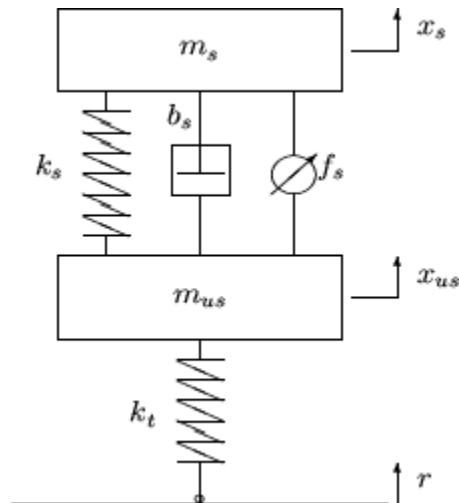
H-Infinity and Mu Synthesis for Active Suspension Control

Conventional passive suspensions employ a spring and damper between the car body and wheel assembly, representing a tradeoff between conflicting performance metrics such as passenger comfort, road holding, and suspension deflection. Active suspensions allow the designer to balance these objectives using a hydraulic actuator, controlled by feedback, between the chassis and wheel assembly.

In this section, you design an active suspension system for a quarter car body and wheel assembly model using the H_∞ control design technique. You will see the tradeoff between passenger comfort, i.e., minimizing car body travel, versus suspension travel as the performance objective.

Quarter Car Suspension Model

The quarter car model shown is used to design active suspension control laws.



The sprung mass m_s represents the car chassis, while the unsprung mass m_{us} represents the wheel assembly. The spring, k_s , and damper, b_s , represent a passive spring and shock absorber that are placed between the car body and the wheel assembly, while the spring k_t serves to model the compressibility of the pneumatic tire. The variables x_s , x_{us} , and r are the car body travel, the

wheel travel, and the road disturbance, respectively. The force f_s , kN, applied between the sprung and unsprung masses, is controlled by feedback and represents the active component of the suspension system. The dynamics of the actuator are ignored in this example, and assume that the control signal is the force f_s . Defining $x_1 := x_s$, $x_2 := \dot{x}_s$, $x_3 := x_{us}$, and $x_4 := \dot{x}_{us}$, the following is the state-space description of the quarter car dynamics.

$$\begin{aligned}\dot{x}_1 &= x_2 \\ \dot{x}_2 &= -\frac{1}{m_s} [k_s(x_1 - x_3) + b_s(x_2 - x_4) - f_s] \\ \dot{x}_3 &= x_4 \\ \dot{x}_4 &= \frac{1}{m_{us}} [k_s(x_1 - x_3) + b_s(x_2 - x_4) - k_t(x_3 - r) - f_s].\end{aligned}$$

The following component values are taken from reference [9].

```
ms = 290;      % kg
mus = 59;     % kg
bs = 1000;    % N/m/s
ks = 16182 ;  % N/m
kt = 190000;  % N/m
```

A linear, time-invariant model of the quarter car model, `qcar`, is constructed from the equations of motion and parameter values. The inputs to the model are the road disturbance and actuator force, respectively, and the outputs are the car body deflection, acceleration, and suspension deflection.

```
A12 = [ 0 1 0 0; [-ks -bs ks bs]/ms ];
A34 = [ 0 0 0 1; [ks bs -ks-kt -bs]/mus];
B12 = [0 0; 0 10000/ms];
B34 = [0 0; [kt -10000]/mus];
C   = [1 0 0 0; A12(2,:); 1 0 -1 0; 0 0 0 0];
D   = [0 0; B12(2,:); 0 0; 0 1];
qcar = ss([A12; A34],[B12; B34],C,D);
```

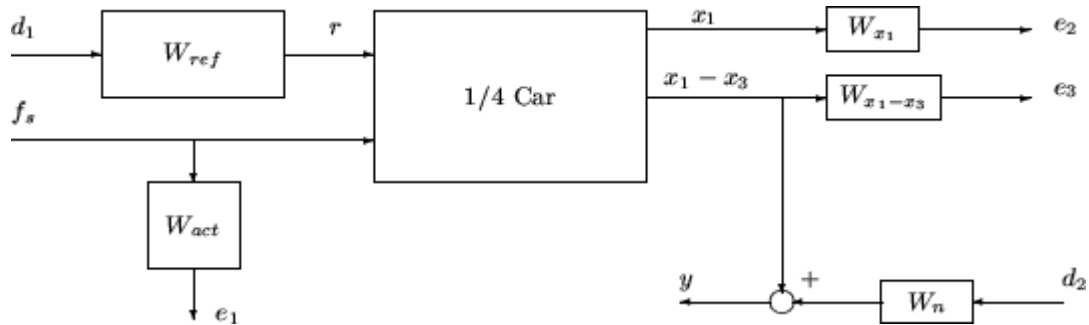
It is well known [8] that the acceleration transfer function has an invariant point at the *tirehop frequency*, 56.7 rad/s. Similarly, the suspension deflection transfer function has an invariant point at the *rattlespace frequency*, 23.3

rad/s. The tradeoff between passenger comfort and suspension deflection is because it is not possible to simultaneously keep both transfer functions small around the tirehop frequency and in the low frequency range.

Linear H-Infinity Controller Design

The design of linear suspension controllers that emphasize either passenger comfort or suspension deflection. The controllers in this section are designed using linear H_∞ synthesis [5]. As is standard in the H_∞ framework, the performance objectives are achieved via minimizing weighted transfer function norms.

Weighting functions serve two purposes in the H_∞ framework: They allow the direct comparison of different performance objectives with the same norm, and they allow for frequency information to be incorporated into the analysis. For more details on H_∞ control design, refer to [4], [6], [7], [11], and [13]. A block diagram of the H_∞ control design interconnection for the active suspension problem is shown below.



The measured output or feedback signal y is the suspension deflection $x_1 - x_3$. The controller acts on this signal to produce the control input, the hydraulic actuator force f_s . The block W_n serves to model sensor noise in the measurement channel. W_n is set to a sensor noise value of 0.01 m.

$$W_n = 0.01;$$

In a more realistic design, W_n would be frequency dependent and would serve to model the noise associated with the displacement sensor. The weight W_{ref}

is used to scale the magnitude of the road disturbances. Assume that the maximum road disturbance is 7 cm and hence choose $W_{ref} = 0.07$.

$W_{ref} = 0.07$;

The magnitude and frequency content of the control force f_s are limited by the weighting function W_{act} . Choose

$$W_{act} = \frac{100}{13} \frac{s+50}{s+500}.$$

The magnitude of the weight increases above 50 rad/s in order to limit the closed-loop bandwidth.

$W_{act} = (100/13)*tf([1 50],[1 500]);$

H-Infinity Control Design 1

The purpose of the weighting functions W_{x_1} and $W_{x_1-x_3}$ is to keep the car deflection and the suspension deflection small over the desired frequency ranges. In the first design, you are designing the controller for passenger comfort, and hence the car body deflection x_1 is penalized.

$W_{x1} = 8*tf(2*pi*5,[1 2*pi*5]);$

The weight magnitude rolls off above $5 \times 2\pi$ rad/s to respect a well-known H_∞ design rule of thumb that requires the performance weights to roll off before an open-loop zero (56.7 rad/s in this case). The suspension deflection weight

$W_{x_1-x_3}$ is not included in this control problem formulation.

You can construct the weighted H_∞ plant model for control design, denoted `qcaric1`, using the `sysic` command. The control signal corresponds to the last input to `qcaric1`, f_s . The car body acceleration, which is noisy, is the measured signal and corresponds to the last output of `qcaric1`.

```
systemnames = 'qcar Wn Wref Wact Wx1';
inputvar = '[ d1; d2; fs ]';
outputvar = '[ Wact; Wx1; qcar(3)+Wn ]';
input_to_qcar = '[ Wref; fs]';
input_to_Wn = '[ d2 ]';
```

```
input_to_Wref = '[ d1 ]';
input_to_Wact = '[ fs ]';
input_to_Wx1 = '[ qcar(1) ]';
qcaric1 = sysic;
```

An H_∞ controller is synthesized with the `hinfsyn` command. There is one control input, the hydraulic actuator force, and one measurement signal, the car body acceleration.

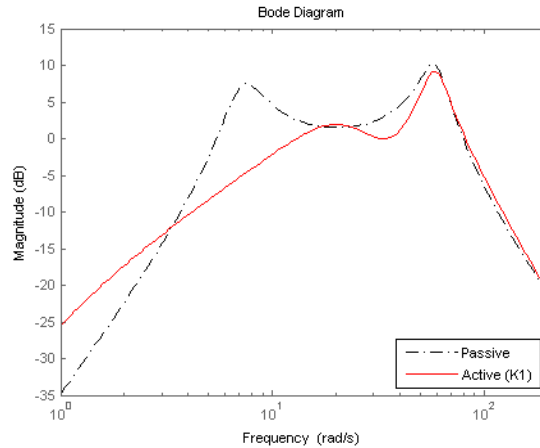
```
ncont = 1;
nmeas = 1;
[K1,Scl1,gam1] = hinfsyn(qcaric1,nmeas,ncont);
CL1 = lft(qcar([1:4 3],1:2),K1);
sprintf('H-infinity controller K1 achieved a norm of %2.5g',gam1)
```

```
ans =
```

```
H-infinity controller K1 achieved a norm of 0.60887
```

You can analyze the H_∞ controller by constructing the closed-loop feedback system `CL1`. Bode magnitude plots of the passive suspension and active suspension are shown in the following figure.

```
bodemag(qcar(3,1),'k-.',CL1(3,1),'r-',logspace(0,2.3,140))
legend('Passive','Active (K1)','Location','SouthEast')
```



H-Infinity Control Design 2

In the second design, you are designing the controller to keep the suspension deflection transfer function small. Hence the road disturbance to suspension deflection x_1-x_3 is penalized via the weighting function W_{x1x3} . The W_{x1x3} weight magnitude rolls off above 10 rad/s to roll off before an open-loop zero (23.3 rad/s) in the design.

$$W_{x1x3} = 25 * tf(1, [1/10 \ 1]);$$

The car deflection weight W_{x_1} is not included in this control problem formulation. You can construct the weighted H_∞ plant model for control design, denoted `qcaric2`, using the `sysic` command. As an alternative, you can create `qcaric2` using `iconnect` objects. The same control and measurements are used as in the first design.

```
M = iconnect;
d = icsignal(2);
fs = icsignal(1);
ycar = icsignal(size(qcar,1));
M.Equation{1} = equate(ycar,qcar*[Wref*d(1); fs]);
M.Input = [d;fs];
M.Output = [Wact*fs;Wx1x3*ycar(1);ycar(2)+Wn*d(2)];
qcaric2 = M.System;
```

The second H_∞ controller is synthesized with the `hinfscn` command.

```
[K2,Sc12,gam2] = hinfscn(qcaric2,nmeas,ncont);
CL2 = lft(qcar([1:4 2],1:2),K2);
sprintf('H-infinity controller K2 achieved a norm of %2.5g',gam2)
```

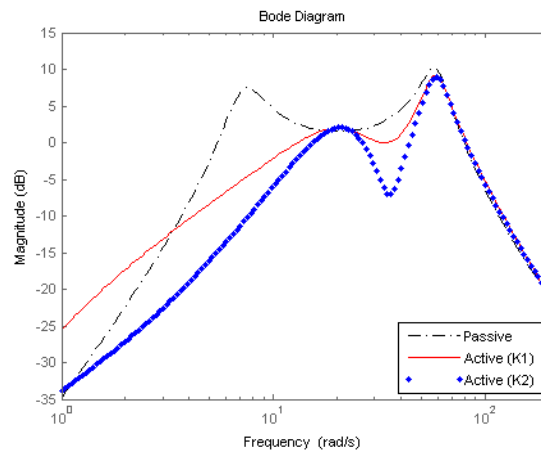
```
ans =
```

```
H-infinity controller K2 achieved a norm of 1.7519
```

Recall that this H_∞ control design emphasizes minimization of suspension deflection over passenger comfort, whereas the first H_∞ design focused on passenger comfort.

You can analyze the H_∞ controller by constructing the closed-loop feedback system `CL2`. Bode magnitude plots of the transfer function from road disturbance to suspension deflection for both controllers and the passive suspension system are shown in the following figure.

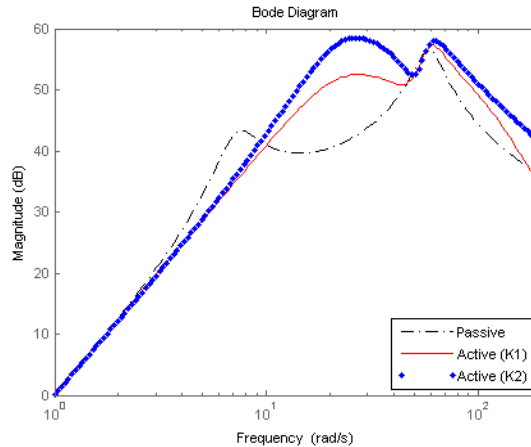
```
bodemag(qcar(3,1),'k-.',CL1(3,1),'r-',CL2(3,1),'b.',logspace(0,2.3,140))
legend('Passive','Active (K1)','Active (K2)','Location','SouthEast')
```



The dotted and solid lines in the figure are the closed-loop frequency responses that result from the different performance weighting functions selected. Observe the reduction in suspension deflection in the vicinity of the tirehop frequency, $\omega_1 = 56.7$ rad/s, and the corresponding increase in the acceleration frequency response in this vicinity. Also, compared to design 1, a reduction in suspension deflection has been achieved for frequencies below the rattlespace frequency, $\omega_2 = 23.3$ rad/s.

The second H_∞ control design attenuates both resonance modes, whereas the first controller focused its efforts on the first mode, the rattlespace frequency at 23.3 rad/s.

```
bodemag(qcar(2,1), 'k-.', CL1(2,1), 'r-', CL2(2,1), 'b.', logspace(0,2.3,140))
legend('Passive', 'Active (K1)', 'Active (K2)', 'Location', 'SouthEast')
```



All the analysis till now has been in the frequency domain. Time-domain performance characteristics are critical to the success of the active suspension system on the car. Time response plots of the two H_∞ controllers are shown in following figures. The dashed, solid, and dotted lines correspond to the passive suspension, H_∞ controller 1, and controller 2 respectively. All responses correspond to the road disturbance $r(t)$:

$$r(t) = a(1 - \cos 8\pi t) \text{ for } 0 \leq t \leq 0.25\text{s}$$

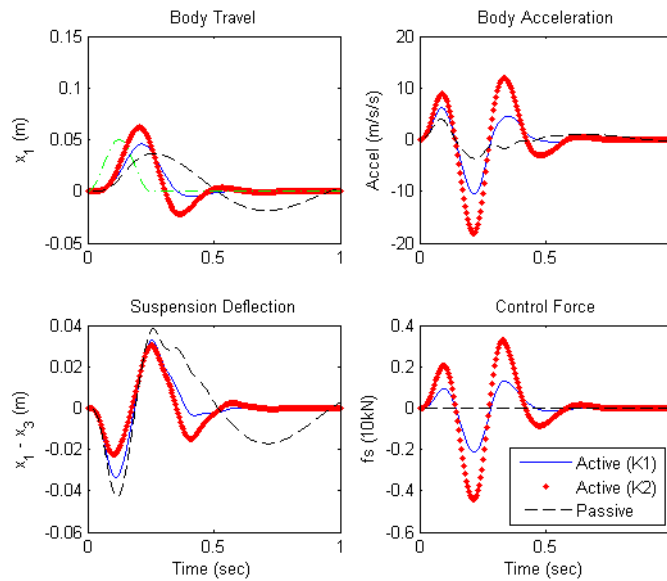
$r(t) = 0$ otherwise.

where $\alpha=0.025$ corresponds to a road bump of peak magnitude 5 cm. In the following plots, observe that the acceleration response of design 1 to the 5 cm bump is very good; however the suspension deflection is larger than for design 2. This is because suspension deflection was not penalized in this design. The suspension deflection response of design 2 to a 5 cm bump is good. However, the acceleration response to the 5 cm bump is much inferior to design 1. Once again this is because car body displacement and acceleration were not penalized in design 2.

```

time = 0:0.005:1;
roaddist = 0*time;
roaddist(1:51) = 0.025*(1-cos(8*pi*time(1:51)));
[p1,t] = lsim(qcar(1:4,1),roaddist,time);
[y1,t] = lsim(CL1(1:4,1),roaddist,time);
[y2,t] = lsim(CL2(1:4,1),roaddist,time);
subplot(221)
plot(t,y1(:,1),'b-',t,y2(:,1),'r.',t,p1(:,1),'k--',t,...
roaddist,'g-.')
    title('Body Travel')
    ylabel('x_1 (m)')
subplot(222)
    plot(t,y1(:,2),'b-',t,y2(:,2),'r.',t,p1(:,2),'k--')
    title('Body Acceleration')
    ylabel('Accel (m/s/s)')
subplot(223)
    plot(t,y1(:,3),'b-',t,y2(:,3),'r.',t,p1(:,3),'k--')
    title('Suspension Deflection')
    xlabel('Time (sec)')
    ylabel('x_1 - x_3 (m)')
subplot(224)
    plot(t,y1(:,4),'b-',t,y2(:,4),'r.',t,p1(:,4),'k--')
    title('Control Force')
    xlabel('Time (sec)')
    ylabel('fs (10kN)')

```



Designs 1 and 2 represent extreme ends of the performance tradeoff spectrum. This section described synthesis of H_∞ to achieve the performance objectives on the active suspension system. Equally, if not more important, is the design of controllers robust to model error or uncertainty.

The goal of every control design is to achieve the desired performance specifications on the nominal model as well as other plants that are *close* to the nominal model. In other words, you want to achieve the performance objectives in the presence of model error or uncertainty. This is called *robust performance*. In the next section, you will design a controller that achieves robust performance using the μ -synthesis control design methodology. The active suspension system again serves as the example. Instead of assuming a perfect actuator, a nominal actuator model with modeling error is introduced into the control problem.

Control Design via Mu Synthesis

The active suspension H_∞ controllers designed in the previous section ignored the hydraulic actuator dynamics. In this section, you will include a first-order model of the hydraulic actuator dynamics as well as an uncertainty model to

account for differences between the actuator model and the actual actuator dynamics.

The nominal model for the hydraulic actuator is

```
actnom = tf(1,[1/60 1]);
```

The actuator model itself is uncertain. You can describe the actuator model error as a set of possible models using a weighting function. At low frequency, below 4 rad/s, it can vary up to 10% from its nominal value. Around 4 rad/s the percentage variation starts to increase and reaches 400% at approximately 800 rad/s. The model uncertainty is represented by the weight W_{unc} , which corresponds to the frequency variation of the model uncertainty and the uncertain LTI dynamic object Δ_{unc} defined as `unc`.

```
Wunc = 0.10*tf([1/4 1],[1/800 1]);
unc = ultidyn('unc',[1 1]);
actmod = actnom*(1+ Wunc*unc)
```

```
actmod =
```

```
Uncertain continuous-time state-space model with 1 outputs, 1 inputs, 2 s
```

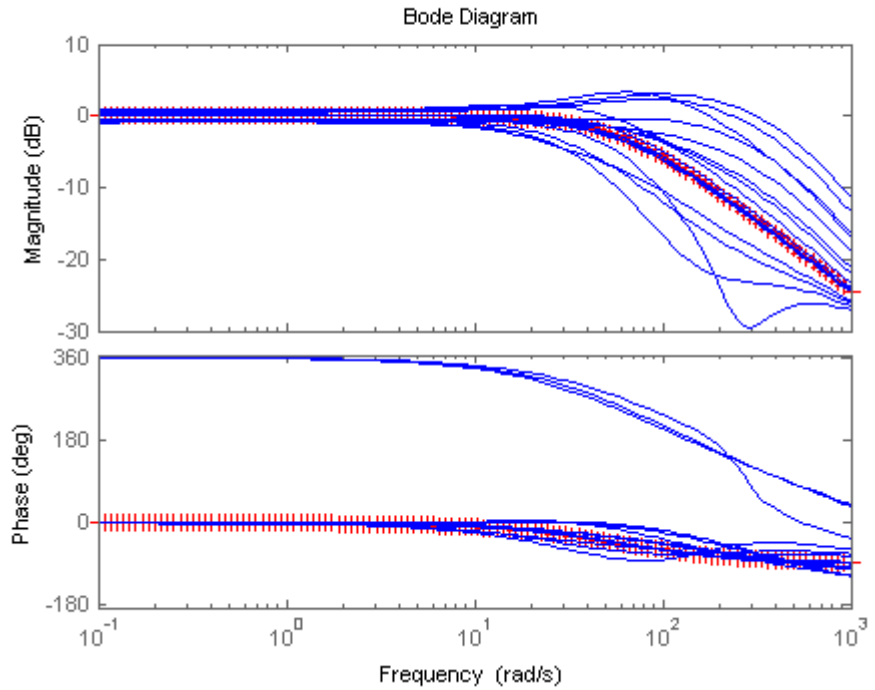
```
The model uncertainty consists of the following blocks:
```

```
unc: Uncertain 1x1 LTI, peak gain = 1, 1 occurrences
```

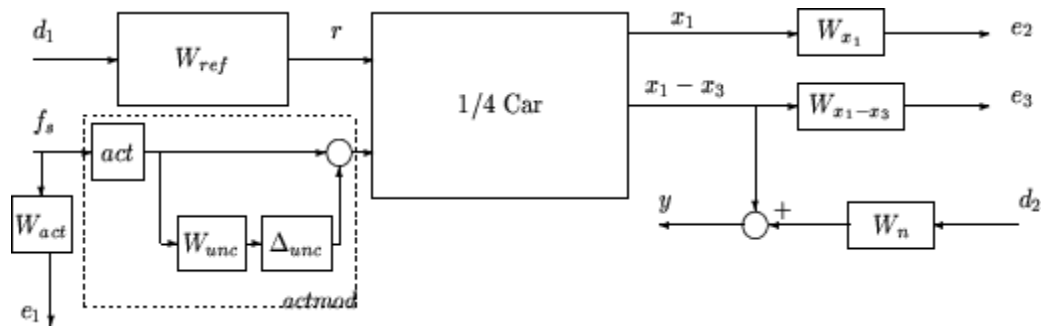
```
Type "actmod.NominalValue" to see the nominal value, "get(actmod)" to see a
```

The actuator model `actmod` is an uncertain state-space system. The following Bode plot shows the nominal actuator model, `actnom`, denoted with a '+' symbol, and 20 random actuator models described by `actmod`. (Because `bodeplot` computes random samples of the `uss` model `actmod`, your plot might not be identical.)

```
bodeplot(actnom,'r+',actmod,'b',logspace(-1,3,120))
```



The uncertain actuator model *actmod* represents the model of the hydraulic actuator used for control. The revised control design interconnection diagram is



You are designing the controller for passenger comfort, as in the first H_∞ control design, hence the car body deflection x_1 is penalized with W_{x1} . The uncertain weighted H_∞ plant model for control design, denoted `qcaricunc`, is using the `sysic` command. As previously described, the control signal corresponds to the last input to `qcaric1`, `fs`. The car body acceleration, which is noisy, is the measured signal and corresponds to the last output of `qcaricunc`.

```
systemnames = 'qcar Wn Wref Wact Wx1 actmod';
inputvar = '[ d1; d2; fs ]';
outputvar = '[ Wact; Wx1; qcar(2)+Wn ]';
input_to_actmod = '[ fs ]';
input_to_qcar = '[ Wref; fs]';
input_to_Wn = '[ d2 ]';
input_to_Wref = '[ d1 ]';
input_to_Wact = '[ fs ]';
input_to_Wx1 = '[ qcar(1) ]';
qcaricunc = sysic;
```

A μ -synthesis controller is synthesized using D - K iteration with the `dksyn` command. The D - K iteration procedure is an approximation to μ -synthesis that attempts to synthesize a controller that achieves robust performance [1], [10], [11], [12]. There is one control input, the hydraulic actuator force, and one measurement signal, the car body acceleration.

```
[Kdk,CLdk,gdk] = dksyn(qcaricunc,nmeas,ncont);
CLdkunc = lft(qcar([1:4 2],1:2)*blkdiag(1,actmod),Kdk);
sprintf('mu-synthesis controller Kdk achieved a norm of %2.5g',gdk)
```

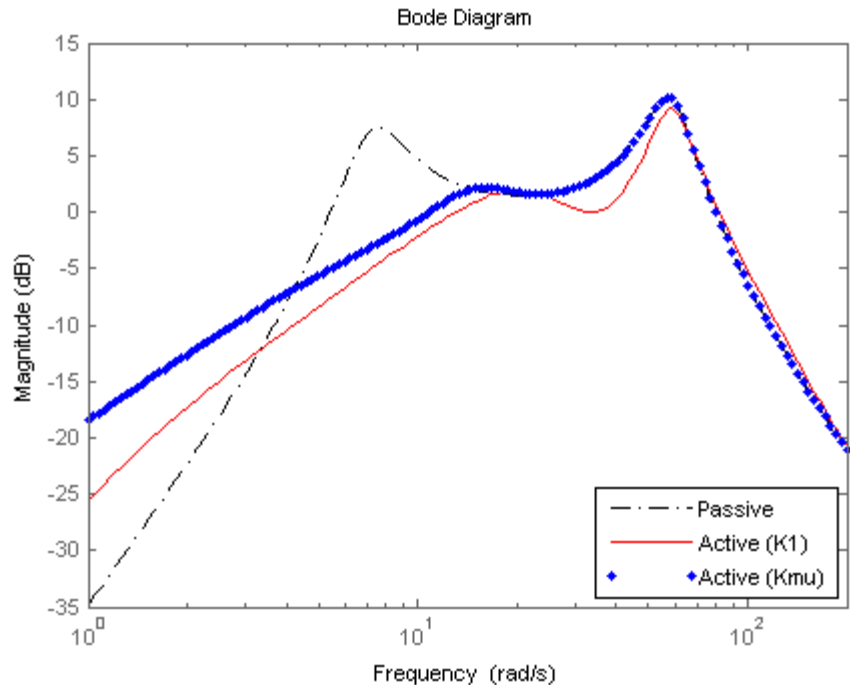
```
ans =
```

```
mu-synthesis controller Kdk achieved a norm of 0.56712
```

You can analyze the performance of the μ -synthesis controller by constructing the closed-loop feedback system `CLdkunc`. Bode magnitude plots of the passive suspension and active suspension systems on the nominal actuator model with H_∞ design 1 and the μ -synthesis controller are shown in the following figure. Note that the μ -synthesis controller better attenuates the first resonant mode at the expense of decreased performance below 3 rad/s.

```
bodemag(qcar(3,1), 'k-.', CL1(3,1), 'r-', CLdkunc.Nominal(3,1), 'b.', ...
```

```
logspace(0,2.3,140)
legend('Passive','Active (K1)','Active (Kmu)','Location','SouthEast')
```



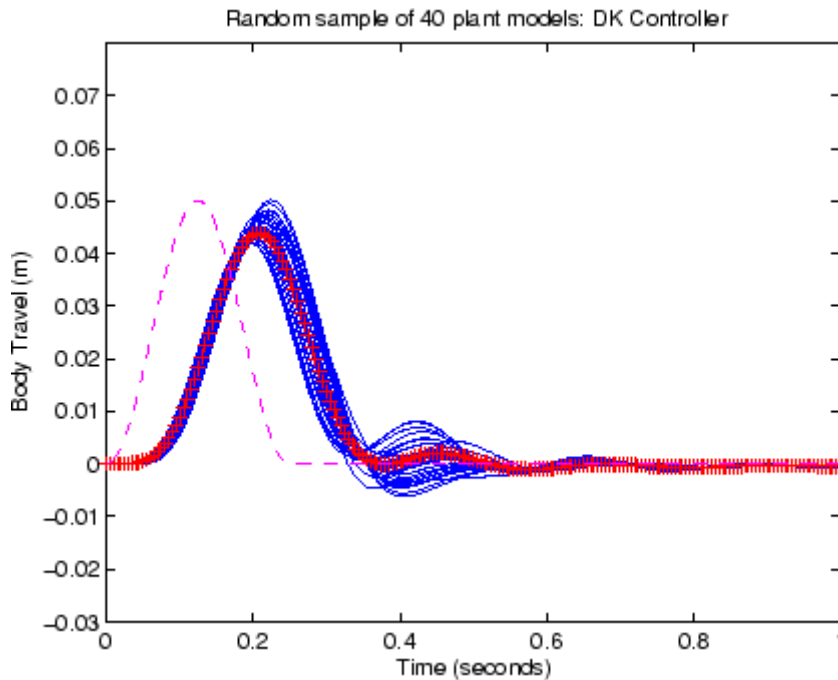
It is important to understand how robust both controllers are in the presence of model error. You can simulate the active suspension system with the H_∞ design 2 and the μ -synthesis controller. The uncertain closed-loop systems, CL2unc and CLdkunc, are formed with K2 and Kdk, respectively. For each uncertain system, 40 random plant models in the model set are simulated. As you can see, both controllers are robust and perform well in the presence of actuator model error. The μ -synthesis controller Kdk achieves slightly better performance than H_∞ design 1.

```
CL2unc = lft(qcar([1:4 2],1:2)*blkdiag(1,actmod),K2);
[CLdkunc40,dksamples] = usample(CLdkunc,40);
CL2unc40 = usubs(CL2unc,dksamples);
nsamp = 40;
```

```

for i=1:nsamp
    [ymusamp,t] = lsim(CLdkunc40(1:4,1,i),roaddist,time);
    plot(t,ymusamp(:,1),'b')
    hold on
end
[ymusamp,t] = lsim(CLdkunc.Nominal(1:4,1),roaddist,time);
plot(t,ymusamp(:,1),'r+',t,roaddist,'m--')

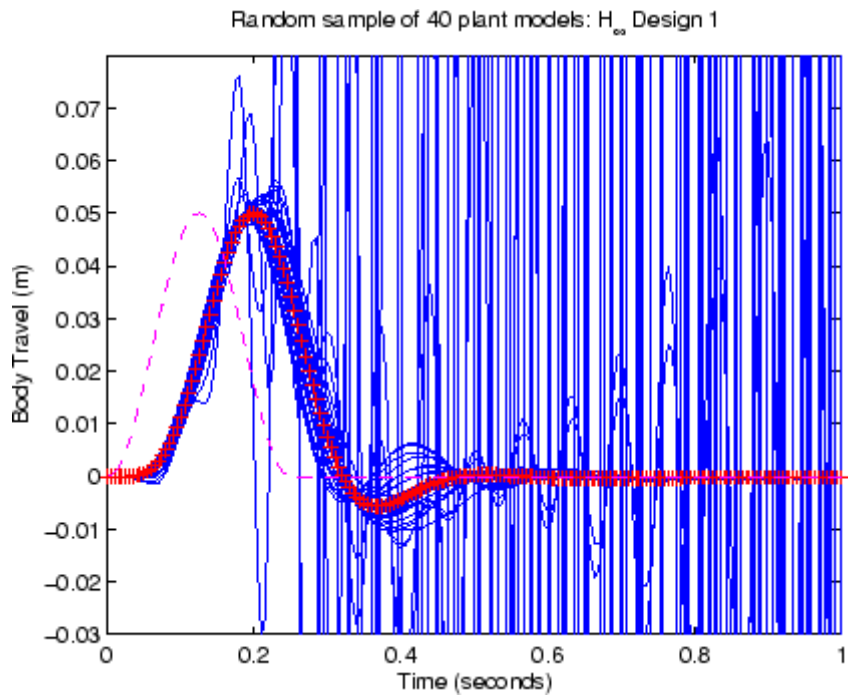
```



```

for i=1:nsamp
    [y1samp,t] = lsim(CL2unc40(1:4,1,i),roaddist,time);
    plot(t,y1samp(:,1),'b')
    hold on
end
[y1samp,t] = lsim(CL2unc.Nominal(1:4,1),roaddist,time);
plot(t,y1samp(:,1),'r+',t,roaddist,'m--')

```



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Tuning Fixed Control Architectures

- “What Is a Fixed-Structure Control System?” on page 6-3
- “Choosing an Approach to Fixed Control Structure Tuning” on page 6-4
- “Difference Between Fixed-Structure Tuning and Traditional H-Infinity Synthesis” on page 6-5
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What Is a Fixed-Structure Control System?

Fixed-structure control systems are control systems that have predefined architectures and controller structures. For example,

- A single-loop SISO control architecture where the controller is a fixed-order transfer function, a PID controller, or a PID controller plus a filter.
- A MIMO control architecture where the controller has fixed order and structure. For example, a 2-by-2 decoupling matrix plus two PI controllers is a MIMO controller of fixed order and structure.
- A multiple-loop SISO or MIMO control architecture, including nested or cascaded loops, with multiple gains and dynamic components to tune.

You can use `systemtune`, `looptune` or `hinfstruct` for frequency-domain tuning of virtually any SISO or MIMO feedback architecture to meet your design requirements. You can use both approaches to tune fixed structure control systems in either MATLAB or Simulink (requires Simulink Control Design™).

Choosing an Approach to Fixed Control Structure Tuning

Robust Control Toolbox includes several commands for frequency-domain tuning of control systems with fixed control structure.

- `systemtune` (or `s1Tunable.systemtune`) — Automatically tunes control systems from high-level design requirements you specify, such as reference tracking, disturbance rejection, and stability margins. `systemtune` jointly tunes all the free parameters of your control system regardless of its architecture and the number of feedback loops it contains.

`systemtune` can also robustly tune control systems for multiple plant models or multiple operating conditions of the plant.

For more information, see “Set Up Your Control System for Tuning with `systemtune`” on page 6-34.

- `looptune` (or `s1Tunable.looptune`) — Automatic tuning of SISO or MIMO feedback loops. `looptune` tunes the a SISO or MIMO feedback loop to high-level design requirements such as loop bandwidth, reference tracking, or stability margins. For more information, see “Set Up Your Control System for Tuning with `looptune`” on page 6-8.
- `hinfstruct` — H_∞ synthesis with fixed-structure controllers. Using `hinfstruct` requires you to formulate the H_∞ synthesis problem and choose weighting functions yourself. For more information, see “What Is `hinfstruct`?” on page 6-18.

Difference Between Fixed-Structure Tuning and Traditional H-Infinity Synthesis

All of the tuning commands `systemtune`, `looptune`, and `hinfstruct` tune the controller parameters by optimizing the H_∞ norm across a closed-loop system (see [1]). However, these functions differ in important ways from traditional H_∞ methods.

Traditional H_∞ synthesis (performed using the `hifsyn` or `loopsyn` commands) designs a full-order, centralized controller. Traditional H_∞ synthesis provides no way to impose structure on the controller and often results in a controller that has high-order dynamics. Thus, the results can be difficult to map to your specific real-world control architecture. Additionally, traditional H_∞ synthesis requires you to express all design requirements in terms of a single weighted MIMO transfer function.

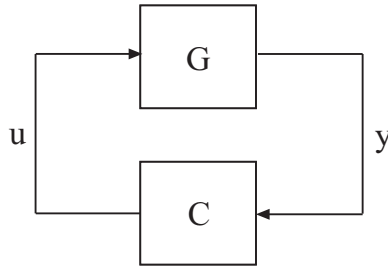
In contrast, structured H_∞ synthesis allows you to describe and tune the specific control system with which you are working. You can specify your control architecture, including the number and configuration of feedback loops. You can also specify the complexity, structure, and parametrization of each tunable component in your control system, such as PID controllers, gains, and fixed-order transfer functions. Additionally, you can easily combine requirements on separate closed-loop transfer functions.

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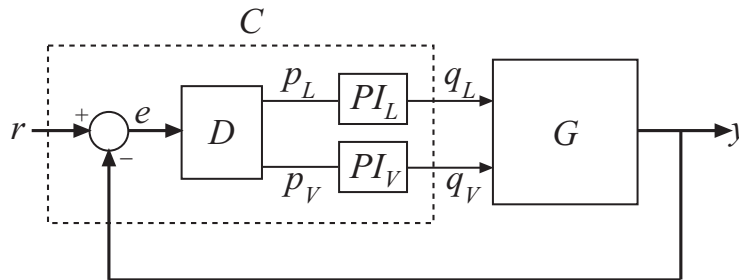
How looptune Sees a Control System

looptune tunes the feedback loop illustrated below to meet default requirements or requirements that you specify.



C represents the controller and G represents the plant. The sensor outputs y (measurements) and actuator outputs u (controls) define the boundary between plant and controller. The controller is the portion of your control system whose inputs are measurements, and whose outputs are control signals. Conversely, the plant is the remainder—the portion of your control system that receives control signals as inputs, and produces measurements as outputs.

For example, in the control system of the following illustration, the controller C receives the measurement y , and the reference signal r . The controller produces the controls q_L and q_V as outputs.



The controller C has a fixed internal structure. C includes a gain matrix D , the PI controllers PI_L and PI_V , and a summing junction. The looptune command tunes free parameters of C such as the gains in D and the

proportional and integral gains of PI_L and PI_V. You can also use looptune to co-tune free parameters in both C and G.

Set Up Your Control System for Tuning with looptune

In this section...
“Set Up Your Control System for looptune in MATLAB” on page 6-8
“Set Up Your Control System for looptune in Simulink” on page 6-8

Set Up Your Control System for looptune in MATLAB

To set up your control system in MATLAB for tuning with looptune:

- 1 Parameterize the tunable elements of your controller. You can use predefined structures such as `ltiblock.pid`, `ltiblock.gain`, and `ltiblock.tf`. Or, you can create your own structure from elementary tunable parameters (`realp`).
- 2 Use model interconnection commands such as `series` and `connect` to build a tunable `genss` model representing the controller `C0`.
- 3 Create a Numeric LTI model representing the plant `G`. For co-tuning the plant and controller, represent the plant as a tunable `genss` model.

Set Up Your Control System for looptune in Simulink

To set up your control system in Simulink for tuning with `sITunable.looptune` (requires Simulink Control Design software):

- 1 Use `sITunable` to create an interface to the Simulink model of your control system. When you create the interface, you specify which blocks to tune in your model.
- 2 Use `sITunable.addControl` and `sITunable.addMeasurement` to specify the control and measurement signals that define the boundaries between plant and controller. Use `sITunable.addSwitch` to mark optional loop-opening or signal injection sites for specifying and assessing open-loop requirements.

The `sITunable` interface automatically linearizes your Simulink model. The `sITunable` interface also automatically parametrizes the blocks that you

specify as tunable blocks. For more information, see the `sITunable` reference page and “Supported Blocks for Tuning in Simulink” on page 6-40.

Related Examples

- “Tune a MIMO Control System for a Specified Bandwidth” on page 6-12
- “Tuning Feedback Loops with LOOPTUNE” on page 6-51

Concepts

- “How looptune Sees a Control System” on page 6-6

Performance and Robustness Specifications for `looptune`

When you use `looptune`, you specify a target control bandwidth as a target open-loop gain crossover, the `wc` input to `looptune`. By default, `looptune` automatically tunes your control system to a loop shape of wc/s to enforce the following requirements:

- Performance—integral action at low frequency
- Roll-off—gain roll-off at frequencies above the crossover frequency
- Stability margins—adequate robustness at frequencies above the crossover frequency. You can override the default stability margins using the `looptuneOptions` command to create an options set to pass to `looptune`.

`looptune` is a wrapper for the `systemtune` command. `looptune` simplifies the setup for less complex problems by automatically tuning your feedback loop to meet the default performance requirements. You can specify additional design requirements using `TuningGoal` requirements objects. The following table summarizes the types of design requirements you can specify.

Design Requirement	Description	How To Specify
Reference Tracking	Requirement that a specified output track a specified input	<code>TuningGoal.Tracking</code>
Stability margins	Gain and phase margins, defined as multi-loop disk margins for MIMO systems (see <code>loopmargin</code>)	<code>TuningGoal.Margins</code>

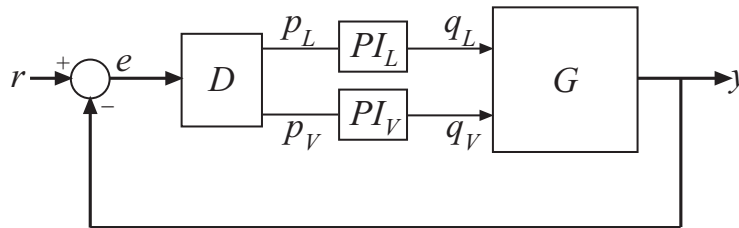
Design Requirement	Description	How To Specify
Gain Limits	Limit on the maximum gain and the gain profile across specified I/Os. For example: <ul style="list-style-type: none"> • Disturbance rejection • Custom roll-off specification • Frequency-weighted peak gain limit • Noise amplification limit 	<ul style="list-style-type: none"> • <code>TuningGoal.Gain</code> • <code>TuningGoal.WeightedGain</code> • <code>TuningGoal.Variance</code> • <code>TuningGoal.WeightedVariance</code>
Custom Loop Shape	Constraints on the open-loop gain profile of a specified SISO or MIMO feedback loop	<code>TuningGoal.LoopShape</code>
Pole Placement	Constraints on location of tunable closed-loop system poles and controller poles	<ul style="list-style-type: none"> • <code>TuningGoal.Poles</code> • <code>TuningGoal.StableController</code>

Related Examples

- “Using Design Requirement Objects” on page 6-74

Tune a MIMO Control System for a Specified Bandwidth

This example shows how to tune the control system of the following illustration to achieve crossover between 0.1 and 1 rad/min, using looptune.



The setpoint signal r , the error signal e , and the output signal y are vector-valued signals of dimension 2. The 2-by-2 plant G is represented by:

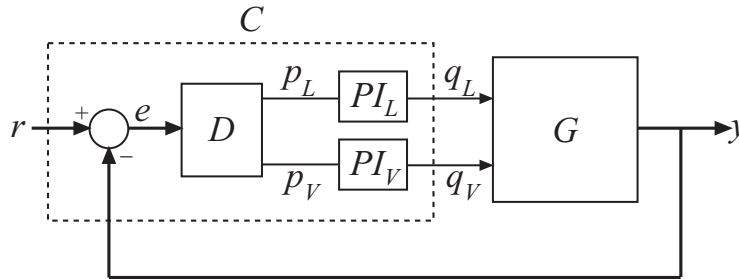
$$G(s) = \frac{1}{75s+1} \begin{bmatrix} 87.8 & -86.4 \\ 108.2 & -109.6 \end{bmatrix}.$$

Create a Numeric LTI model representing the plant.

```
s = tf('s');
G = 1/(75*s+1)*[87.8 -86.4; 108.2 -109.6];
G.InputName = {'qL', 'qV'};
G.OutputName = 'y';
```

Name the inputs and outputs of your model using the InputName and OutputName properties. Doing so tells looptune how to interconnect your plant and controller.

The controller, C , has a fixed structure that includes three tunable components: the 2-by-2 decoupling matrix, D , and the two PI controllers, PI_L and PI_V . (For more information about this control system, see Decoupling Controller for a Distillation Column.)



C receives the measurement signal y and the reference r as inputs. C produces the control signals q_L and q_V .

Use Control Design Blocks to represent the tunable components of the controller.

```
D = ltiblock.gain('Decoupler',eye(2));
D.InputName = 'e';
D.OutputName = {'pL','pV'};
```

```
PI_L = ltiblock.pid('PI_L','pi');
PI_L.InputName = 'pL';
PI_L.OutputName = 'qL';
```

```
PI_V = ltiblock.pid('PI_V','pi');
PI_V.InputName = 'pV';
PI_V.OutputName = 'qV';
```

Create the summing junction using `sumblk`.

```
sum1 = sumblk('e = r - y',2);
```

Connect the components using model interconnection commands.

```
C0 = connect(PI_L,PI_V,D,sum1,{'r','y'},{'qL','qV'});
```

The `genss` model `C0` represents the controller structure. `C0` specifies which controller parameters are tunable, and contains the initial values of those parameters.

Tune the control system.

```
wc = [0.1,1];
options = looptuneOptions('RandomStart',5);
[G,C,gam,Info] = looptune(G,C0,wc,options);
```

```
Final: Peak gain = 0.949, Iterations = 27
```

The `looptune` algorithm uses the input and output names of `C0` and `G` to form the closed loop system for tuning. The input `wc` specifies that the crossover frequency of each loop in the system falls between 0.1 and 1 rad/min.

Setting `RandomStart` to 5 causes `looptune` to perform five additional optimization runs, beginning from parameter values it chooses randomly. Most runs return 0.949 as optimal gain value, suggesting that this local minimum has a wide region of attraction and is likely to be the global optimum.

The output `gam` is a normalized parameter indicating the degree of success meeting the tuning requirements. `gam` is the best value of the peak gain over all optimization runs. `gam <=1` indicates that all requirements are satisfied. A `gam` value much greater than one indicates failure to meet some requirement. In this example, the best value of `gam = 0.949` indicates success meeting the specified crossover requirement.

`looptune` also returns `C`, the tuned version of `C0` that yields the best peak gain value. Examine the tuned values of all components using `showTunable`.

```
showTunable(C)
```

```
Decoupler =
```

```

d =
      u1      u2
y1  1.075 -0.8273
y2 -1.581  1.328
```

```
Name: Decoupler
```

```
Static gain.
```

```
-----
PI_L =
```

$$K_p + K_i * \frac{1}{s}$$

with $K_p = 2.82$, $K_i = 0.0398$

Name: PI_L

Continuous-time PI controller in parallel form.

PI_V =

$$K_p + K_i * \frac{1}{s}$$

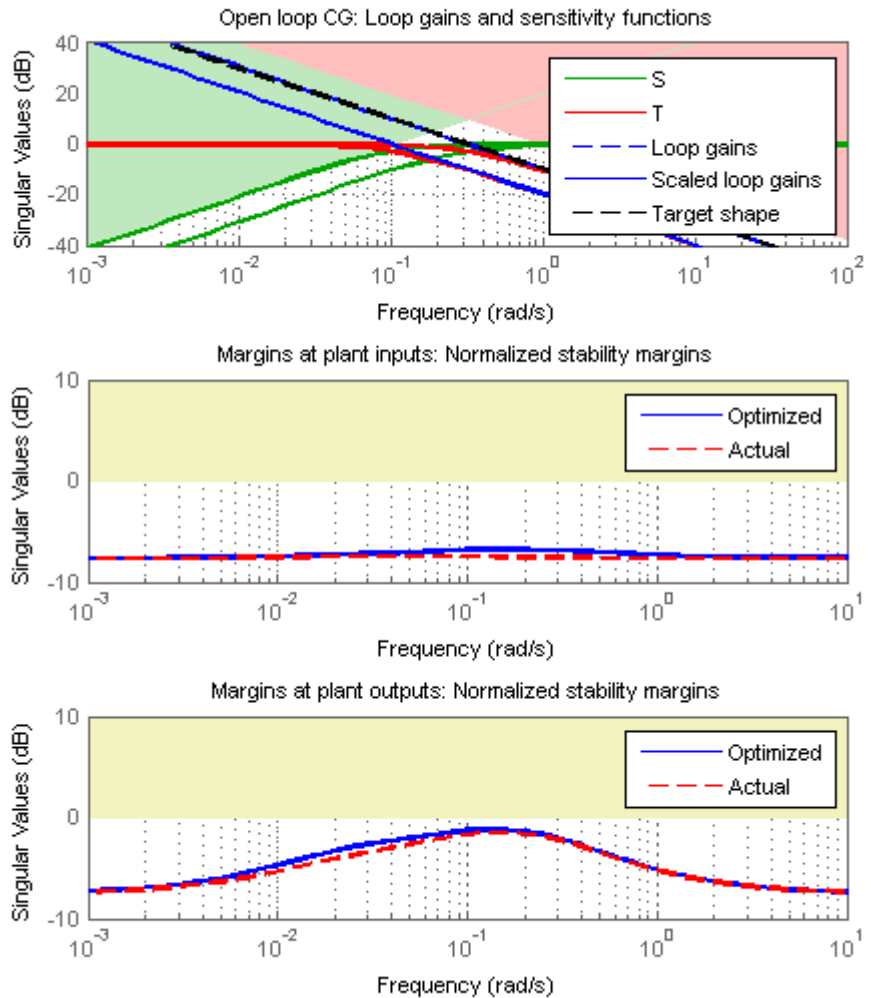
with $K_p = -1.94$, $K_i = -0.0303$

Name: PI_V

Continuous-time PI controller in parallel form.

Validate the frequency domain response of the tuned result using loopview.

loopview(G,C,Info)

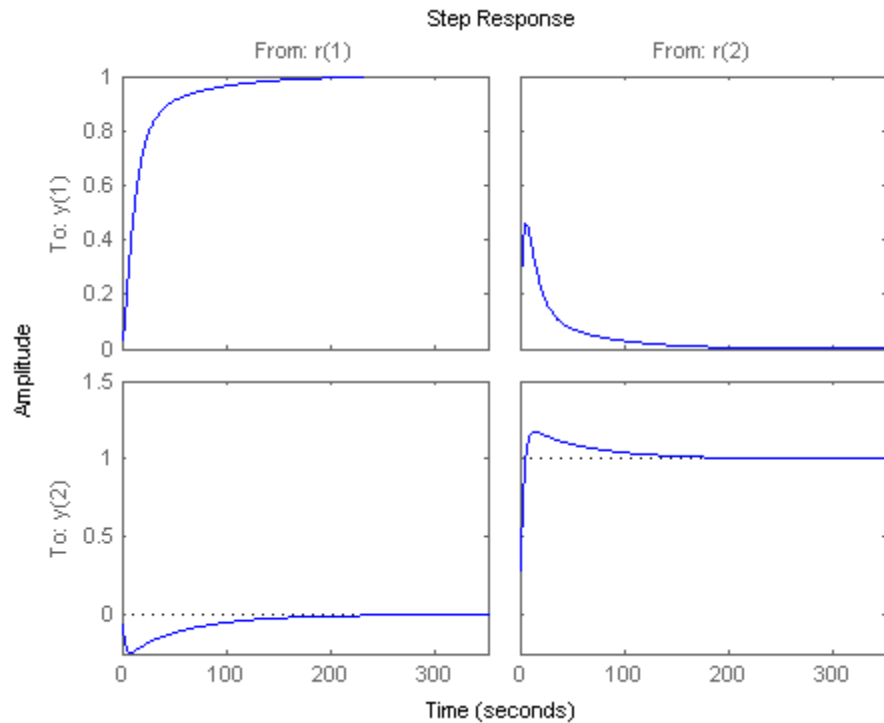


The first plot shows that the open-loop gain crossovers fall within the specified interval $[0.1, 1]$. This plot also includes the maximum and tuned values of the sensitivity function $S = (I - GC)^{-1}$ and complementary sensitivity $T = I - S$.

The second and third plots show that the MIMO stability margins of the tuned system (blue curve) do not exceed the upper limit (yellow curve).

Validate the time domain response using Control System Toolbox analysis commands such as `step`.

```
T = connect(G,C,'r','y');
step(T)
```



The tuned controller provides good setpoint tracking with reasonably small crosstalk between channels. To impose further constraints on crosstalk, use the `TuningGoal.Gain` requirements object.

What Is hinfstruct?

`hinfstruct` lets you use the frequency-domain methods of H_∞ synthesis to tune control systems that have predefined architectures and controller structures.

To use `hinfstruct`, you describe your control system as a Generalized LTI model that keeps track of the tunable components of your system. `hinfstruct` tunes those parameters by minimizing the closed-loop gain from the system inputs to the system outputs (the H_∞ norm).

`hinfstruct` is the counterpart of `hinfsyn` for fixed-structure controllers. The methodology and algorithm behind `hinfstruct` are described in [1].

Formulating Design Requirements as H-Infinity Constraints

Control design requirements are typically performance measures such as response speed, control bandwidth, roll-off, and steady-state error. To use `hinfstruct`, first express the design requirements as constraints on the closed-loop gain.

You can formulate design requirements in terms of the closed-loop gain using loop shaping. *Loop shaping* is a common systematic technique for defining control design requirements for H_∞ synthesis. In loop shaping, you first express design requirements as open-loop gain requirements.

For example, a requirement of good reference tracking and disturbance rejection is equivalent to high (>1) open-loop gain at low frequency. A requirement of insensitivity to measurement noise or modeling error is equivalent to a low (<1) open-loop gain at high frequency. You can then convert these open-loop requirements to constraints on the closed-loop gain using weighting functions.

This formulation of design requirements results in a H_∞ constraint of the form:

$$\|H(s)\|_\infty < 1,$$

where $H(s)$ is a closed-loop transfer function that aggregates and normalizes the various requirements.

For an example of how to formulate design requirements for H_∞ synthesis using loop shaping, see “Fixed-Structure H-infinity Synthesis with HINFSTRUCT” on page 6-211.

For more information about constructing weighting functions from design requirements, see “H-Infinity Performance” on page 5-9.

Structured H-Infinity Synthesis Workflow

Performing structured H_∞ synthesis requires the following steps:

- 1** Formulate your design requirements as H_∞ constraints, which are constraints on the closed-loop gains from specific system inputs to specific system outputs.
- 2** Build tunable models of the closed-loop transfer functions of Step 1.
- 3** Tune the control system using `hinfstruct`.
- 4** Validate the tuned control system.

Build Tunable Closed-Loop Model for Tuning with hinfstruct

In the previous step you expressed your design requirements as a constraint on the H_∞ norm of a closed-loop transfer function $H(s)$.

The next step is to create a Generalized LTI model of $H(s)$ that includes all of the fixed and tunable elements of the control system. The model also includes any weighting functions that represent your design requirements. There are two ways to obtain this tunable model of your control system:

- Construct the model using Control System Toolbox commands.
- Obtain the model from a Simulink model using Simulink Control Design commands.

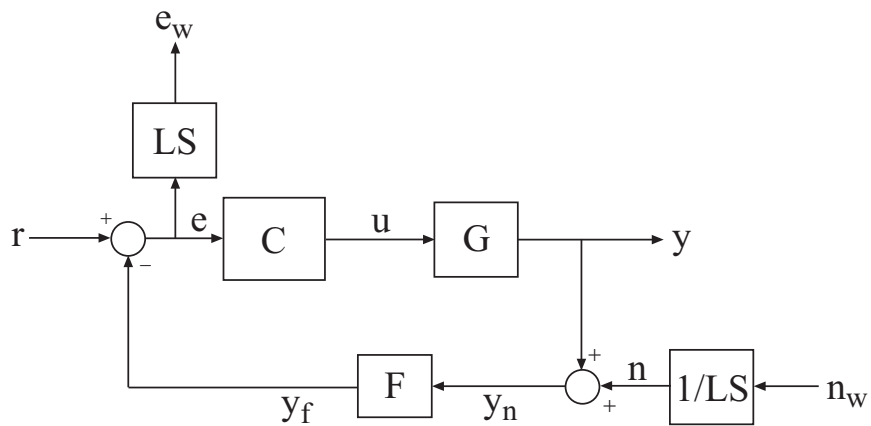
Constructing the Closed-Loop System Using Control System Toolbox Commands

To construct the tunable generalized linear model of your closed-loop control system in MATLAB:

- 1** Use commands such as `tf`, `zpk`, and `ss` to create numeric linear models that represent the fixed elements of your control system and any weighting functions that represent your design requirements.
- 2** Use tunable models (either Control Design Blocks or Generalized LTI models) to model the tunable elements of your control system. For more information about tunable models, see “Models with Tunable Coefficients” in the *Control System Toolbox User’s Guide*.
- 3** Use model-interconnection commands such as `series`, `parallel`, and `connect` to construct your closed-loop system from the numeric and tunable models.

Example: Modeling a Control System With a Tunable PI Controller and Tunable Filter

This example shows how to construct a tunable generalized linear model of the following control system for tuning with `hinfstruct`.



This block diagram represents a head-disk assembly (HDA) in a hard disk drive. The architecture includes the plant G in a feedback loop with a PI controller C and a low-pass filter, $F = a/(s+a)$. The tunable parameters are the PI gains of C and the filter parameter a .

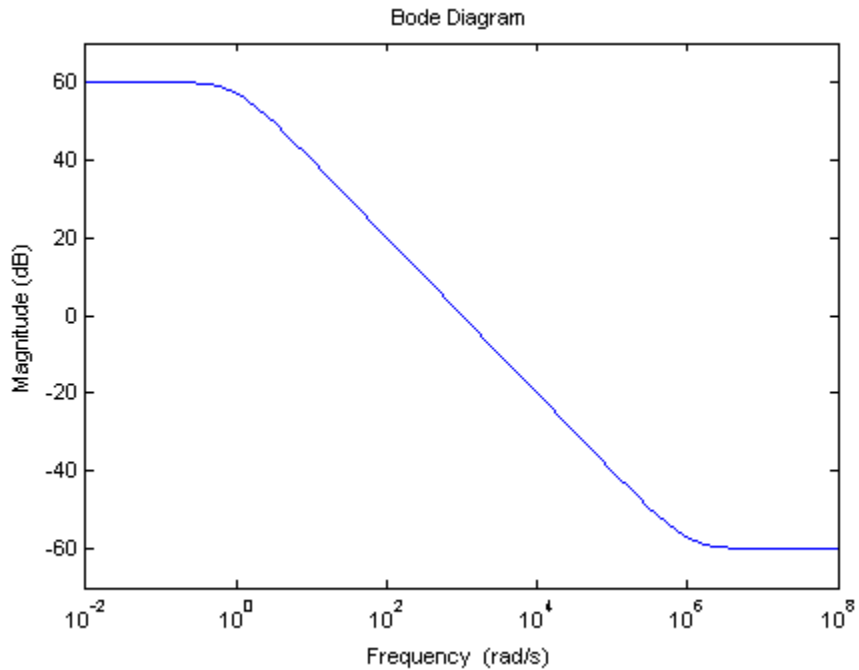
The block diagram also includes the weighting functions LS and $1/LS$, which express the loop-shaping requirements. Let $T(s)$ denote the closed-loop transfer function from inputs (r, n_w) to outputs (y, e_w) . Then, the H_∞ constraint:

$$\|T(s)\|_\infty < 1$$

approximately enforces the target open-loop response shape LS . For this example, the target loop shape is

$$LS = \frac{1 + 0.001 \frac{s}{\omega_c}}{0.001 + \frac{s}{\omega_c}}$$

This value of LS corresponds to the following open-loop response shape.



To tune the HDA control system with `hinfstruct`, construct a tunable model of the closed-loop system $T(s)$, including the weighting functions, as follows.

- 1 Load the plant G from a saved file.

```
load hinfstruct_demo G
```

G is a 9th-order SISO state-space (ss) model.

- 2 Create a tunable model of the PI controller.

You can use the predefined Control Design Block `ltiblock.pid` to represent a tunable PI controller.

```
C = ltiblock.pid('C', 'pi');
```

3 Create a tunable model of the low-pass filter.

Because there is no predefined Control Design Block for the filter $F = a/(s+a)$, use `realp` to represent the tunable filter parameter a . Then create a tunable `genss` model representing the filter.

```
a = realp('a',1);
F = tf(a,[1 a]);
```

4 Specify the target loop shape LC.

```
wc = 1000;
s = tf('s');
LS = (1+0.001*s/wc)/(0.001+s/wc);
```

5 Label the inputs and outputs of all the components of the control system.

Labeling the I/Os allows you to connect the elements to build the closed-loop system $T(s)$.

```
Wn = 1/LS; Wn.InputName = 'nw'; Wn.OutputName = 'n';
We = LS; We.InputName = 'e'; We.OutputName = 'ew';
C.InputName = 'e'; C.OutputName = 'u';
F.InputName = 'yn'; F.OutputName = 'yf';
```

6 Specify the summing junctions in terms of the I/O labels of the other components of the control system.

```
Sum1 = sumblk('e = r - yf');
Sum2 = sumblk('yn = y + n');
```

7 Use `connect` to combine all the elements into a complete model of the closed-loop system $T(s)$.

```
T0 = connect(G,Wn,We,C,F,Sum1,Sum2,{'r','nw'},{'y','ew'});
```

`T0` is a `genss` object, which is a Generalized LTI model representing the closed-loop control system with weighting functions. The `Blocks` property of `T0` contains the tunable blocks `C` and `a`.

```
T0.Blocks
```

```
ans =
```



```
C: [1x1 ltiblock.pid]
a: [1x1 realp]
```

For more information about generalized models of control systems that include both numeric and tunable components, see “Models with Tunable Coefficients” in the Control System Toolbox documentation.

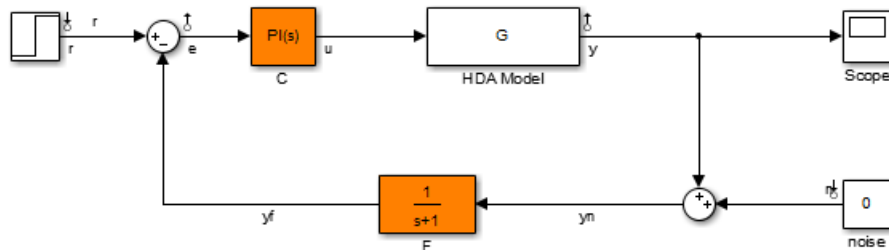
You can now use `hinfstruct` to tune the parameters of this control system. See “Tune the Controller Parameters” on page 6-28.

Constructing the Closed-Loop System Using Simulink Control Design Commands

If you have a Simulink model of your control system and Simulink Control Design software, use `s1Tunable` to create an interface to the Simulink model of your control system. When you create the interface, you specify which blocks to tune in your model. The `s1Tunable` interface allows you to extract a closed-loop model for tuning with `hinfstruct`.

Example: Creating a Weighted Tunable Model of Control System Starting From a Simulink Model

This example shows how to construct a tunable generalized linear model of the control system in the Simulink model `rct_diskdrive`.



To create a generalized linear model of this control system (including loop-shaping weighting functions):

- 1 Open the model.

```
open('rct_diskdrive');
```

- 2** Create an `sITunable` interface to the model. The interface allows you to specify the tunable blocks and extract linearized open-loop and closed-loop responses. (For more information about the interface, see the `sITunable` reference page.)

```
ST0 = sITunable('rct_diskdrive',{'C','F'});
```

This command specifies that `C` and `F` are the tunable blocks in the model. The `sITunable` interface automatically parametrizes these blocks. The default parametrization of the transfer function block `F` is a transfer function with two free parameters. Because `F` is a low-pass filter, you must constrain its coefficients. To do so, specify a custom parameterization of `F`.

```
a = realp('a',1); % filter coefficient
setBlockParam(ST0,'F',tf(a,[1 a]));
```

- 3** Extract a tunable model of the closed-loop transfer function you want to tune.

```
T0 = getIOTransfer(ST0,{'r','n'},{'y','e'});
```

This command returns a `genss` model of the linearized closed-loop transfer function from the reference and noise inputs `r,n` to the measurement and error outputs `y,e`. The error output is needed for the loop-shaping weighting function.

- 4** Define the loop-shaping weighting functions and append them to `T0`.

```
wc = 1000;
s = tf('s');
LS = (1+0.001*s/wc)/(0.001+s/wc);
```

```
T0 = blkdiag(1,LS) * T0 * blkdiag(1,1/LS);
```

The generalized linear model `T0` is a tunable model of the closed-loop transfer function $T(s)$, discussed in “Example: Modeling a Control System With a Tunable PI Controller and Tunable Filter” on page 6-21. $T(s)$ is a weighted closed-loop model of the control system of `rct_diskdrive`. Tuning `T0` to enforce the H_∞ constraint

$$\|T(s)\|_{\infty} < 1$$

approximately enforces the target loop shape LS.

You can now use `hinfstruct` to tune the parameters of this control system. See “Tune the Controller Parameters” on page 6-28.

Tune the Controller Parameters

After you obtain the `genss` model representing your control system, use `hinfstruct` to tune the tunable parameters in the `genss` model .

`hinfstruct` takes a tunable linear model as its input.

For example, you can tune controller parameters for the example discussed in “Build Tunable Closed-Loop Model for Tuning with `hinfstruct`” on page 6-21 using the following command:

```
[T,gamma,info] = hinfstruct(T0);
```

This command returns the following outputs:

- `T`, a `genss` model object containing the tuned values of `C` and `a`.
- `gamma`, the minimum peak closed-loop gain of `T` achieved by `hinfstruct`.
- `info`, a structure containing additional information about the minimization runs.

Interpret the Outputs of hinfstruct

Output Model is Tuned Version of Input Model

T contains the same tunable components as the input closed-loop model T0. However, the parameter values of T are now tuned to minimize the H_∞ norm of this transfer function.

Interpreting gamma

gamma is the smallest H_∞ norm achieved by the optimizer. Examine gamma to determine how close the tuned system is to meeting your design constraints. If you normalize your H_∞ constraints, a final gamma value of 1 or less indicates that the constraints are met. A final gamma value exceeding 1 by a small amount indicates that the constraints are nearly met.

The value of gamma that hinfstruct returns is a local minimum of the gain minimization problem. For best results, use the RandomStart option to hinfstruct to obtain several minimization runs. Setting RandomStart to an integer $N > 0$ causes hinfstruct to run the optimization N additional times, beginning from parameter values it chooses randomly. For example:

```
opts = hinfstructOptions('RandomStart',5);  
[T,gamma,info] = hinfstruct(T0,opts);
```

You can examine gamma for each run to identify an optimization result that meets your design requirements.

For more details about hinfstruct, its options, and its outputs, see the hinfstruct and hinfstructOptions reference pages.

Validate the Controller Design

To validate the `hinfstruct` control design, analyze the tuned output models described in “Interpret the Outputs of `hinfstruct`” on page 6-29. Use these tuned models to examine the performance of the tuned system.

Validating the Design in MATLAB

This example shows how to obtain the closed-loop step response of a system tuned with `hinfstruct` in MATLAB.

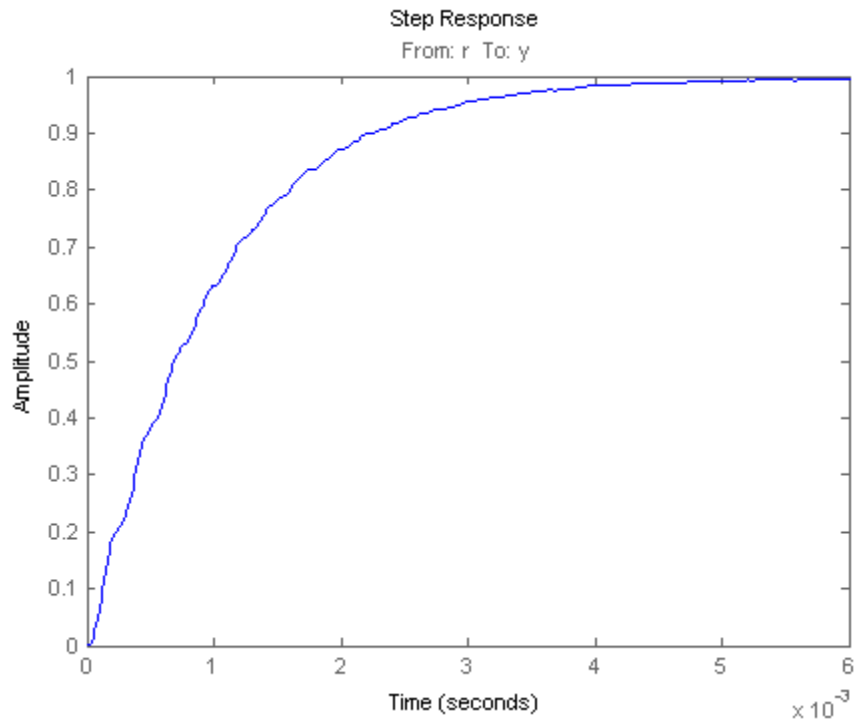
You can use the tuned versions of the tunable components of your system to build closed-loop or open-loop numeric LTI models of the tuned control system. You can then analyze open-loop or closed-loop performance using other Control System Toolbox tools.

In this example, create and analyze a closed-loop model of the HDA system tuned in “Tune the Controller Parameters” on page 6-28. To do so, extract the tuned controller values from the tuned closed-loop model. Use `getBlockValue` to get the tuned value of a tunable Control Design Block. Use `getValue` to evaluate a `genss` model using parameter values from another `genss` model.

```
Ctuned = getBlockValue(T, 'C');  
Ftuned = getValue(F,T);
```

Then construct the tuned closed-loop response of the control system from `r` to `y`.

```
Try = connect(G,Ctuned,Ftuned,Sum1,Sum2, 'r', 'y');  
step(Try)
```



Validating the Design in Simulink

This example shows how to write tuned values to your Simulink model for validation.

The `sITunable` interface linearizes your Simulink model. As a best practice, validate the tuned parameters in your nonlinear model. You can use the `sITunable` interface to do so.

In this example, write tuned parameters to the `rct_diskdrive` system tuned in “Tune the Controller Parameters” on page 6-28.

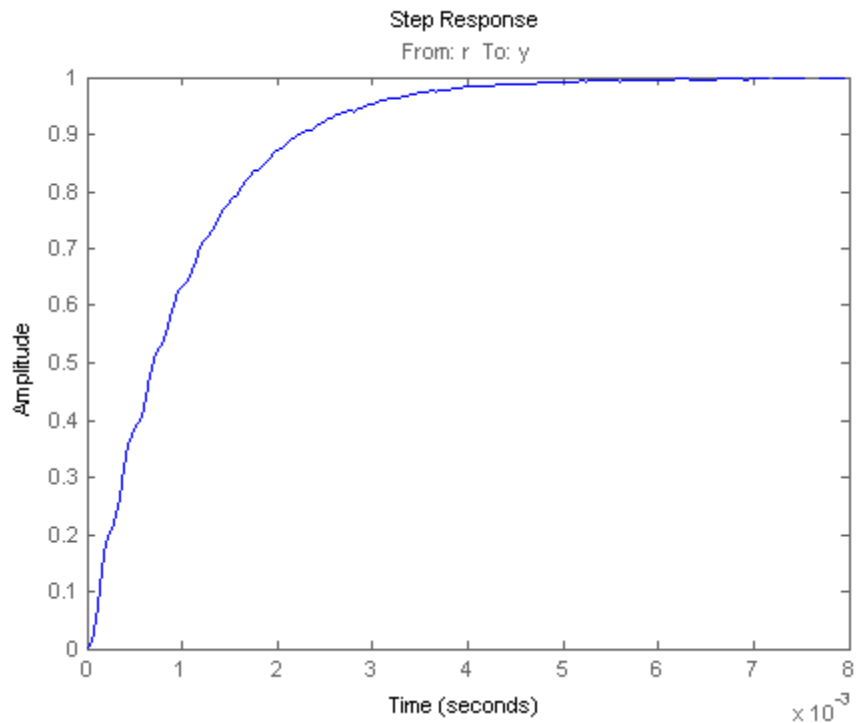
Make a copy of the `sITunable` description of the control system, to preserve the original parameter values. Then propagate the tuned parameter values to the copy.

```
ST = copy(ST0);  
setBlockValue(ST,T);
```

This command writes the parameter values from the tuned, weighted closed-loop model T to the corresponding parameters in the interface ST.

You can examine the closed-loop responses of the linearized version of the control system represented by ST. For example:

```
Try = getIOTransfer(ST,'r','y');
```



Since `hinfstruct` tunes a linearized version of your system, you should also validate the tuned controller in the full nonlinear Simulink model. To do so, write the parameter values from the `sITunable` interface to the Simulink model.


```
writeBlockValue(ST)
```

You can now simulate the model using the tuned parameter values to validate the controller design.

Set Up Your Control System for Tuning with `system`

`system` tunes the free parameters of your control system to meet tuning requirements that you specify. To use `system`, you create a tunable model of your control system in either MATLAB or Simulink.

Set Up Your Control System for `system` in MATLAB

To set up your control system in MATLAB for tuning with `system`:

- 1** Parameterize the tunable elements of your control system. You can use predefined structures such as `ltiblock.pid`, `ltiblock.gain`, and `ltiblock.tf`. Or, you can create your own structure from elementary tunable parameters (`realp`).
- 2** Use model interconnection commands such as `feedback` and `connect` to build a closed-loop model of the overall control system as an interconnection of fixed and tunable components. Use `loopswitch` blocks to mark optional loop-opening or signal injection sites for specifying and assessing open-loop requirements. `system` tunes the resulting `genss` model of the control system.

Set Up Your Control System for `system` in Simulink

To set up your control system in Simulink for tuning with `sITunable.system` (requires Simulink Control Design software):

- 1** Use `sITunable` to create an interface to the Simulink model of your control system. When you create the interface, you specify which blocks to tune in your model.
- 2** Use `sITunable.addIO`, `sITunable.addControl`, and `sITunable.addMeasurement` to mark the input and output signals you need to specify and assess closed-loop requirements.
- 3** Use `sITunable.addSwitch`, `sITunable.addControl`, and `sITunable.addMeasurement` to mark optional loop-opening sites for specifying and assessing open-loop requirements.

The `sITunable` interface automatically linearizes your Simulink model. The `sITunable` interface also automatically parametrizes the blocks that you

specify as tunable blocks. For more information, see the `sITunable` reference page and “Supported Blocks for Tuning in Simulink” on page 6-40.

After setting up your control system, define your design requirements using `TuningGoal` objects as described in “Specifying Design Requirements for `systune`” on page 6-36

.

Related Examples

- “Using Design Requirement Objects” on page 6-74
- “Tuning Control Systems with `SYSTUNE`” on page 6-43

Specifying Design Requirements for systune

When you tune a control system with `systune`, you specify tuning requirements such as reference tracking, maximum gain from specified inputs to specified outputs, or stability margins. You can designate tuning requirements as either soft constraints or hard constraints. `systune` tunes the free parameters of your control system to come as close as possible to satisfying the soft constraints, subject to satisfying the hard constraints.

Specify your tuning requirements using `TuningGoal` requirements objects. The following table summarizes the types of design requirements you can specify for `systune`.

Design Requirement	Description	How To Specify
Reference Tracking	Requirement that a specified output track a specified input	<code>TuningGoal.Tracking</code>
Stability margins	Gain and phase margins, defined as multi-loop disk margins for MIMO systems (see <code>loopmargin</code>)	<code>TuningGoal.Margins</code>
Gain Limits	Limit on the maximum gain and the gain profile across specified I/Os. For example: <ul style="list-style-type: none"> • Disturbance rejection • Custom roll-off specification • Frequency-weighted peak gain limit • Noise amplification limit 	<ul style="list-style-type: none"> • <code>TuningGoal.Gain</code> • <code>TuningGoal.WeightedGain</code> • <code>TuningGoal.Variance</code> • <code>TuningGoal.WeightedVariance</code>

Design Requirement	Description	How To Specify
Custom Loop Shape	Constraints on the open-loop gain profile of a specified SISO or MIMO feedback loop	TuningGoal.LoopShape
Pole Placement	Constraints on location of tunable closed-loop system poles and controller poles	<ul style="list-style-type: none">• TuningGoal.Poles• TuningGoal.StableController

Related Examples

- “Using Design Requirement Objects” on page 6-74
- “Tuning Control Systems with SYSTUNE” on page 6-43

Tune Controller Against Set of Plant Models

`systemtune` can simultaneously tune the parameters of multiple models or control configurations. For example you can:

- Tune a single controller against a range of plant models, to help ensure that the tuned control system is robust against parameter variations
- Tune for reliable control by simultaneously to multiple plant configurations that represent different failure modes of a system

In either case, `systemtune` finds values for tunable parameters that best satisfy the specified tuning objectives for all models.

To tune a controller parameters against a set of plant models:

- 1 Create an array of `genss` models that represent the control systems or configurations to tune against.
- 2 Specify your tuning objectives using `TuningGoal` requirements objects such as `TuningGoal.Tracking`, `TuningGoal.Gain`, or `TuningGoal.Margins`. If you want to limit a tuning requirement to a subset of the models in the model array, set the `Models` property of the `TuningGoal` requirement.

For example, suppose you have a model array whose first entry represents your control system under normal operating conditions, and whose second entry represents the system in a failure mode. Suppose further that `Req` is a `TuningGoal.Tracking` requirement that only applies to the normally operating system. To enforce the requirement only on the first entry, use the following command:

```
Req.Models = [1];
```

- 3 Provide the model array and tuning requirements as input argument to `systemtune`.

`systemtune` jointly tunes the tunable parameters for all models in the array to best meet the tuning requirements as you specify them. `systemtune` returns an array containing the corresponding tuned models. You can use `getBlockValue` or `showBlockValue` to access the tuned values of the control elements.

Related Examples

- “Reliable Control of a Fighter Jet” on page 6-200

Supported Blocks for Tuning in Simulink

sITunable automatically assigns a parameterization to certain Simulink blocks. You can write tuned parameter values back to the Simulink model using `sITunable.writeBlockValue`.

Supported blocks include the following:

Automatically Parametrized Blocks	Simulink Library
Gain	Math Operations
LTI System	Control System Toolbox
Discrete Filter	Discrete
PID Controller (one-degree-of-freedom only)	<ul style="list-style-type: none"> • Continuous • Discrete
State-Space blocks	<ul style="list-style-type: none"> • Continuous • Discrete • Simulink Extras Additional Linear
Zero-pole blocks	<ul style="list-style-type: none"> • Continuous • Discrete
Transfer function blocks	<ul style="list-style-type: none"> • Continuous • Discrete

LTI blocks, state-space blocks, transfer function blocks, and zero-pole blocks discretized using the Model Discretizer are also supported.

Tuning Other Blocks

You can specify tuned blocks in the `sITunable` interface other than the automatically parametrized blocks. `sITunable` assigns a state-space parametrization to such blocks based upon the block linearization. Use

`slTunable.getBlockParam` to examine the parametrization of a block. Use `slTunable.setBlockParam` to override the default parametrization.

For blocks that are not automatically parametrized, you cannot write parameter values back to the block using `slTunable.writeBlockValue`.

Speed Up Tuning with Parallel Computing Toolbox Software

If you have the Parallel Computing Toolbox™ software installed, you can use parallel computing to speed up tuning of fixed-structure control systems. When you run multiple randomized optimization starts, parallel computing speeds up tuning by distributing the optimization runs among MATLAB workers.

To distribute randomized optimization runs among workers:

Start a worker pool of MATLAB sessions using the Parallel Computing Toolbox command `matlabpool`. For example:

```
matlabpool('open')
```

Create a `systuneOptions`, `looptuneOptions`, or `hinfstructOptions` set that specifies multiple random starts. For example, the following options set specifies 20 random restarts to run in parallel for tuning with `looptune`:

```
options = systuneOptions('RandomStart',20,'UseParallel',true);
```

Setting `UseParallel` to `true` enables parallel processing by distributing the randomized starts among available MATLAB workers in the pool.

Use the options set when you call `systune`, `looptune` or `hinfstruct`. For example, if you have already created a tunable control system model, `CL0`, and tunable controller, and tuning requirement vectors `SoftReqs` and `HardReqs`, the following command uses parallel computing to tune the control system of `CL0` with `systune`.

```
[CL,fSoft,gHard,info] = systune(CL0,SoftReq,Hardreq,options);
```

For an example of using parallel computing to tune a Simulink model, see “Using Parallel Computing to Accelerate Tuning” on page 6-105.

To learn more about configuring a worker pool of MATLAB sessions, see the Parallel Computing Toolbox documentation.

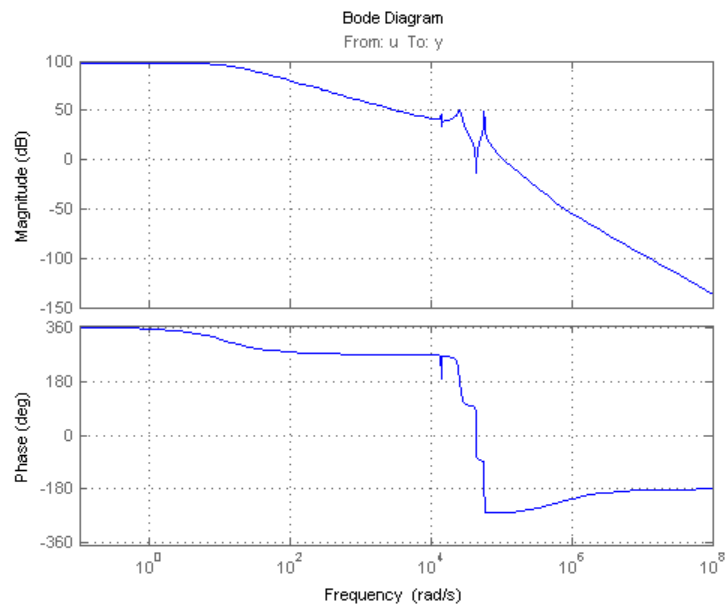
Tuning Control Systems with SYSTUNE

The `systune` command can jointly tune the gains of your control system regardless of its architecture and number of feedback loops. This example outlines the `systune` workflow on a simple application.

Head-Disk Assembly Control

This example uses a 9th-order model of the head-disk assembly (HDA) in a hard-disk drive. This model captures the first few flexible modes in the HDA.

```
load rctExamples G
bode(G), grid
```



We use the feedback loop shown below to position the head on the correct track. This control structure consists of a PI controller and a low-pass filter in the return path. The head position y should track a step change r with a response time of about one millisecond, little or no overshoot, and no steady-state error.

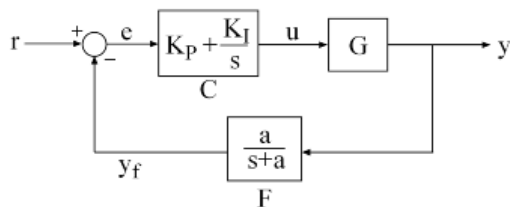


Figure 1: Control Structure

You can use `systemtune` to directly tune the PI gains and filter coefficient a subject to a variety of time- and frequency-domain requirements.

Specifying the Tunable Elements

There are two tunable elements in the control structure of Figure 1: the PI controller $C(s)$ and the low-pass filter

$$F(s) = \frac{a}{s+a}.$$

You can use the `ltiblock.pid` class to parameterize the PI block:

```
C0 = ltiblock.pid('C','pi'); % tunable PI
```

To parameterize the lowpass filter $F(s)$, create a tunable real parameter a and construct a first-order transfer function with numerator a and denominator $s + a$:

```
a = realp('a',1); % filter coefficient
F0 = tf(a,[1 a]); % filter parameterized by a
```

See the *"Building Tunable Models"* example for an overview of available tunable elements.

Building a Tunable Closed-Loop Model

Next build a closed-loop model of the feedback loop in Figure 1. To facilitate open-loop analysis and specify open-loop requirements such as desired stability margins, add a loop opening switch at the plant input u :

```
LSU = loopswitch('u');
```

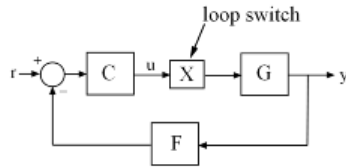


Figure 2: Loop Switch Block

Use feedback to build a model of the closed-loop transfer from reference r to head position y :

```
T0 = feedback(G*LSU*C0,F0); % closed-loop transfer from r to y
T0.InputName = 'r';
T0.OutputName = 'y';
```

The result $T0$ is a generalized state-space model (genss) that depends on the tunable elements C and F .

Specifying the Design Requirements

The TuningGoal package contains a variety of control design requirements for specifying the desired behavior of the control system. These include requirements on the response time, deterministic and stochastic gains, loop shape, stability margins, and pole locations. Here we use two requirements to capture the control objectives:

- **Tracking requirement** : The position y should track the reference r with a 1 millisecond response time
- **Stability margin requirement** : The feedback loop should have 6dB of gain margin and 45 degrees of phase margin

Use the `TuningGoal.Tracking` and `TuningGoal.Margins` objects to capture these requirements. Note that the margins requirement applies to the open-loop response measured at the plant input `u` (location marked by the `loopswitch` block `LSU`).

```
Req1 = TuningGoal.Tracking('r','y',0.001);
Req2 = TuningGoal.Margins('u',6,45);
```

Tuning the Controller Parameters

You can now use `sysstune` to tune the PI gain and filter coefficient α . This function takes the tunable closed-loop model `T0` and the requirements `Req1,Req2`. Use a few randomized starting points to improve the chances of getting a globally optimal design.

```
rng('default')
Options = sysstuneOptions('RandomStart',3);
[T,fSoft,~,Info] = sysstune(T0,[Req1,Req2],Options);
```

```
Final: Soft = 1.35, Hard = -Inf, Iterations = 157
Final: Soft = 2.61, Hard = -Inf, Iterations = 104
Final: Soft = 2.78e+03, Hard = -Inf, Iterations = 180
      Min decay rate 1e-07 is close to lower bound 1e-07
Final: Soft = 1.35, Hard = -Inf, Iterations = 51
```

All requirements are normalized so a requirement is satisfied when its value is less than 1. Here the final value is slightly greater than 1, indicating that the requirements are nearly satisfied. Use the output `fSoft` to see the tuned value of each requirement. Here we see that the first requirement (tracking) is slightly violated while the second requirement (margins) is satisfied.

`fSoft`

```
fSoft =
      1.3461      0.6326
```

The first output `T` of `systune` is the "tuned" closed-loop model. Use `showTunable` or `getBlockValue` to access the tuned values of the PI gains and filter coefficient:

```
getBlockValue(T,'C') % tuned value of PI controller
```

```
ans =
```

$$K_p + K_i * \frac{1}{s}$$

```
with Kp = 0.00104, Ki = 0.0122
```

```
Name: C
```

```
Continuous-time PI controller in parallel form.
```

```
showTunable(T) % tuned values of all tunable elements
```

```
C =
```

$$K_p + K_i * \frac{1}{s}$$

```
with Kp = 0.00104, Ki = 0.0122
```

```
Name: C
```

```
Continuous-time PI controller in parallel form.
```

```
-----
```

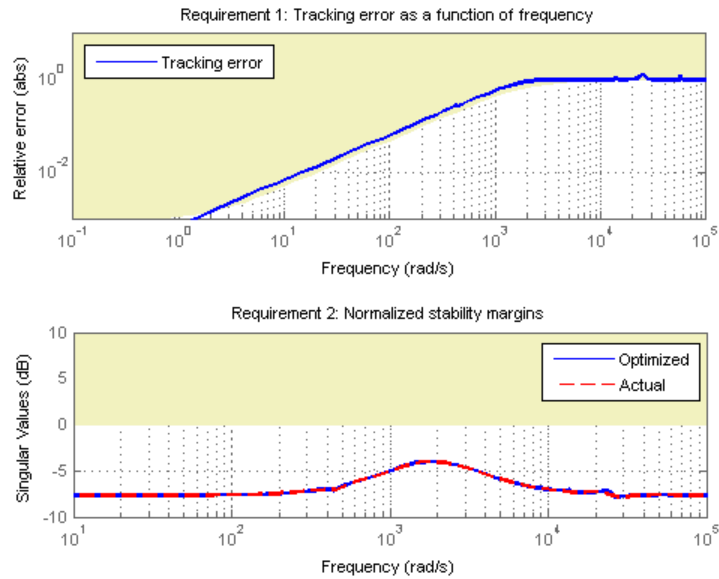
```
a = 3.19e+03
```

Validating Results

First use `viewSpec` to inspect how the tuned system does against each requirement. The first plot shows the tracking error as a function of frequency, and the second plot shows the normalized disk margins as a function of

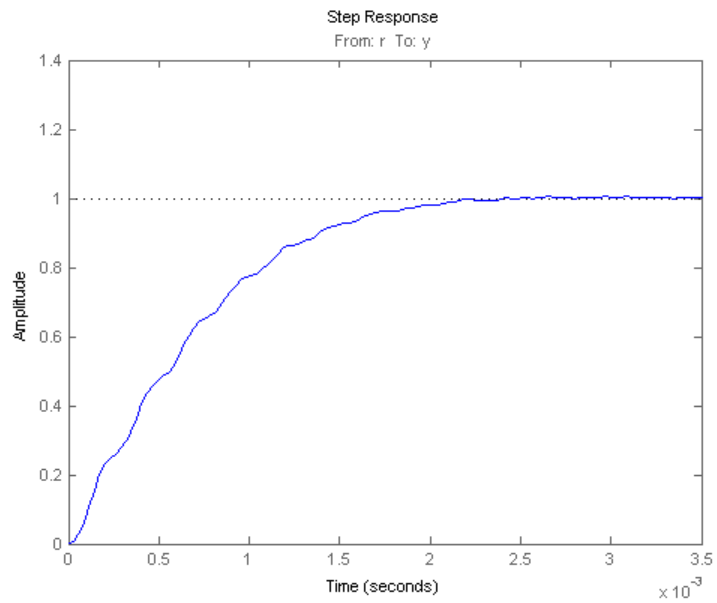
frequency (see `loopmargin`). See the *"Creating Design Requirements"* example for details.

```
clf, viewSpec([Req1 Req2],T,Info)
```



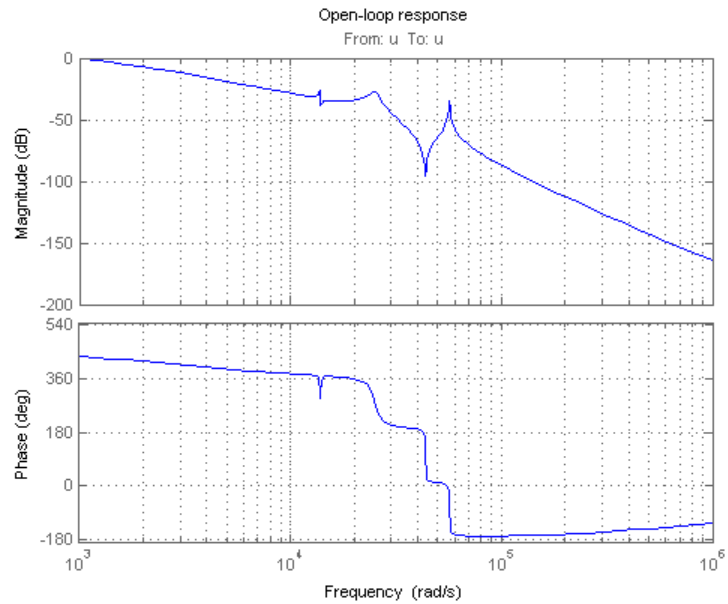
Next plot the closed-loop step response from reference r to head position y . The response has no overshoot but wobbles a little.

```
clf, step(T)
```

To investigate further, use `getLoopTransfer` to get the open-loop response at the plant input.

```
L = getLoopTransfer(T, 'u');  
bode(L, {1e3, 1e6}), grid  
title('Open-loop response')
```



The wobble is due to the first resonance after the gain crossover. To eliminate it, you could add a notch filter to the feedback loop and tune its coefficients along with the lowpass coefficient and PI gains using systune.

Tuning Feedback Loops with LOOPTUNE

This example shows the basic workflow of tuning feedback loops with the `looptune` command. `looptune` is similar to `systemtune` and meant to facilitate loop shaping design by automatically generating the tuning requirements.

Engine Speed Control

This example uses a simple engine speed control application as illustration. The control system consists of a single PID loop and the PID controller gains must be tuned to adequately respond to step changes in the desired speed. Specifically, we want the response to settle in less than 5 seconds with little or no overshoot.

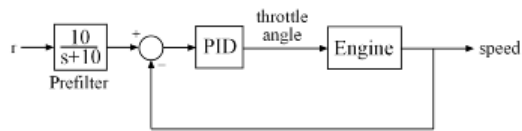
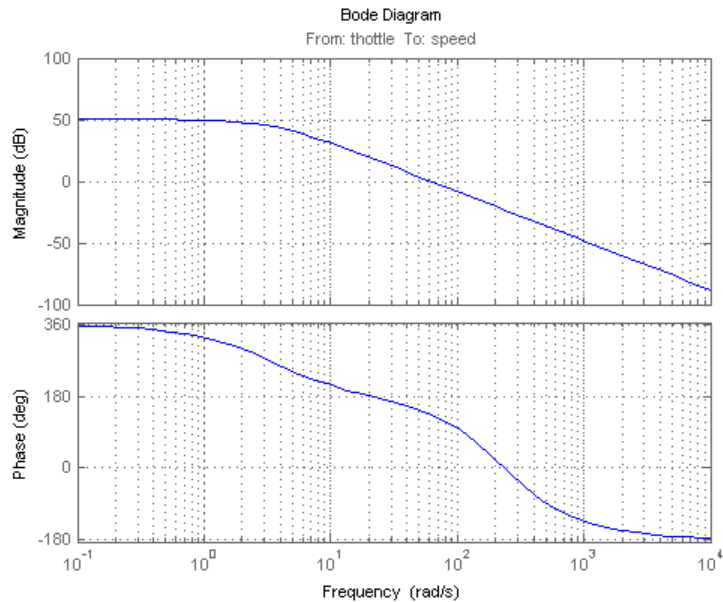


Figure 1: Engine Speed Control Loop

We use the following fourth-order model of the engine dynamics.

```
load rctExamples Engine
bode(Engine), grid
```



Specifying the Tunable Elements

We need to tune the four PID gains to achieve the desired performance. Use the `ltiblock.pid` class to parameterize the PID controller.

```
PID0 = ltiblock.pid('SpeedController','pid')
```

```
PID0 =
```

```
Parametric continuous-time PID controller "SpeedController" with formula:
```

$$K_p + K_i * \frac{1}{s} + K_d * \frac{s}{T_f * s + 1}$$

```
and tunable parameters Kp, Ki, Kd, Tf.
```

```
Type "pid(PID0)" to see the current value and "get(PID0)" to see all proper
```

Building a Tunable Model of the Feedback Loop

looptune tunes the generic SISO or MIMO feedback loop of Figure 2. This feedback loop models the interaction between the plant and the controller. Note that this is a *positive* feedback interconnection.

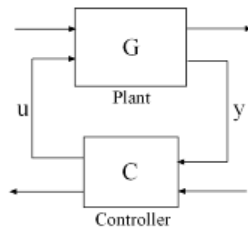


Figure 2: Generic Feedback Loop

For the speed control loop, the plant G is the engine model and the controller C consists of the PID and the prefilter F .

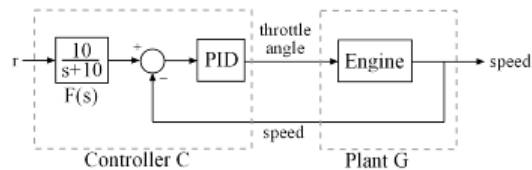


Figure 3: Feedback Loop for Engine Speed Control

To use looptune, create models for G and C in Figure 3. Assign names to the inputs and outputs of each model to specify the feedback paths between plant and controller. Note that the controller C has two inputs: the speed reference "r" and the speed measurement "speed".

```
F = tf(10,[1 10]); % prefilter
```

```
G = Engine;
G.InputName = 'throttle';
G.OutputName = 'speed';

CO = PID0 * [F , -1];
CO.InputName = {'r','speed'};
CO.OutputName = 'throttle';
```

Here CO is a generalized state-space model (genss) that depends on the tunable PID block PID0.

Tuning the Controller Parameters

You can now use `looptune` to tune the PID gains subject to a simple control bandwidth requirement. To achieve the 5-second settling time, the gain crossover frequency of the open-loop response should be approximately 1 rad/s. Given this basic requirement, `looptune` automatically shapes the open-loop response to provide integral action, high-frequency roll-off, and adequate stability margins. Note that you could specify additional requirements to further constrain the design, see *"Decoupling Controller for a Distillation Column"* for an example.

```
wc = 1; % target gain crossover frequency
```

```
[~,C,~,Info] = looptune(G,CO,wc);
```

```
Final: Peak gain = 0.92, Iterations = 3
```

The final value is less than 1, indicating that the desired bandwidth was achieved with adequate roll-off and stability margins. `looptune` returns the tuned controller C. Use `getBlockValue` to retrieve the tuned value of the PID block.

```
PIDT = getBlockValue(C,'SpeedController')
```

```
PIDT =
```

$$K_p + K_i * \frac{1}{s} + K_d * s$$

$$s \quad T_f s + 1$$

with $K_p = 0.000484$, $K_i = 0.00324$, $K_d = 0.000835$, $T_f = 1$

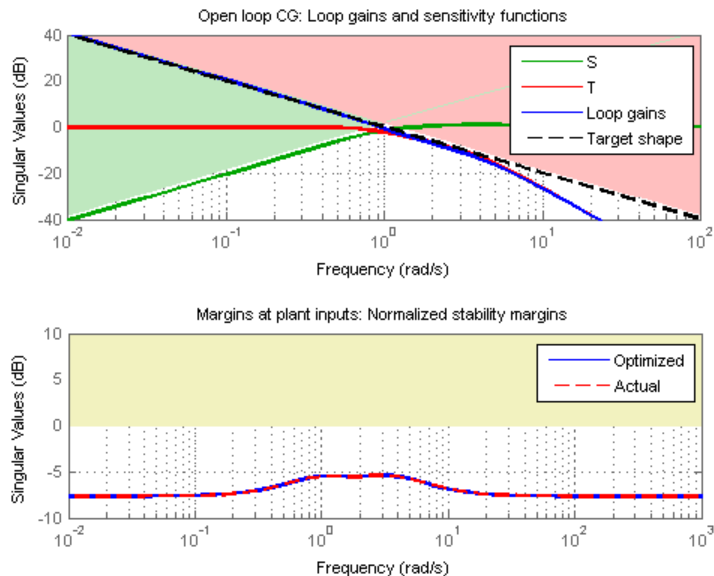
Name: SpeedController

Continuous-time PIDF controller in parallel form.

Validating Results

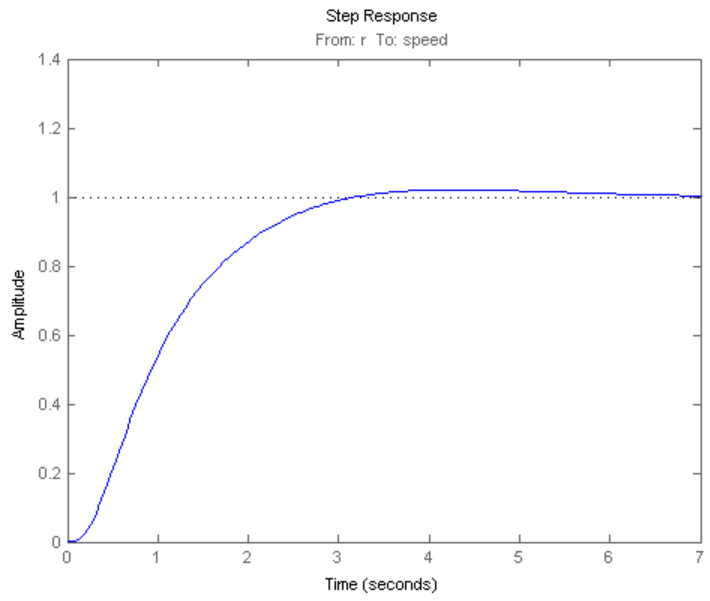
Use `loopview` to validate the design and visualize the loop shaping requirements implicitly enforced by `looptune`.

```
clf, loopview(G,C,Info)
```



Next plot the closed-loop response to a step command in engine speed. The tuned response satisfies our requirements.

```
T = connect(G,C,'r','speed'); % closed-loop transfer from r to speed
clf, step(T)
```



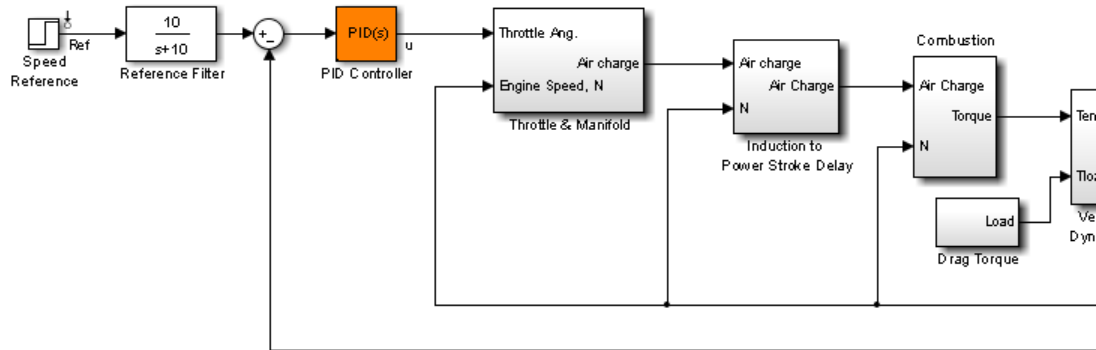
Tuning Control Systems in Simulink

This example shows how to use `systemtune` or `looptune` to automatically tune control systems modeled in Simulink.

Engine Speed Control

For this example we use the following model of an engine speed control system:

```
open_system('rct_engine_speed')
```



The control system consists of a single PID loop and the PID controller gains must be tuned to adequately respond to step changes in the desired speed. Specifically, we want the response to settle in less than 5 seconds with little or no overshoot. While `pidtune` is a faster alternative for tuning a single PID controller, this simple example is well suited for an introduction to the `systemtune` and `looptune` workflows in Simulink.

Controller Tuning with SYSTUNE

The `sITunable` interface provides a convenient gateway to `systemtune` for control systems modeled in Simulink. This interface lets you specify which blocks in the Simulink model are tunable and what signals are of interest for open- or closed-loop validation. Create an `sITunable` instance for the `rct_engine_speed` model and list the "PID Controller" block (highlighted in

orange) as tunable. Note that all linearization points in the model (signals "Ref" and "Speed" here) are automatically available as analysis I/Os.

```
ST0 = s1Tunable('rct_engine_speed', 'PID Controller');
```

The PID block is initialized with its value in the Simulink model, which you can access using `getBlockValue`. Note that the proportional and derivative gains are initialized to zero.

```
ST0.getBlockValue('PID Controller')
```

```
ans =
```

$$Ki * \frac{1}{s}$$

```
with Ki = 0.01
```

```
Name: PID_Controller
Continuous-time I-only controller.
```

Next create a reference tracking requirement to capture the target settling time. Use the signal names in the Simulink model to refer to the reference and output signals, and use a two-second response time target to ensure settling in less than 5 seconds.

```
TrackReq = TuningGoal.Tracking('Ref', 'Speed', 2);
```

You can now tune the control system ST0 subject to the requirement TrackReq.

```
ST1 = systune(ST0, TrackReq);
```

```
Final: Soft = 1.07, Hard = -Inf, Iterations = 99
```

The final value is close to 1 indicating that the tracking requirement is met. `systune` returns a "tuned" version ST1 of the control system described by ST0. Again use `getBlockValue` to access the tuned values of the PID gains:

```
ST1.getBlockValue('PID Controller')
```

```
ans =
```

$$K_p + K_i * \frac{1}{s} + K_d * \frac{s}{T_f * s + 1}$$

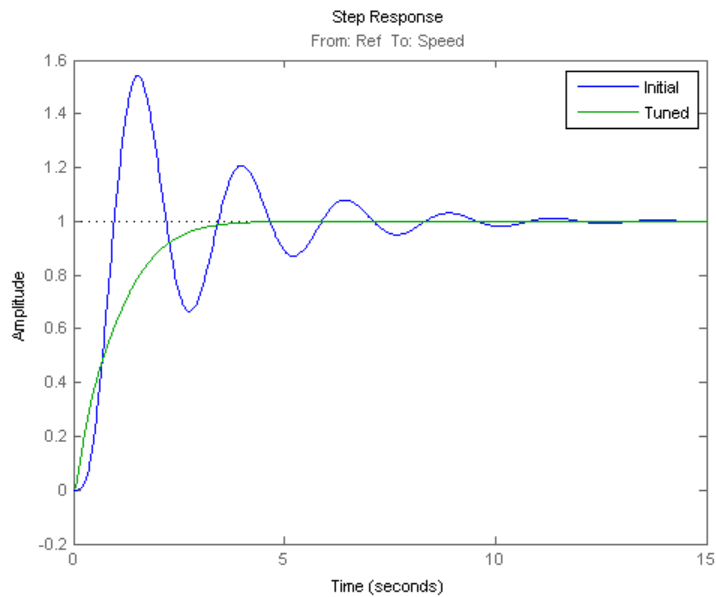
```
with Kp = 0.00189, Ki = 0.00357, Kd = 0.000473, Tf = 0.00176
```

```
Name: PID_Controller
```

```
Continuous-time PIDF controller in parallel form.
```

To simulate the closed-loop response to a step command in speed, get the initial and tuned transfer functions from speed command "Ref" to "Speed" output and plot their step responses:

```
T0 = ST0.getIOTransfer('Ref','Speed');
T1 = ST1.getIOTransfer('Ref','Speed');
step(T0,T1)
legend('Initial','Tuned')
```



Controller Tuning with LOOPTUNE

You can also use `looptune` to tune control systems modeled in Simulink. The `looptune` workflow is very similar to the `systemtune` workflow. One difference is that `looptune` needs to know the boundary between the plant and controller, which is specified in terms of *controls* and *measurements* signals.

```
STO = slTunable('rct_engine_speed', 'PID Controller');

% Specify plant/controller boundary
STO.addControl('u');
STO.addMeasurement('Speed');

% Add the "Ref" signal to the list of analysis I/Os
STO.addIO('Ref', 'in')
```

For a single loop the performance is essentially captured by the response time, or equivalently by the open-loop crossover frequency. Based on first-order characteristics the crossover frequency should exceed 1 rad/s for

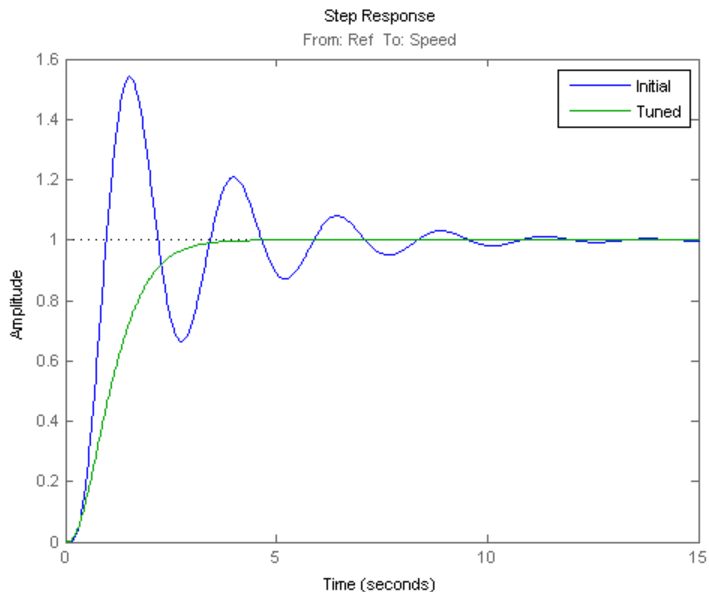
the closed-loop response to settle in less than 5 seconds. You can therefore tune the PID loop using 1 rad/s as target 0-dB crossover frequency.

```
wc = 1;  
ST1 = ST0.looptune(wc);
```

```
Final: Peak gain = 0.961, Iterations = 10
```

Again the final value is close to 1, indicating that the target control bandwidth was achieved. As with `systeme`, use `getIOTransfer` to compute and plot the closed-loop response from speed command to actual speed. The result is very similar to that obtained with `systeme`.

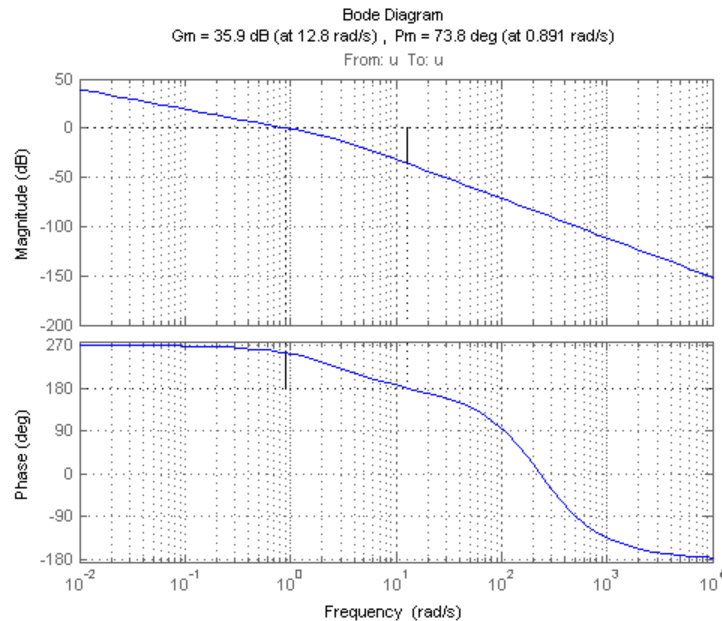
```
T0 = ST0.getIOTransfer('Ref','Speed');  
T1 = ST1.getIOTransfer('Ref','Speed');  
step(T0,T1)  
legend('Initial','Tuned')
```



You can also perform open-loop analysis, for example, compute the gain and phase margins at the plant input u .

```
% Note: -1 because |margin| expects the negative-feedback loop transfer
L = ST1.getLoopTransfer('u',-1);
```

```
margin(L), grid
```

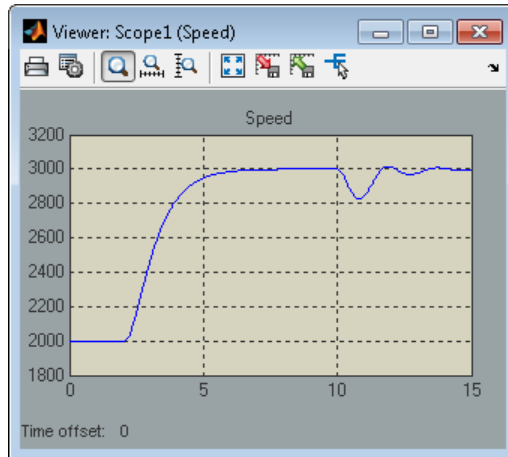


Validation in Simulink

Once you are satisfied with the `sys tune` or `looptune` results, you can upload the tuned controller parameters to Simulink for further validation with the nonlinear model.

```
ST1.writeBlockValue()
```

You can now simulate the engine response with the tuned PID controller.



The nonlinear simulation results closely match the linear responses obtained in MATLAB.

Comparison of PI and PID Controllers

Closer inspection of the tuned PID gains suggests that the derivative term contributes little because of the large value of the Tf coefficient.

```
ST1.showTunable()
```

```
Block "rct_engine_speed/PID Controller" =
```

```
Value =
```

$$K_p + K_i * \frac{1}{s} + K_d * \frac{s}{T_f s + 1}$$

```
with Kp = 0.000653, Ki = 0.00282, Kd = 0.0021, Tf = 46.7
```

```
Name: PID_Controller
```

```
Continuous-time PIDF controller in parallel form.
```

This suggests using a simpler PI controller instead. To do this, you need to override the default parameterization for the "PID Controller" block:

```
ST0.setBlockParam('PID Controller',ltiblock.pid('C','pi'))
```

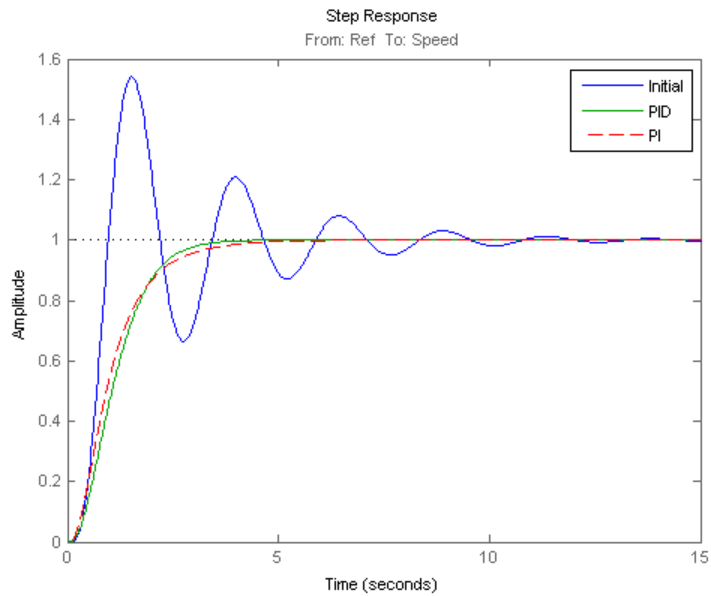
This specifies that the "PID Controller" block should now be parameterized as a mere PI controller. Next re-tune the control system for this simpler controller:

```
ST2 = ST0.looptune(wc);
```

```
Final: Peak gain = 0.915, Iterations = 5
```

Again the final value is less than one indicating success. Compare the closed-loop response with the previous ones:

```
T2 = ST2.getIOTransfer('Ref','Speed');
step(T0,T1,T2,'r--')
legend('Initial','PID','PI')
```



Clearly a PI controller is sufficient for this application.

Building Tunable Models

This example shows how to create tunable models of control systems for use with `system` or `looptune`.

Background

You can tune the gains and parameters of your control system with `system` or `looptune`. To use these commands, you need to construct a tunable model of the control system that identifies and parameterizes its tunable elements. This is done by combining numeric LTI models of the fixed elements with parametric models of the tunable elements.

Using Pre-Defined Tunable Elements

You can use one of the following "parametric" blocks to model commonly encountered tunable elements:

- **ltiblock.gain**: Tunable gain
- **ltiblock.pid**: Tunable PID controller
- **ltiblock.pid2**: Tunable two-degree-of-freedom PID controller
- **ltiblock.tf**: Tunable transfer function
- **ltiblock.ss**: Tunable state-space model.

For example, create a tunable model of the feedforward/feedback configuration of Figure 1 where C is a tunable PID controller and F is a tunable first-order transfer function.

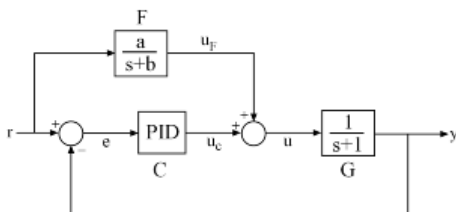


Figure 1: Control System with Feedforward and Feedback Paths

First model each block in the block diagram, using suitable parametric blocks for C and F .

```
G = tf(1,[1 1]);
C = ltiblock.pid('C','pid'); % tunable PID block
F = ltiblock.tf('F',0,1);    % tunable first-order transfer function
```

Then use `connect` to build a model of the overall block diagram. To specify how the blocks are connected, label the inputs and outputs of each block and model the summing junctions using `sumblk`.

```
G.u = 'u'; G.y = 'y';
C.u = 'e'; C.y = 'uC';
F.u = 'r'; F.y = 'uF';

% Summing junctions
S1 = sumblk('e = r-y');
S2 = sumblk('u = uF + uC');

T = connect(G,C,F,S1,S2,'r','y')
```

T =

```
Generalized continuous-time state-space model with 1 outputs, 1 inputs, 3
C: Parametric PID controller, 1 occurrences.
F: Parametric SISO transfer function, 0 zeros, 1 poles, 1 occurrences.
```

Type "ss(T)" to see the current value, "get(T)" to see all properties, and

This creates a generalized state-space model T of the closed-loop transfer function from r to y . This model depends on the tunable blocks C and F and you can use `systemtune` to automatically tune the PID gains and feedforward coefficients a, b subject to your performance requirements. Use `showTunable` to see the current value of the tunable blocks.

```
showTunable(T)
```

```

C =

      1
Ki * ---
      s

with Ki = 0.001

Name: C
Continuous-time I-only controller.
-----
F =

      10
-----
      s + 10

Name: F
Continuous-time transfer function.

```

Interacting with the Tunable Parameters

You can adjust the parameterization of the tunable elements *C* and *F* by interacting with the objects C and F. Use get to see their list of properties.

```

get(C)

      Kp: [1x1 param.Continuous]
      Ki: [1x1 param.Continuous]
      Kd: [1x1 param.Continuous]
      Tf: [1x1 param.Continuous]
IFormula: ''
DFormula: ''
      Ts: 0
TimeUnit: 'seconds'
InputName: {'e'}
InputUnit: {''}
InputGroup: [1x1 struct]
OutputName: {'uC'}

```

```

OutputUnit: {''}
OutputGroup: [1x1 struct]
    Name: 'C'
    Notes: {}
    UserData: []

```

A PID controller has four tunable parameters K_p , K_i , K_d , T_f , and C contains a description of each of these parameters. Parameter attributes include current value, minimum and maximum values, and whether the parameter is free or fixed.

C.Kp

```

ans =

    Name: 'Kp'
    Value: 0
    Minimum: -Inf
    Maximum: Inf
    Free: 1
    Scale: 1
    Info: [1x1 struct]

```

1x1 param.Continuous

Set the corresponding attributes to override defaults. For example, you can fix the time constant T_f to the value 0.1 by

```

C.Tf.Value = 0.1;
C.Tf.Free = false;

```

Creating Custom Tunable Elements

For tunable elements not covered by the pre-defined blocks listed above, you can create your own parameterization in terms of elementary real parameters (realp). Consider the low-pass filter

$$F(s) = \frac{a}{s + a}$$

where the coefficient a is tunable. To model this tunable element, create a real parameter a and define F as a transfer function whose numerator and denominator are functions of a . This creates a generalized state-space model F of the low-pass filter parameterized by the tunable scalar a .

```
a = realp('a',1); % real tunable parameter, initial value 1
F = tf(a,[1 a])
```

$F =$

```
Generalized continuous-time state-space model with 1 outputs, 1 inputs, 1
a: Scalar parameter, 2 occurrences.
```

Type "ss(F)" to see the current value, "get(F)" to see all properties, and

Similarly, you can use real parameters to model the notch filter

$$N(s) = \frac{s^2 + 2\zeta_1\omega_n s + \omega_n^2}{s^2 + 2\zeta_2\omega_n s + \omega_n^2}$$

with tunable coefficients $\omega_n, \zeta_1, \zeta_2$.

```
wn = realp('wn',100);
zeta1 = realp('zeta1',1); zeta1.Maximum = 1; % zeta1 <= 1
zeta2 = realp('zeta2',1); zeta2.Maximum = 1; % zeta2 <= 1
N = tf([1 2*zeta1*wn wn^2],[1 2*zeta2*wn wn^2]); % tunable notch filter
```

You can also create tunable elements with matrix-valued parameters. For example, model the observer-based controller $C(s)$ with equations

$$\frac{dx}{dt} = Ax + Bu + L(y - Cx), \quad u = -Kx$$

and tunable gain matrices K and L .

```
% Plant with 6 states, 2 controls, 3 measurements
[A,B,C] = ssdata(rss(6,3,2));
```

```
K = realp('K',zeros(2,6));
L = realp('L',zeros(6,3));
```

```
C = ss(A-B*K-L*C,L,-K,0)
```

```
C =
```

```
Generalized continuous-time state-space model with 2 outputs, 3 inputs, 6
```

```
  K: Parametric 2x6 matrix, 2 occurrences.
```

```
  L: Parametric 6x3 matrix, 2 occurrences.
```

```
Type "ss(C)" to see the current value, "get(C)" to see all properties, and
```

Enabling Open-Loop Requirements

The `sys tune` command takes a closed-loop model of the overall control system, like the tunable model `T` built at the beginning of this example. Such models do not readily support open-loop analysis or open-loop specifications such as loop shapes and stability margins. To gain access to open-loop responses, insert a `loopswitch` block as shown in Figure 2.

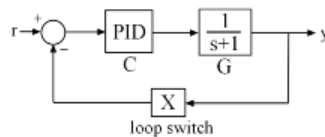


Figure 2: Loop Switch Block

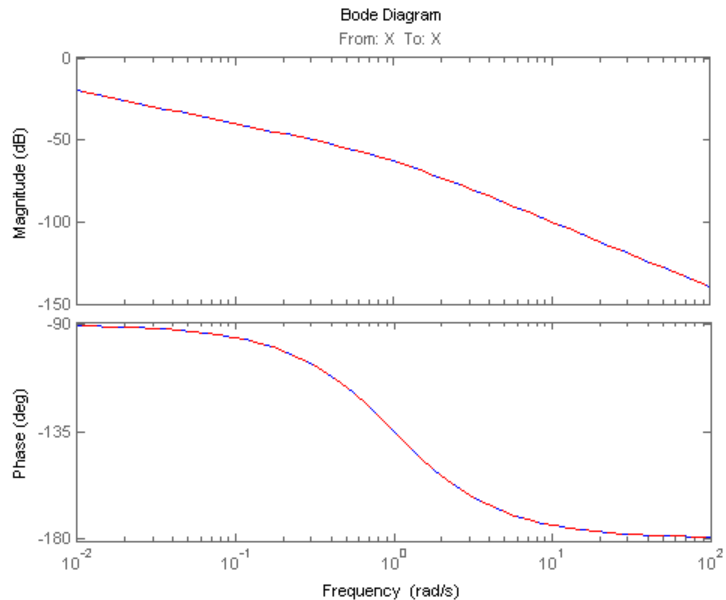
The `loopswitch` block is an open/closed switch that can be used to open feedback loops or to measure open-loop responses. For example, construct a closed-loop model `T` of the feedback loop of Figure 2 where `C` is a tunable PID.

```
G = tf(1,[1 1]);
```

```
C = ltiblock.pid('C','pid');
LS = loopswitch('X');
T = feedback(G*C,LS);
```

By default the loop switch "X" is closed and T models the closed-loop transfer from r to y . However, you can now use `getLoopTransfer` to compute the (negative-feedback) loop transfer function measured at the location "X". Note that this loop transfer function is $L = GC$ for the feedback loop of Figure 2.

```
L = getLoopTransfer(T,'X',-1); % loop transfer at "X"
clf, bode(L,'b',G*C,'r--')
```



You can also refer to the location "X" when specifying target loop shapes or stability margins for `systune`. The requirement then applies to the loop transfer measured at this location.

```
% Target loop shape for loop transfer at "X"
Req1 = TuningGoal.LoopShape('X',tf(5,[1 0]));

% Target stability margins for loop transfer at "X"
```



```
Req2 = TuningGoal.Margins('X',6,40);
```

In general, the loop identifiers (names of the loop opening sites marked by a `loopswitch` block) are specified in the `LoopID` property of `loopswitch` blocks. For single-loop switches, the loop identifier defaults to the block name. For multi-loop switches, the default identifiers are the block name followed by an index.

```
LS = loopswitch('Y',2); % two-channel switch
LS.LoopID
```

```
ans =
```

```
 'Y(1)'  
 'Y(2)'
```

You can override these default names by modifying the `LoopID` property.

```
% Relabel feedback channels to "Inner" and "Outer".
LS.LoopID = {'Inner' ; 'Outer'};
LS.LoopID
```

```
ans =
```

```
 'Inner'  
 'Outer'
```

Using Design Requirement Objects

This example gives a tour of available design requirements for control system tuning with `systune` or `looptune`.

Background

The `systune` and `looptune` commands let you tune the parameters of fixed-structure control systems subject to a variety of time- and frequency-domain requirements. The `TuningGoal` package is the repository for such design requirements.

Tracking Requirement

The `TuningGoal.Tracking` requirement enforces setpoint tracking and loop decoupling objectives. For example

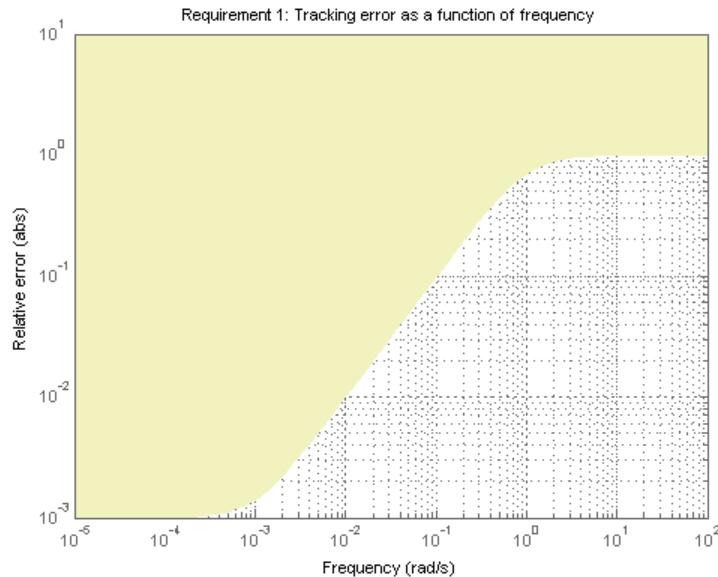
```
R1 = TuningGoal.Tracking('r','y',2);
```

specifies that the output `y` should track the reference `r` with a two-second response time. Similarly

```
R2 = TuningGoal.Tracking({'Vsp','wsp'},{'V','w'},2);
```

specifies that `V` should track `Vsp` and `w` should track `wsp` with minimum cross-coupling between the two responses. Tracking requirements are converted into frequency-domain constraints on the tracking error as a function of frequency. For the first requirement `R1`, for example, the gain from `r` to the tracking error $e = r - y$ should be small at low frequency and approach 1 (100%) at frequencies greater than 1 rad/s (bandwidth for a two-second response time). You can use `viewSpec` to visualize this frequency-domain constraint.

```
viewSpec(R1)
```



Note that the yellow region indicates where the requirement is violated.

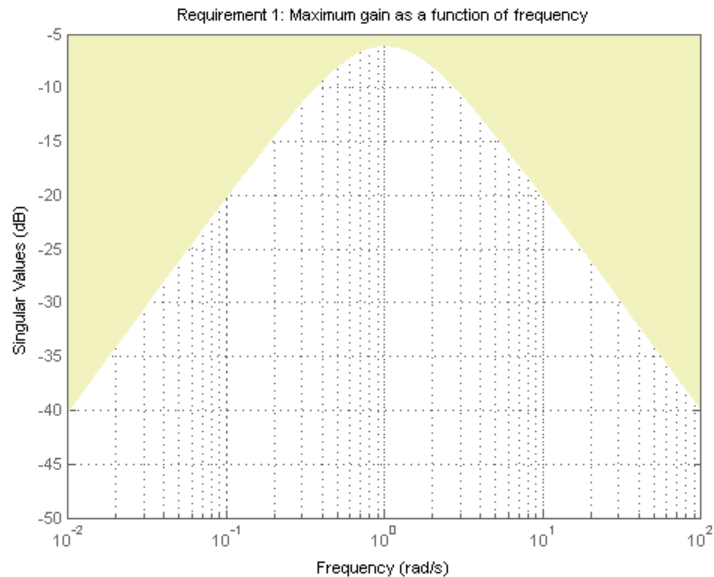
Gain Requirement

The `TuningGoal.Gain` requirement enforces gain limits on SISO or MIMO closed-loop transfer functions. This requirement is useful to enforce adequate disturbance rejection and roll off, limit sensitivity and control effort, and prevent saturation. For MIMO transfer functions, "gain" refers to the largest singular value of the frequency response matrix. The gain limit can be frequency dependent. For example

```
s = tf('s');
R1 = TuningGoal.Gain('d','y',s/(s+1)^2);
```

specifies that the gain from `d` to `y` should not exceed the magnitude of the transfer function $s/(s+1)^2$.

```
viewSpec(R1)
```

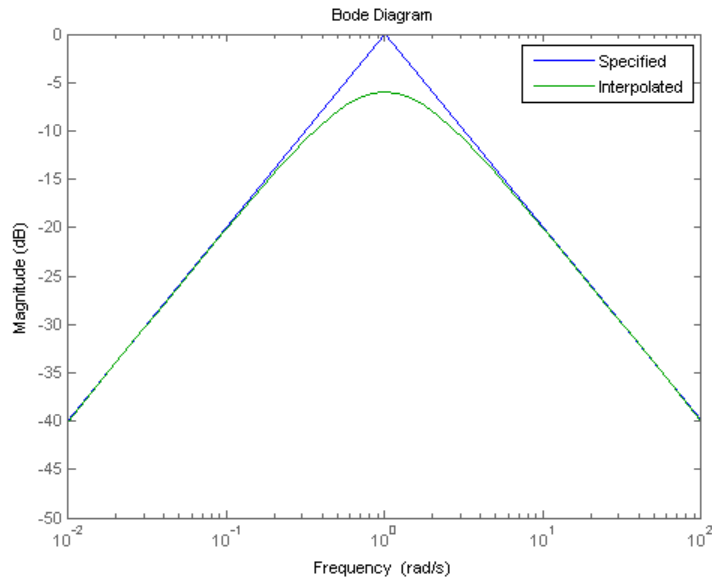


It is often convenient to just sketch the asymptotes of the desired gain profile. For example, instead of the transfer function $s/(s+1)^2$, we could just specify gain values of 0.01, 1, 0.01 at the frequencies 0.01, 1, 100, the point (1, 1) being the breakpoint of the two asymptotes s and $1/s$.

```
Asymptotes = frd([0.01, 1, 0.01], [0.01, 1, 100]);
R2 = TuningGoal.Gain('d', 'y', Asymptotes);
```

The requirement object automatically turns this discrete gain profile into a gain limit defined at all frequencies.

```
bodemag(Asymptotes, R2.MaxGain)
legend('Specified', 'Interpolated')
```



Variance Requirement

The `TuningGoal.Variance` requirement limits the noise variance amplification from specified inputs to specified outputs. In technical terms, this requirement constrains the H_2 norm of a closed-loop transfer function. This requirement is preferable to `TuningGoal.Gain` when the input signals are random processes and the average gain matters more than the peak gain. For example,

```
R = TuningGoal.Variance('n','y',0.1);
```

limits the output variance of `y` to 0.1^2 for a unit-variance white-noise input `n`.

Weighted Gain and Weighted Variance Requirements

The `TuningGoal.WeightedGain` and `TuningGoal.WeightedVariance` requirements are generalizations of the `TuningGoal.Gain` and `TuningGoal.Variance` requirements. These requirements constrain the H_∞ or H_2 norm of a frequency-weighted closed-loop transfer function

$W_L(s)T(s)W_R(s)$, where $W_L(s)$ and $W_R(s)$ are user-defined weighting functions. For example

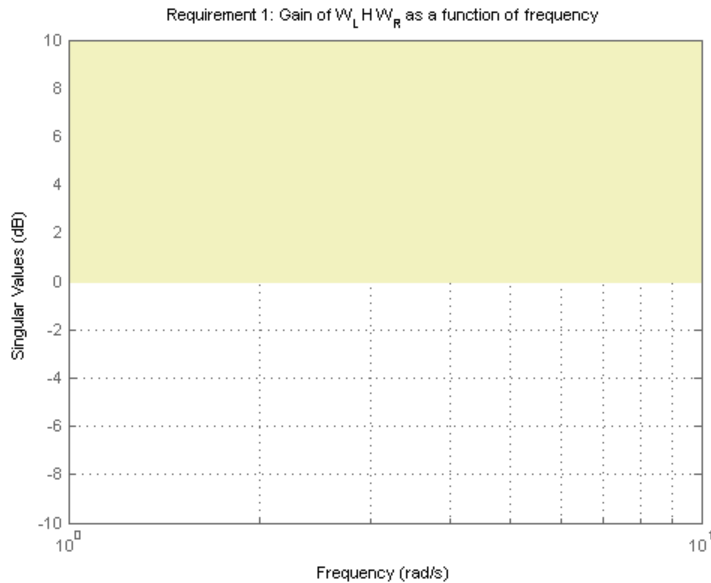
```
WL = blkdiag(1/(s+0.001),s/(0.001*s+1));
WR = [];
R = TuningGoal.WeightedGain('r',{ 'e','y'},WL,[]);
```

specifies the constraint

$$\left\| \begin{pmatrix} \frac{1}{s+0.001} T_{re} \\ \frac{s}{0.001s+1} T_{ry} \end{pmatrix} \right\|_{\infty} < 1$$

Note that this is a normalized gain constraint (unit bound across frequency).

```
viewSpec(R)
```



The `TuningGoal.WeightedVariance` requirement is useful to specify LQG-like performance objectives. For example, the LQG cost

$$J = \lim_{T \rightarrow \infty} E \left(\frac{1}{T} \int_0^T (x^T Q x + u^T R u) dt \right)$$

can be expressed as

$$J = \left\| \begin{pmatrix} W_z & 0 \\ 0 & W_u \end{pmatrix} T(s) \right\|_2^2$$

where $T(s)$ is the closed-loop transfer from process and measurement noise to the states x and controls u , and W_z, W_u are Cholesky factors of Q and R (see `chol`).

Loop Shape Requirement

The `TuningGoal.LoopShape` requirement is used to shape the open-loop response gain(s), a design approach known as *loop shaping*. For example,

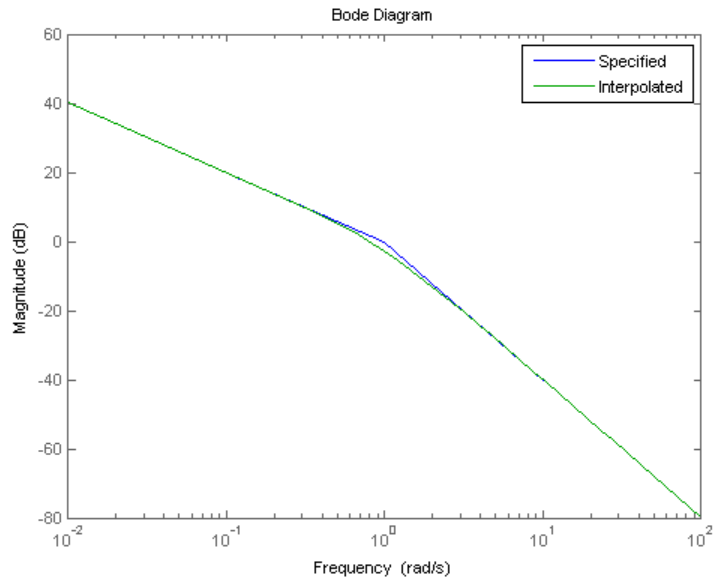
```
R1 = TuningGoal.LoopShape('u', 1/s);
```

specifies that the open-loop response measured at the location "u" should look like a pure integrator (as far as its gain is concerned). In MATLAB, the location "u" should be marked using a `loopswitch` block, see the *"Building Tunable Models"* example for details. In Simulink, this location should be registered as control, measurement, or switch using the `addControl`, `addMeasurement`, `addSwitch` methods of the `sITunable` tuning interface.

As with other gain specifications, you can just specify the asymptotes of the desired loop shape using a few frequency points. For example, to specify a loop shape with gain crossover at 1 rad/s, -20 dB/decade slope before 1 rad/s, and -40 dB/decade slope after 1 rad/s, just specify that the gain at the frequencies 0.1, 1, 10 should be 10, 1, 0.01, respectively.

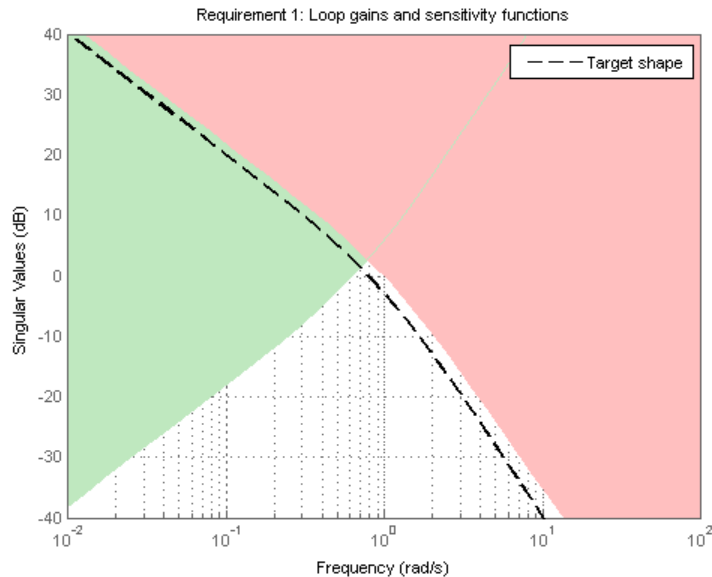
```
LS = frd([10, 1, 0.01], [0.1, 1, 10]);
R2 = TuningGoal.LoopShape('u', LS);
```

```
bodemag(LS, R2.LoopGain)
legend('Specified', 'Interpolated')
```



Loop shape requirements are constraints on the open-loop response L . For tuning purposes, they are converted into closed-loop gain constraints on the sensitivity function $S = 1/(1 + L)$ and complementary sensitivity function $T = L/(1 + L)$. Use `viewSpec` to visualize the target loop shape and corresponding gain bounds on S (green) and T (red).

```
viewSpec(R2)
```

Stability Margins Requirement

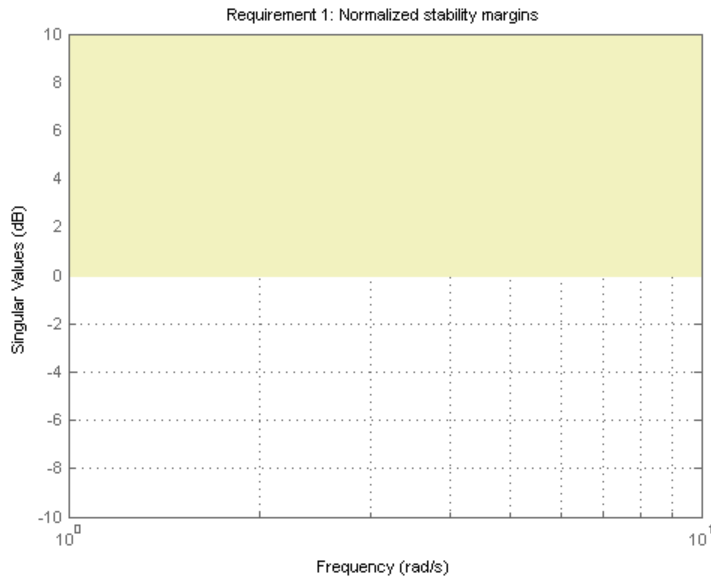
The `TuningGoal.Margins` requirement enforces minimum amounts of gain and phase margins at the specified loop opening site(s). For MIMO feedback loops, this requirement uses the notion of *disk margins*, which guarantee stability for concurrent gain and phase variations of the specified amount in all feedback channels (see `loopmargin` for details). For example,

```
R = TuningGoal.Margins('u',6,45);
```

enforces ± 6 dB of gain margin and 45 degrees of phase margin at the location "u". In MATLAB, this location should be marked with a `loopswitch` block in the overall model of the control system. In Simulink, it should be registered as control, measurement, or switch using the `addControl`, `addMeasurement`, `addSwitch` methods of the `sITunable` tuning interface. Stability margins are typically measured at the plant inputs or plant outputs or both.

The target gain and phase margin values are converted into a normalized gain constraint on some appropriate closed-loop transfer function. The desired margins are achieved at frequencies where the gain is less than 1.

`viewSpec(R)`



Closed-Loop Pole Requirement

The `TuningGoal.Poles` requirement constrains the location of the closed-loop poles. You can enforce some minimum decay rate

$$\text{Re}(s) < -\alpha$$

or constrain the pole magnitude to

$$|s| < \omega_{\max}$$

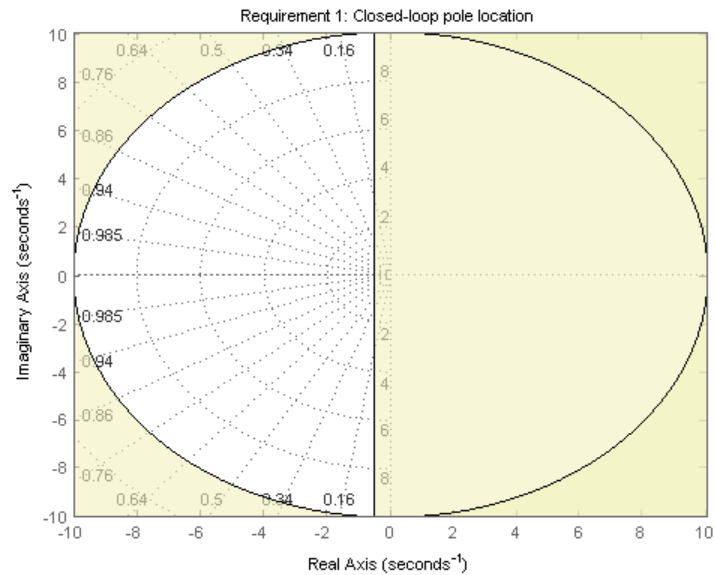
For example

```
R = TuningGoal.Poles();
```

```
R.MinDecay = 0.5;
R.MaxFrequency = 10;
```

constrains the closed-loop poles to lie in the white region below.

```
viewSpec(R)
```



Increasing the MinDecay value results in better damping and faster transients. Decreasing the MaxFrequency value prevents fast dynamics and high-frequency oscillations.

Stable Controller Requirement

The `TuningGoal.StableController` requirement enforces stability of a given control element (compensator). The tuning algorithm may produce unstable compensators for unstable plants. You can prevent this by constraining the location of the compensator poles. For example, if your compensator is parameterized as a second-order transfer function,

```
C = ltiblock.tf('C',1,2);
```

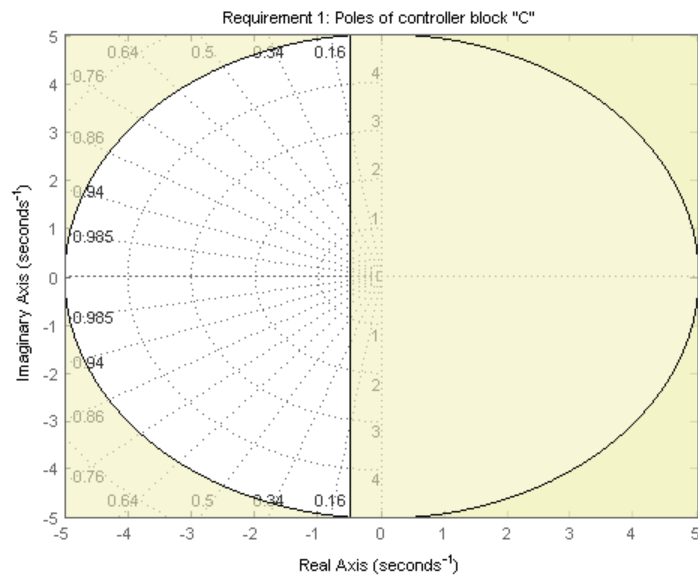
you can force it to have stable dynamics by adding the requirement

```
R = TuningGoal.StableController('C');
```

You can further constrain the minimum decay rate and maximum natural frequency of its poles, for example,

```
R.MinDecay = 0.5;
R.MaxFrequency = 5;
```

```
viewSpec(R)
```



Configuring Requirements

All requirements discussed above are objects that can be further configured by modifying their default attributes. The display shows the list of such attributes. For example

```
R = TuningGoal.Gain('d','y',1)
```

```

R =

  TuningGoal.Gain
  Package: TuningGoal

  Properties:
    Input: {'d'}
    Output: {'y'}
    MaxGain: [1x1 zpk]
    Focus: [0 Inf]
    Models: NaN
    Openings: {0x1 cell}
    Name: ''

```

Three attributes are shared by multiple requirements. The `Focus` property specifies the frequency band in which the requirement is active. For example,

```
R.Focus = [1 20];
```

limits the gain from `d` to `y` in the frequency interval [1,20] only. The `Models` property specifies which models the requirement applies to (in the context of tuning for multiple plant models). For example,

```
R.Models = [2 3 5];
```

indicates that the requirement only applies to the second, third, and fifth model in the model array supplied to `system`. Finally, the `Openings` property lets you specify additional loop openings. For example

```
R = TuningGoal.Margins('Inner',6,45);
R.Openings = 'Outer';
```

specifies stability margins for the inner loop with the outer loop open. All loop opening sites should be registered with `loopswitch` blocks in MATLAB and with the `addControl`, `addMeasurement`, `addSwitch` methods of the `slTunable` tuning interface in Simulink.

Validating Results

This example shows how to interpret and validate tuning results from `systemtune`.

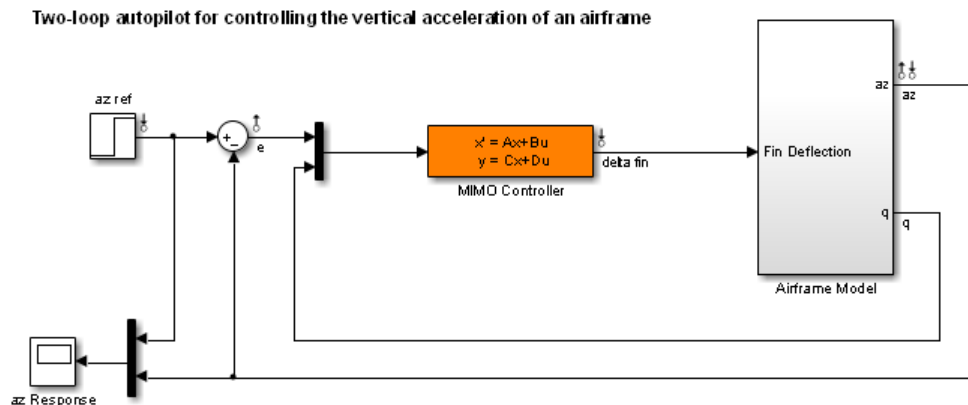
Background

You can tune the parameters of your control system with `systemtune` or `looptune`. The design specifications are captured using `TuningGoal` requirement objects. This example shows how to interpret the results from `systemtune`, graphically verify the design requirements, and perform additional open- and closed-loop analysis.

Controller Tuning with SYSTUNE

We use an autopilot tuning application as illustration, see the *"Tuning of a Two-Loop Autopilot"* example for details. The tuned compensator is the "MIMO Controller" block highlighted in orange in the model below.

```
open_system('rct_airframe2')
```



See `airframe_demo` to see how you can tune this MIMO controller with the `LOOPTUNE` command.

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The setup and tuning steps are repeated below for completeness.

```

ST0 = sITunable('rct_airframe2','MIMO Controller');
ST0.addControl('delta fin');

% Compensator parameterization
C0 = ltiblock.ss('C',2,1,2);
C0.d.Value(1) = 0; C0.d.Free(1) = false;
ST0.setBlockParam('MIMO Controller',C0)

% Requirements
Req1 = TuningGoal.Tracking('az ref','az',1); % tracking
Req2 = TuningGoal.Gain('delta fin','delta fin',tf(25,[1 0])); % roll-off
Req3 = TuningGoal.Margins('delta fin',7,45); % margins
MaxGain = frd([2 200 200],[0.02 2 200]);
Req4 = TuningGoal.Gain('delta fin','az',MaxGain); % disturbance rejection

% Tuning
Opt = systuneOptions('RandomStart',3);
rng('default')
[ST1,fSoft,~,Info] = ST0.systune([Req1,Req2,Req3,Req4],Opt);

Final: Soft = 1.5, Hard = -Inf, Iterations = 57
Final: Soft = 1.49, Hard = -Inf, Iterations = 93
Final: Soft = 1.15, Hard = -Inf, Iterations = 57
Final: Soft = 1.15, Hard = -Inf, Iterations = 82

```

Interpreting Results

`systune` run three optimizations from three different starting points and returned the best overall result. The first output `ST` is an `sITunable` interface representing the tuned control system. The second output `fSoft` contains the final values of the four requirements for the best design.

`fSoft`

`fSoft =`

```

    1.1477    1.1477    0.5458    1.1477

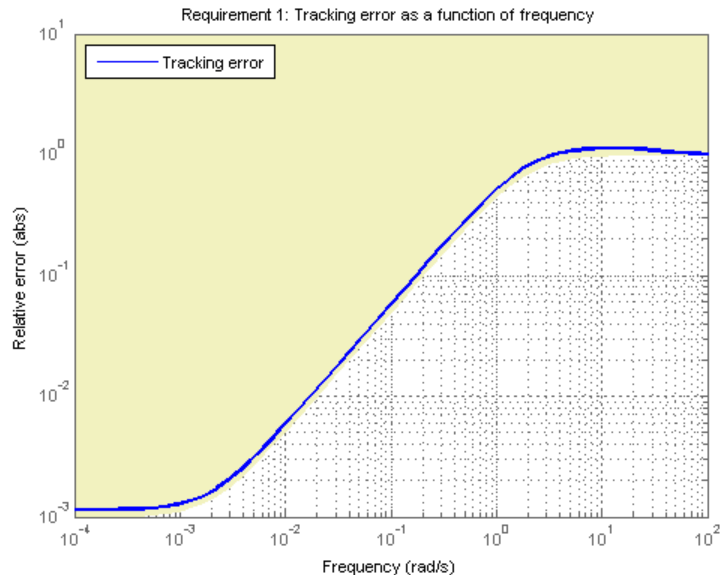
```

Requirements are normalized so a requirement is satisfied if and only if its value is less than 1. Inspection of `fSoft` reveals that Requirements 1,2,4 are active and slightly violated while Requirement 3 (stability margins) is satisfied.

Verifying Requirements

Use `viewSpec` to graphically inspect each requirement. This is useful to understand whether small violations are acceptable or what causes large violations. Make sure to provide the structure `Info` returned by `systeme` to properly account for scalings and other parameters computed by the optimization algorithms. First verify the tracking requirement.

```
viewSpec(Req1,ST1,Info)
```

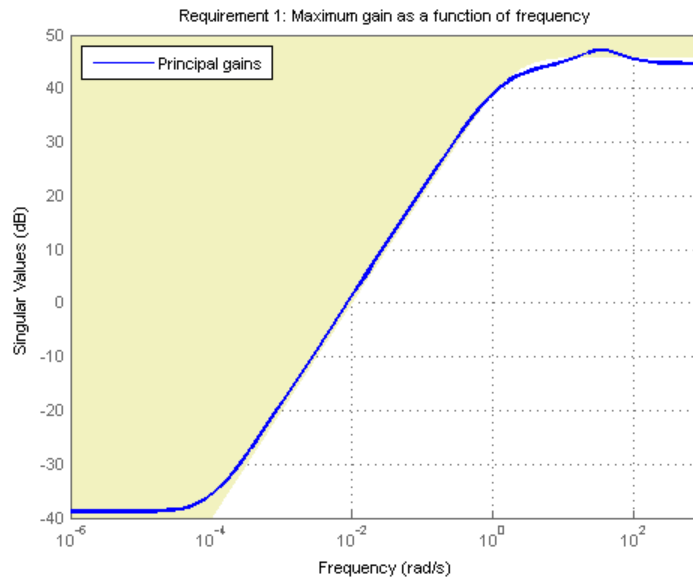


We observe a slight violation across frequency, suggesting that setpoint tracking will perform close to expectations. Similarly, verify the disturbance rejection requirement.

```
viewSpec(Req4,ST1,Info)
```

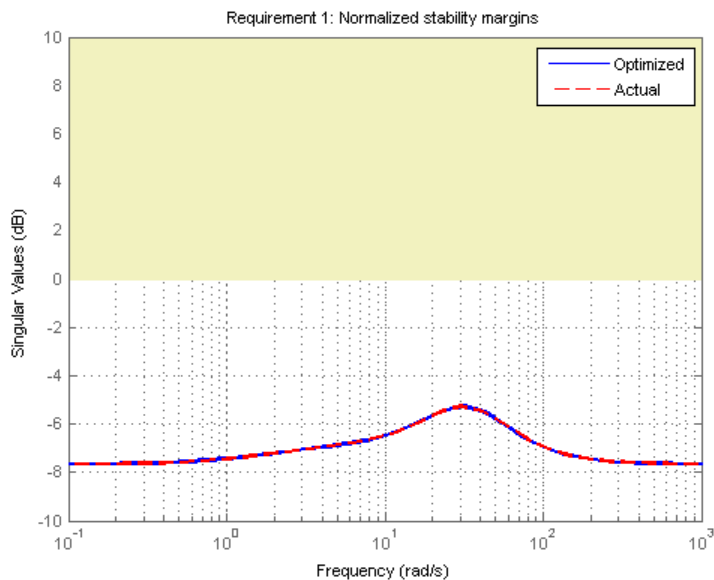


```
legend('location','NorthWest')
```



Most of the violation is at low frequency with a small bump near 35 rad/s, suggesting possible damped oscillations at this frequency. Finally, verify the stability margin requirement.

```
viewSpec(Req3,ST1,Info)
```



This requirement is satisfied at all frequencies, with the smallest margins achieved near the crossover frequency as expected. To see the actual margin values at a given frequency, click on the red curve and read the values from the data tip.

Evaluating Requirements

You can also use `evalSpec` to evaluate each requirement, that is, compute its contribution to the soft and hard constraints. For example

```
[H1,f1] = evalSpec(Req1,ST1,Info);
```

returns the value `f1` of the requirement and the underlying frequency-weighted transfer function `H1` used to compute it. You can verify that `f1` matches the first entry of `fSoft` and coincides with the peak gain of `H1`.

```
[f1 fSoft(1) getPeakGain(H1,1e-6)]
```

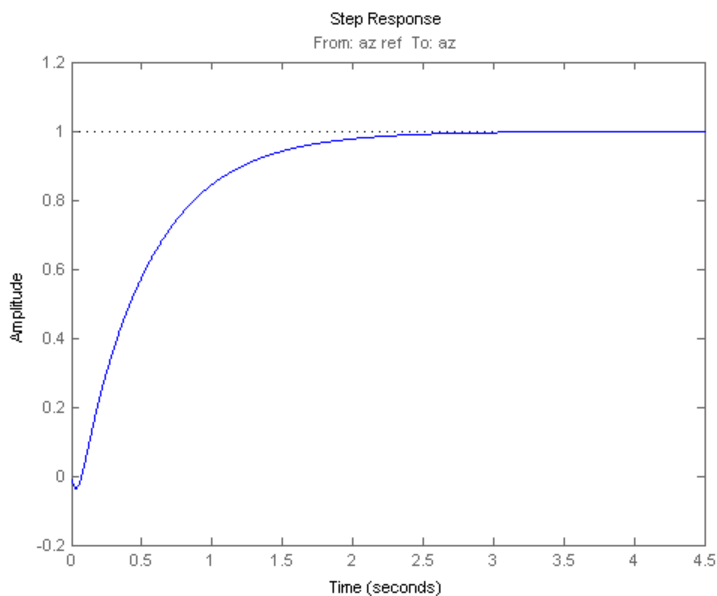
```
ans =
```

```
1.1477    1.1477    1.1477
```

Analyzing System Responses

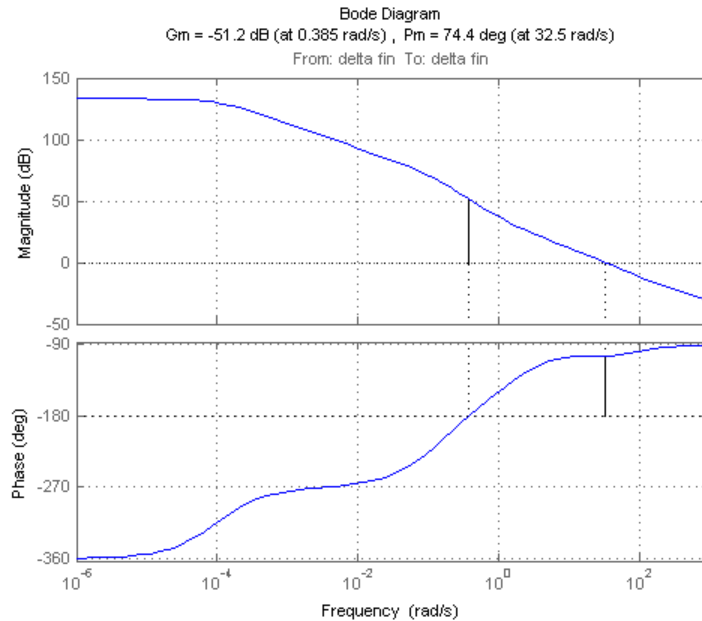
In addition to verifying requirements, you can perform basic open- and closed-loop analysis using `getIOTransfer` and `getLoopTransfer`. For example, verify tracking performance in the time domain by plotting the response `az` to a step command `azref` for the tuned system `ST1`.

```
T = ST1.getIOTransfer('az ref','az');
step(T)
```



Also plot the open-loop response measured at the plant input `delta fin`. You can use this plot to assess the classical gain and phase margins at the plant input.

```
L = ST1.getLoopTransfer('delta fin',-1); % negative-feedback loop transfer
margin(L), grid
```



Soft vs Hard Requirements

So far we have treated all four requirements equally in the objective function. Alternatively, you can use a mix of soft and hard constraints to differentiate between must-have and nice-to-have requirements. For example, you could treat Requirements 3,4 as hard constraints and optimize the first two requirements subject to these constraints. For best results, do this only after obtaining a reasonable design with all requirements treated equally.

```
[ST2,fSoft,gHard,Info] = ST1.systune([Req1 Req2],[Req3 Req4]);
```

```
Final: Soft = 1.31, Hard = 0.99979, Iterations = 137
```

```
fSoft
```

```
fSoft =
```

1.3072 1.3120

gHard

gHard =

0.4756 0.9998

Here **fSoft** contains the final values of the first two requirements (soft constraints) and **gHard** contains the the final values of the last two requirements (hard constraints). The hard constraints are satisfied since all entries of **gHard** are less than 1. As expected, the best value of the first two requirements went up as the optimizer strived to strictly enforce the fourth requirement.

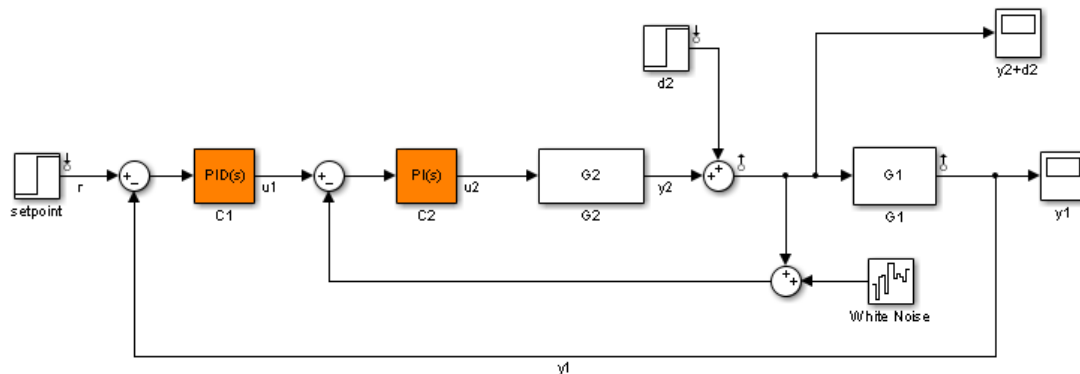
Tuning Multi-Loop Control Systems

This example shows how to jointly tune the inner and outer loops of a cascade architecture with the `system` command.

Cascaded PID Loops

Cascade control is often used to achieve smooth tracking with fast disturbance rejection. The simplest cascade architecture involves two control loops (inner and outer) as shown in the block diagram below. The inner loop is typically faster than the outer loop to reject disturbances before they propagate to the outer loop.

```
open_system('rct_cascade')
```



Plant Models and Bandwidth Requirements

In this example, the inner loop plant G_2 is

$$G_2(s) = \frac{3}{s + 2}$$

and the outer loop plant G_1 is

$$G_1(s) = \frac{10}{(s + 1)^3}$$

```
G2 = zpk([], -2, 3);
G1 = zpk([], [-1 -1 -1], 10);
```

We use a PI controller in the inner loop and a PID controller in the outer loop. The outer loop must have a bandwidth of at least 0.2 rad/s and the inner loop bandwidth should be ten times larger for adequate disturbance rejection.

Tuning the PID Controllers with SYSTUNE

When the control system is modeled in Simulink, use the `sITunable` interface in Simulink Control Design™ to set up the tuning task. List the tunable blocks and mark the signals `y1` and `y2` as loop opening locations where to measure open-loop transfers and specify loop shapes.

```
STO = sITunable('rct_cascade', {'C1', 'C2'});
STO.addSwitch({'y1', 'y2'})
```

You can query the current values of `C1` and `C2` in the Simulink model using `showTunable`. The control system is unstable for these initial values as confirmed by simulating the Simulink model.

```
STO.showTunable()
```

```
Block "rct_cascade/C1" =
```

```
Value =
```

$$K_p + K_i * \frac{1}{s}$$

```
with Kp = 0.1, Ki = 0.1
```

```
Name: C1
```

```
Continuous-time PI controller in parallel form.
```

```
-----
Block "rct_cascade/C2" =
```

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = 0.1$, $K_i = 0.1$

Name: C2

Continuous-time PI controller in parallel form.

Next use "LoopShape" requirements to specify the desired bandwidths for the the inner and outer loops. Use $0.2/s$ as target loop shape for the outer loop to enforce integral action with a gain crossover frequency at 0.2 rad/s:

```
% Outer loop bandwidth = 0.2
s = tf('s');
Req1 = TuningGoal.LoopShape('y1',0.2/s); % loop transfer measured at y1
Req1.Name = 'Outer Loop';
```

Use $2/s$ for the inner loop to make it ten times faster (higher bandwidth) than the outer loop. To constrain the inner loop transfer, make sure to open the outer loop by specifying y1 as a loop opening:

```
% Inner loop bandwidth = 2
Req2 = TuningGoal.LoopShape('y2',2/s); % loop transfer measured at y2
Req2.Openings = 'y1'; % with outer loop opened at y1
Req2.Name = 'Inner Loop';
```

You can now tune the PID gains in C1 and C2 with systune:

```
[ST,fSoft,~,Info] = ST0.systune([Req1,Req2]);
```

```
Final: Soft = 0.86, Hard = -Inf, Iterations = 96
```

Use showTunable to see the tuned PID gains.

```
ST.showTunable()
```


Block "rct_cascade/C1" =

Value =

$$K_p + K_i * \frac{1}{s} + K_d * \frac{s}{T_f * s + 1}$$

with $K_p = 0.0493$, $K_i = 0.0186$, $K_d = 0.042$, $T_f = 0.022$

Name: C1

Continuous-time PIDF controller in parallel form.

Block "rct_cascade/C2" =

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = 0.697$, $K_i = 1.64$

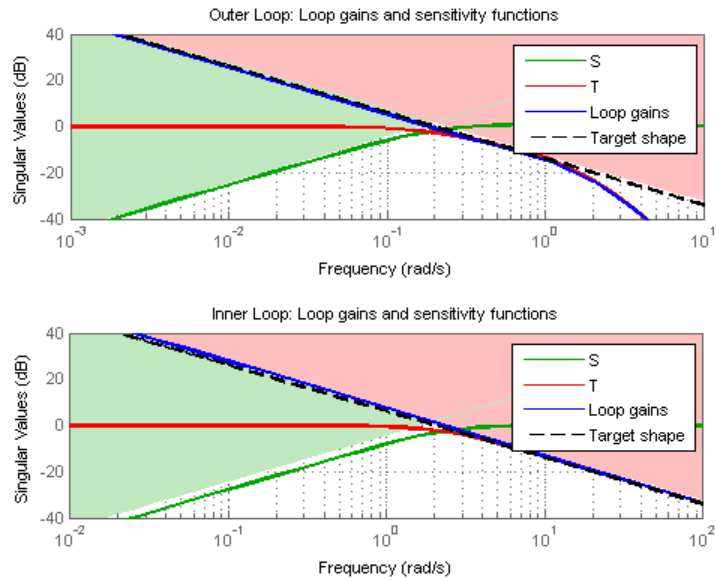
Name: C2

Continuous-time PI controller in parallel form.

Validating the Design

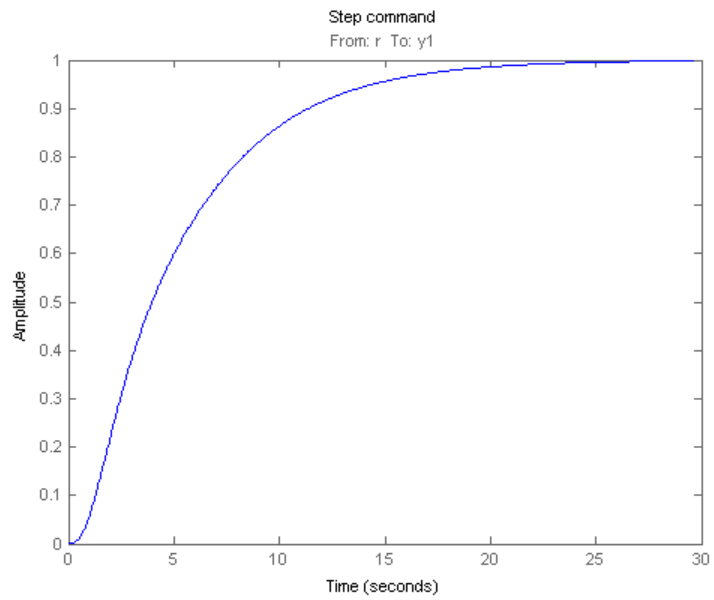
The final value is less than 1 which means that `sysTune` successfully met both loop shape requirements. Confirm this by inspecting the tuned control system `ST` with `viewSpec`

```
viewSpec([Req1,Req2],ST,Info)
```

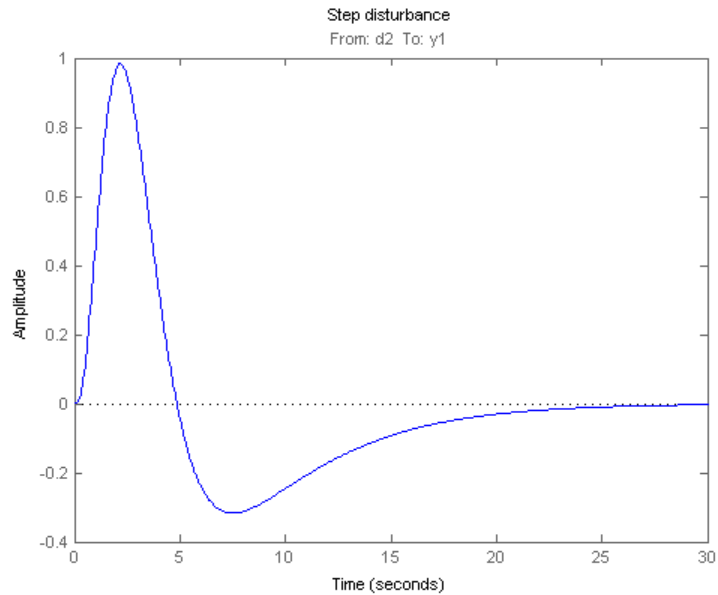


Note that the inner and outer loops have the desired gain crossover frequencies. To further validate the design, plot the tuned responses to a step command r and step disturbance d_2 :

```
% Response to a step command
H = ST.getIOTransfer('r','y1');
clf, step(H,30), title('Step command')
```



```
% Response to a step disturbance  
H = ST.getIOTransfer('d2','y1');  
step(H,30), title('Step disturbance')
```



Once you are satisfied with the linear analysis results, use `writeBlockValue` to write the tuned PID gains back to the Simulink blocks. You can then conduct a more thorough validation in Simulink.

```
ST.writeBlockValue
```

Equivalent Workflow in MATLAB

If you do not have a Simulink model of the control system, you can perform the same steps using LTI models of the plant and Control Design blocks to model the tunable elements.

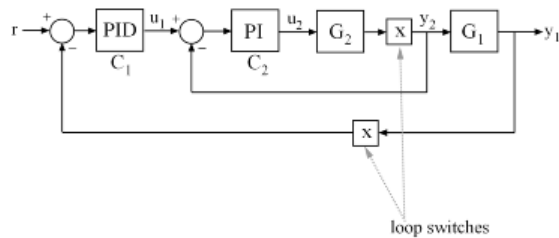


Figure 1: Cascade Architecture

First create parametric models of the tunable PI and PID controllers.

```
C1 = ltiblock.pid('C1','pid');
C2 = ltiblock.pid('C2','pi');
```

Then use loopswitch blocks to mark the loop opening locations y_1 and y_2 .

```
LS1 = loopswitch('y1');
LS2 = loopswitch('y2');
```

Finally, create a closed-loop model T_0 of the overall control system by closing each feedback loop. The result is a generalized state-space model depending on the tunable elements C_1 and C_2 .

```
InnerCL = feedback(LS2*G2*C2,1);
T0 = feedback(G1*InnerCL*C1,LS1);
T0.InputName = 'r';
T0.OutputName = 'y1';
```

You can now tune the PID gains in C_1 and C_2 with `systemtune`.

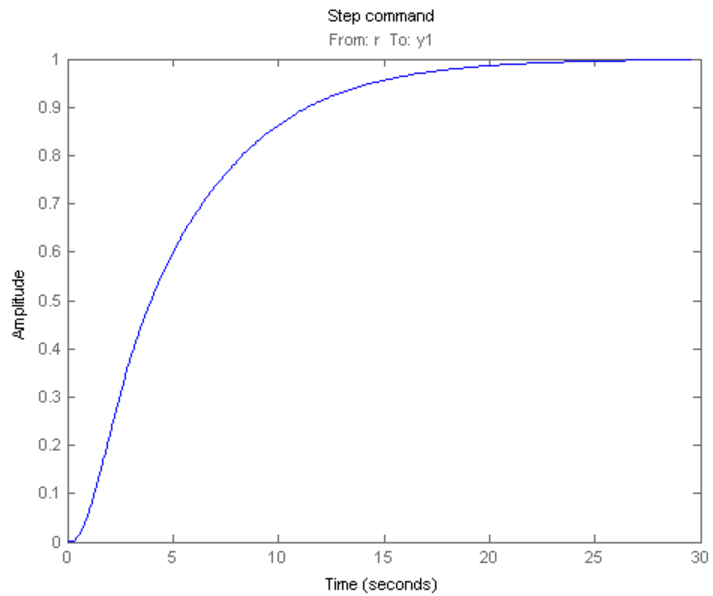
```
[T,fSoft,~,Info] = systemtune(T0,[Req1,Req2]);
```

```
Final: Soft = 0.861, Hard = -Inf, Iterations = 64
```

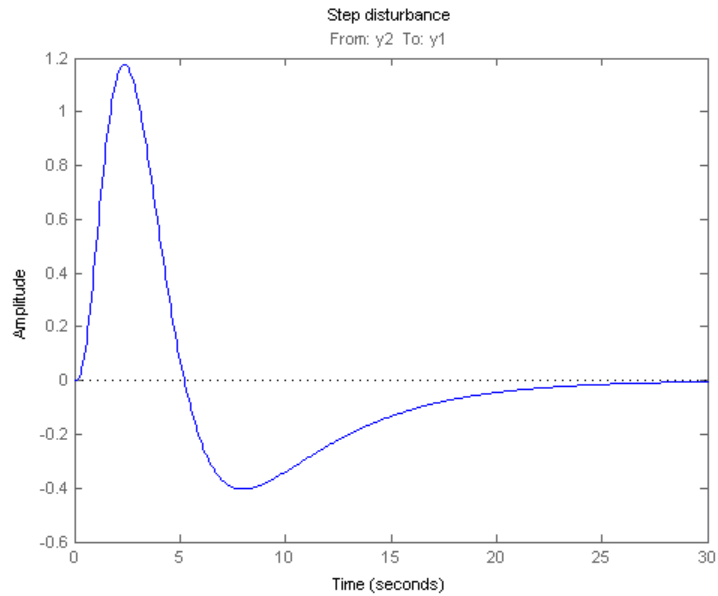
As before, use `getIOTransfer` to compute and plot the tuned responses to a step command r and step disturbance entering at the location y_2 :

```
% Response to a step command
```

```
H = getIOTransfer(T,'r','y1');  
clf, step(H,30), title('Step command')
```

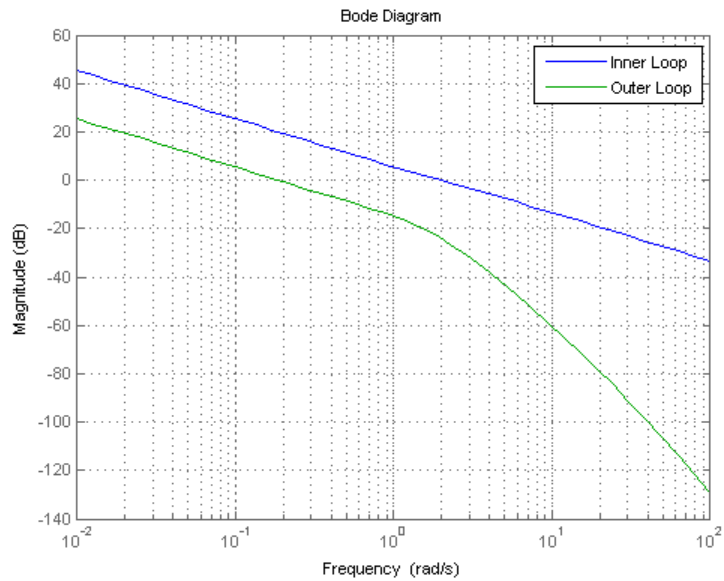


```
% Response to a step disturbance  
H = getIOTransfer(T,'y2','y1');  
step(H,30), title('Step disturbance')
```



You can also plot the open-loop gains for the inner and outer loops to validate the bandwidth requirements. Note the -1 sign to compute the negative-feedback open-loop transfer:

```
L1 = getLoopTransfer(T, 'y1', -1);           % crossover should be at .2
L2 = getLoopTransfer(T, 'y2', -1, 'y1');    % crossover should be at 2
bodemag(L2, L1, {1e-2, 1e2}), grid
legend('Inner Loop', 'Outer Loop')
```



Using Parallel Computing to Accelerate Tuning

This example shows how to leverage the Parallel Computing Toolbox™ to accelerate multi-start strategies for tuning fixed-structure control systems.

Background

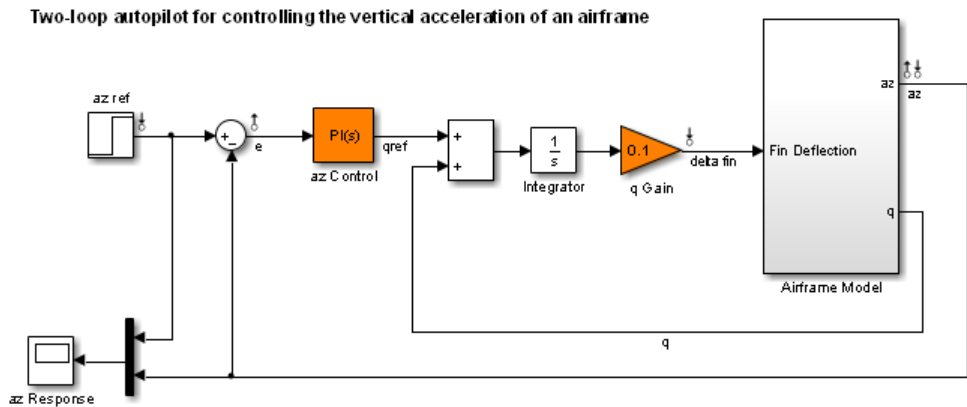
Both `systune` and `looptune` use local optimization methods for tuning the control architecture at hand. To mitigate the risk of ending up with a locally optimal but globally poor design, it is recommended to run several optimizations starting from different randomly generated initial points. If you have a multi-core machine or have access to distributed computing resources, you can significantly speed up this process using the Parallel Computing Toolbox.

This example shows how to parallelize the tuning of an airframe autopilot with `looptune`. See the example "Tuning of a Two-Loop Autopilot" for more details about this application of `looptune`.

Autopilot Tuning

The airframe dynamics and autopilot are modeled in Simulink.

```
open_system('rct_airframe1')
```



See `airframe_demo` to see how you can tune the autopilot gains with the `LOOP TUNE` command.

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The autopilot consists of two cascaded loops whose tunable elements include two PI controller gains ("az Control" block) and one gain in the pitch-rate loop ("q Gain" block). The vertical acceleration az should track the command az_{ref} with a 1 second response time. Use `sITunable` to configure this tuning task (see "Tuning of a Two-Loop Autopilot" example for details):

```
STO = sITunable('rct_airframe1',{ 'az Control', 'q Gain' });
STO.addControl('delta fin');
STO.addMeasurement({'az', 'q'});

% Design requirements
wc = [3,12]; % bandwidth
TrackReq = TuningGoal.Tracking('az ref','az',1); % tracking
```

Parallel Tuning with `LOOP TUNE`

We are ready to tune the autopilot gains with `looptune`. To minimize the risk of getting a poor-quality local minimum, run 20 optimizations starting from 20 randomly generated values of the three gains. To enable parallel processing of these 20 runs, open the MATLAB pool and configure the `looptune` options to enable parallel processing:

```
rng('default')
matlabpool open
Options = looptuneOptions('RandomStart',20,'UseParallel',true);
```

Starting matlabpool using the 'local' profile ... connected to 8 workers.

Next call `looptune` to launch the tuning algorithm. The 20 runs are automatically distributed across available computing resources:

```
[ST,gam,Info] = ST0.looptune(wc,TrackReq,Options);
```

```
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.067)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0392)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.081)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Peak gain = 1.23, Iterations = 39
Final: Peak gain = 62.1, Iterations = 38
      Min decay rate 2.1e-07 is close to lower bound 1e-07
Final: Peak gain = 1.23, Iterations = 112
```

Most runs return 1.23 as optimal gain value, suggesting that this local minimum has a wide region of attraction and is likely to be the global optimum. Use `showBlockValue` to see the corresponding gain values:

```
showBlockValue(ST)
```

Block "rct_airframe1/az Control" =

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = 0.00165$, $K_i = 0.00166$

Name: az_Control

Continuous-time PI controller in parallel form.

Block "rct_airframe1/q Gain" =

Value =

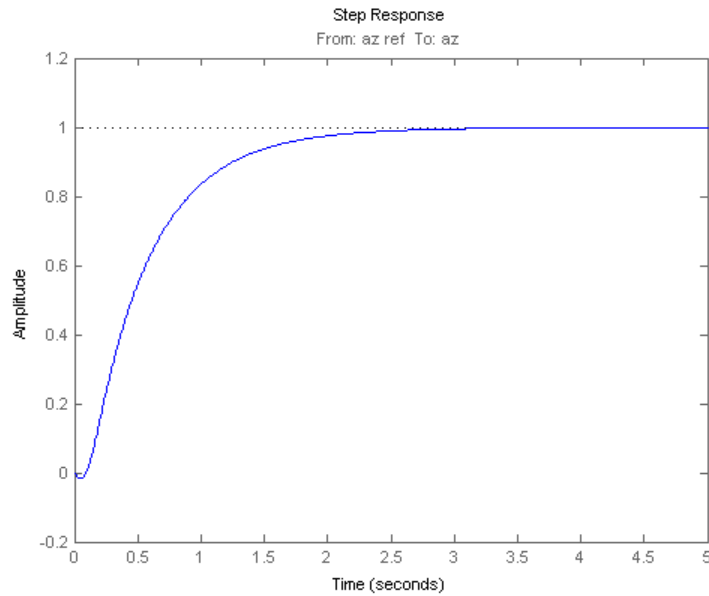
$$d = \frac{u1}{y1 \ 1.985}$$

Name: q_Gain

Static gain.

Plot the closed-loop response for this set of gains:

```
T = ST.getIOTransfer('az ref','az');
step(T,5)
```



You can now close the MATLAB pool to release the distributed computing resources used by `looptune`.

```
matlabpool close
```

```
Sending a stop signal to all the workers ... stopped.
```

Decoupling Controller for a Distillation Column

This example shows how to use Robust Control Toolbox™ to decouple the two main feedback loops in a distillation column.

Distillation Column Model

This example uses a simple model of the distillation column shown below.

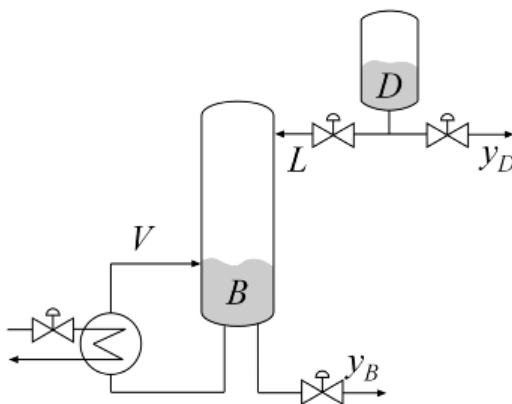


Figure 1: Distillation Column

In the so-called LV configuration, the controlled variables are the concentrations y_D and y_B of the chemicals D (tops) and B (bottoms), and the manipulated variables are the reflux L and boilup V . This process exhibits strong coupling and large variations in steady-state gain for some combinations of L and V . For more details, see Skogestad and Postlethwaite, *Multivariable Feedback Control*.

The plant is modeled as a first-order transfer function with inputs L, V and outputs y_D, y_B :

$$G(s) = \frac{1}{75s + 1} \begin{pmatrix} 87.8 & -86.4 \\ 108.2 & -109.6 \end{pmatrix}$$

```

s = tf('s','TimeUnit','minutes');
G = [87.8 -86.4 ; 108.2 -109.6]/(75*s+1);
G.InputName = {'L','V'};
G.OutputName = {'yD','yB'};

```

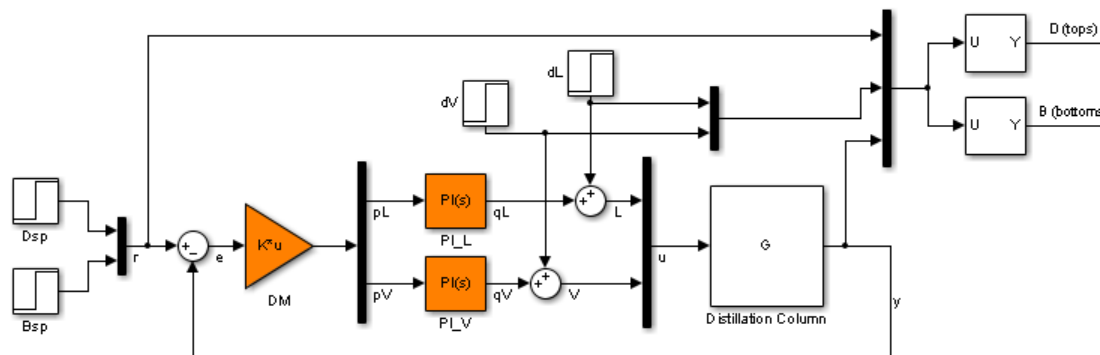
Control Architecture

The control objectives are as follows:

- Independent control of the tops and bottoms concentrations by ensuring that a change in the tops setpoint D_{sp} has little impact on the bottoms concentration B and vice versa
- Response time of about 15 minutes
- Fast rejection of input disturbances affecting the effective reflux L and boilup V

To achieve these objectives we use the control architecture shown below. This architecture consists of a static decoupling matrix DM in series with two PI controllers for the reflux L and boilup V .

```
open_system('rct_distillation')
```



Decoupling controller for a Distillation Column
(see companion demo "distill_demo")

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Controller Tuning in Simulink with LOOPTUNE

The `looptune` command provides a quick way to tune MIMO feedback loops. When the control system is modeled in Simulink, you just specify the tuned blocks, the control and measurement signals, and the desired bandwidth, and `looptune` automatically sets up the problem and tunes the controller parameters. `looptune` shapes the open-loop response to provide integral action, roll-off, and adequate MIMO stability margins.

Use the `sITunable` interface to specify the tuned blocks, plant/controller boundary, and signals for closed-loop validation.

```
% Tuned blocks (specified by their name in the model)
STO = sITunable('rct_distillation',{'PI_L','PI_V','DM'});

% Control and measurement signals
STO.addControl({'L','V'})
STO.addMeasurement('y')

% Analysis I/Os
STO.addIO('r','in')           % reference inputs
STO.addIO({'dL','dV'},'in')  % disturbances
```

Use `TuningGoal` objects to express the design requirements. Specify that each loop should respond to a step command in about 15 minutes with minimum cross-coupling.

```
TR = TuningGoal.Tracking('r','y',15);
```

Set the control bandwidth by using `[0.1,0.5]` as gain crossover band for the open-loop response.

```
wc = [0.1,0.5];
```

To enforce fast disturbance rejection, limit the gain from disturbance inputs to `yD`, `yB` as shown in the plot below. This seeks to reject disturbances at low frequency and attenuate their effect up to the control bandwidth `wc`. The desired gain profile is sketched with a few frequency points.

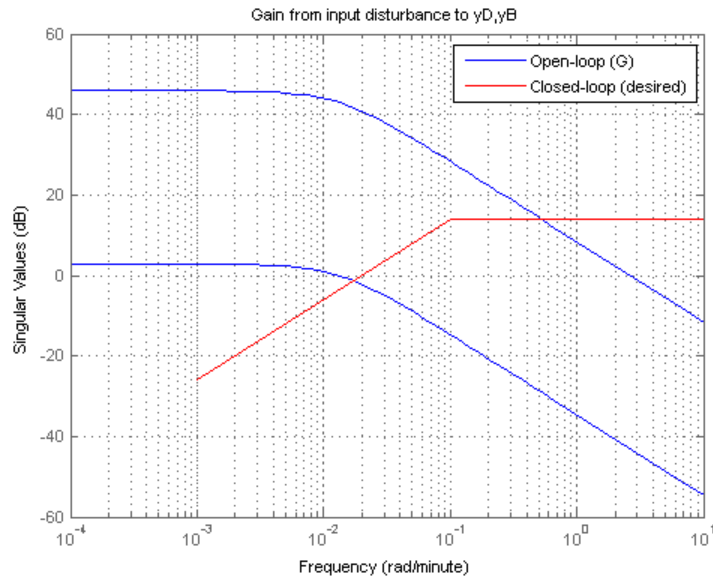
```
MaxGain = frd([0.05 5 5],[0.001 0.1 10],'TimeUnit','min');
DR = TuningGoal.MaxGain({'dL','dV'},'y',MaxGain);
```



```

% Plot gain limit and compare with open-loop gain
sigma(G,'b',MaxGain,'r'), grid
title('Gain from input disturbance to yD,yB')
legend('Open-loop (G)', 'Closed-loop (desired)')

```



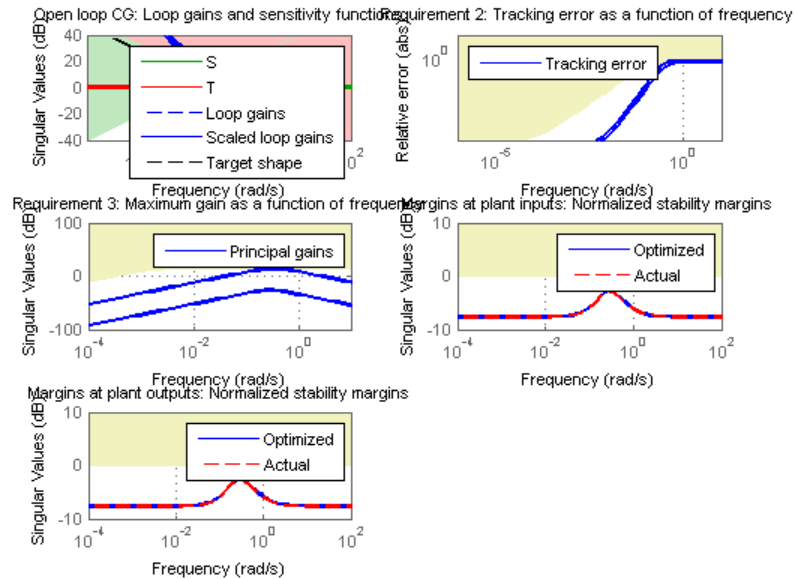
Next use `looptune` to tune the controller blocks `PI_L`, `PI_V`, and `DM` subject to these requirements.

```
[ST,gam,Info] = ST0.looptune(wc,TR,DR);
```

Final: Peak gain = 1.03, Iterations = 92

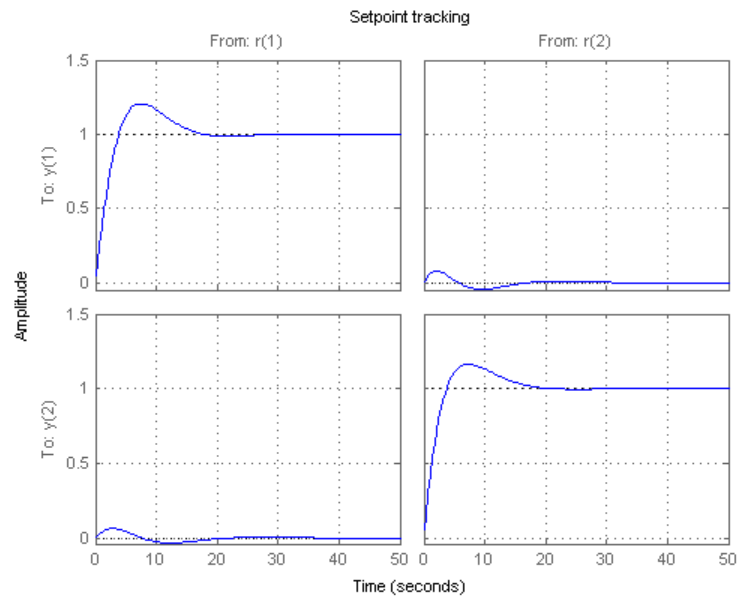
The final value is near 1 which indicates that all requirements were met. Use `loopview` to check the resulting design. The responses should stay outside the shaded areas.

```
ST.loopview(Info)
```

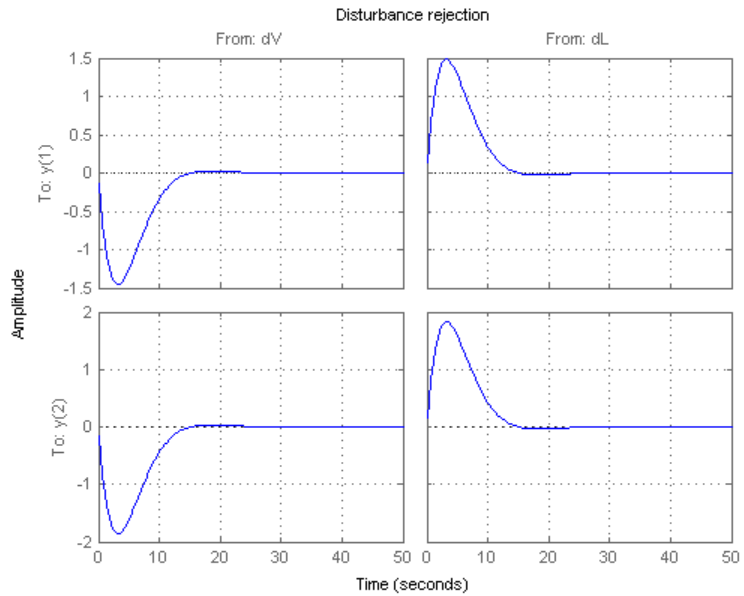


Finally, use `getIOTransfer` to access and plot the closed-loop responses from reference and disturbance to the tops and bottoms concentrations. The tuned responses show a good compromise between tracking and disturbance rejection.

```
clf
Ttrack = ST.getIOTransfer('r','y');
step(Ttrack,50), grid, title('Setpoint tracking')
```



```
Treject = ST.getIOTransfer({'dV','dL'},'y');  
step(Treject,50), grid, title('Disturbance rejection')
```



Equivalent Workflow in MATLAB

If you do not have a Simulink model of the control system, you can use LTI objects and Control Design blocks to create a MATLAB representation of the following block diagram.

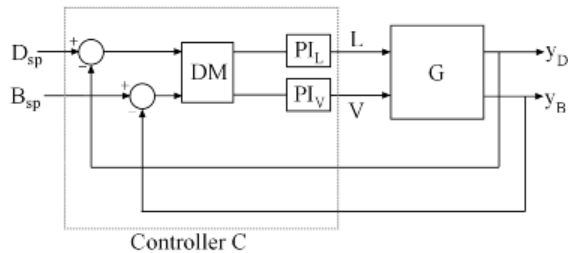


Figure 2: Block Diagram of Control System

First parameterize the tunable elements using Control Design blocks. Use the `ltiblock.gain` block to parameterize DM:

```
DM = ltiblock.gain('Decoupler',eye(2));
```

This creates a 2x2 static gain with four tunable parameters. Similarly, use the `ltiblock.pid` block to parameterize the two PI controllers:

```
PI_L = ltiblock.pid('PI_L','pi'); PI_L.TimeUnit = 'minutes';
PI_V = ltiblock.pid('PI_V','pi'); PI_V.TimeUnit = 'minutes';
```

At this point, the tunable elements are over-parameterized because multiplying DM by two and dividing the PI coefficients by two does not change the overall controller. To eliminate redundant parameters, normalize the PI controllers by fixing their proportional gain K_p to 1:

```
PI_L.Kp.Value = 1; PI_L.Kp.Free = false;
PI_V.Kp.Value = 1; PI_V.Kp.Free = false;
```

Next construct a model C_0 of the controller C in Figure 2.

```
C0 = blkdiag(PI_L,PI_V) * DM * [eye(2) -eye(2)];
```

```
% Note: I/O names should be consistent with those of G
C0.InputName = {'Dsp','Bsp','yD','yB'};
C0.OutputName = {'L','V'};
```

Now tune the controller parameters with `looptune` as done previously.

```
% Tracking requirement
TR = TuningGoal.Tracking({'Dsp','Bsp'},{'yD','yB'},15);

% Disturbance rejection requirement
DR = TuningGoal.MaxGain({'L','V'},{'yD','yB'},MaxGain);

% Crossover band
wc = [0.1,0.5];

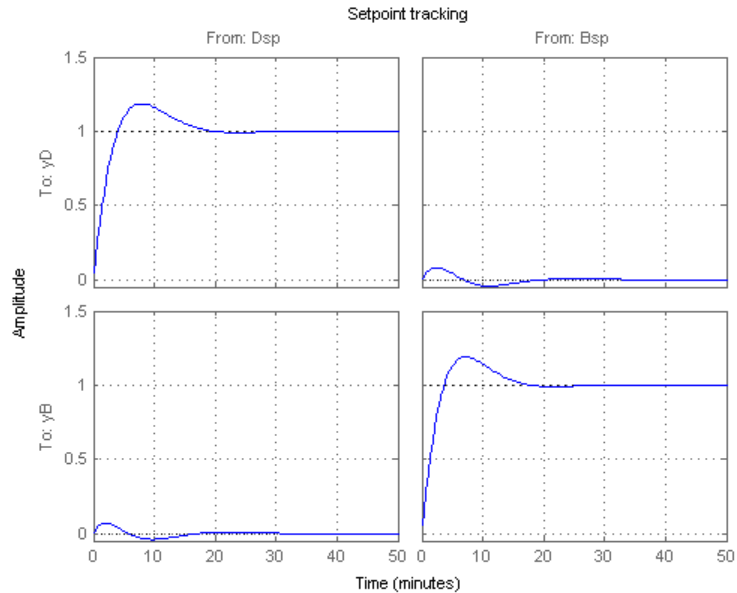
[~,C] = looptune(G,C0,wc,TR,DR);

Final: Peak gain = 1.04, Iterations = 77
```

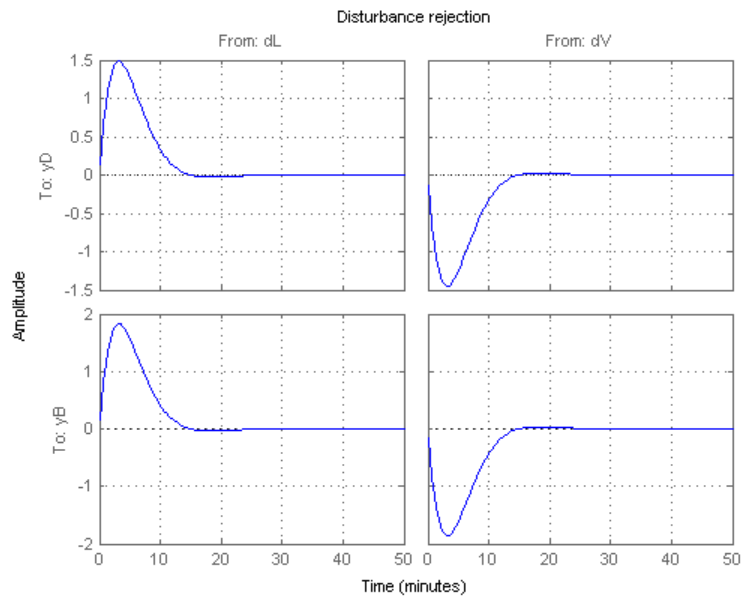
To validate the design, close the loop with the tuned compensator **C** and simulate the step responses for setpoint tracking and disturbance rejection. Also compare the open- and closed-loop disturbance rejection characteristics in the frequency domain.

```
Tcl = connect(G,C,{'Dsp','Bsp','L','V'},{'yD','yB'});
```

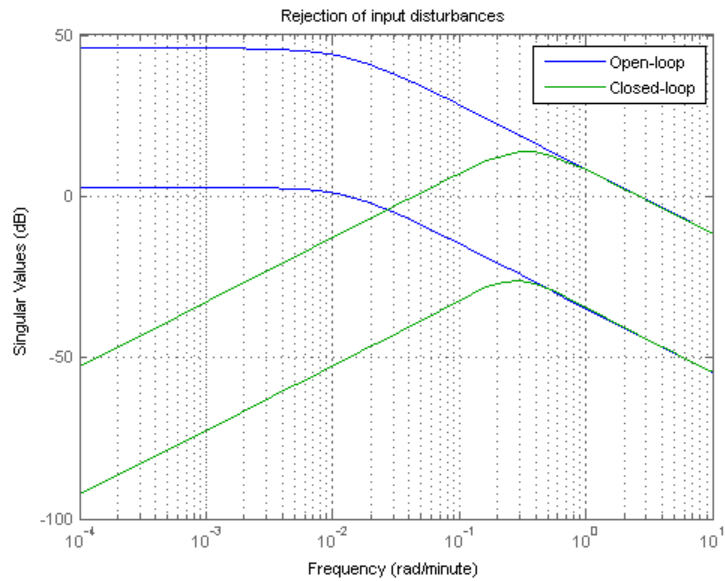
```
Ttrack = Tcl(:,[1 2]);
step(Ttrack,50), grid, title('Setpoint tracking')
```



```
Treject = Tcl(:,[3 4]);
Treject.InputName = {'dL','dV'};
step(Treject,50), grid, title('Disturbance rejection')
```



```
clf, sigma(G,Treject), grid
title('Rejection of input disturbances')
legend('Open-loop','Closed-loop')
```



The results are similar to those obtained in Simulink.

Tuning of a Digital Motion Control System

This example shows how to use Robust Control Toolbox™ to tune a digital motion control system.

Motion Control System

The motion system under consideration is shown below.

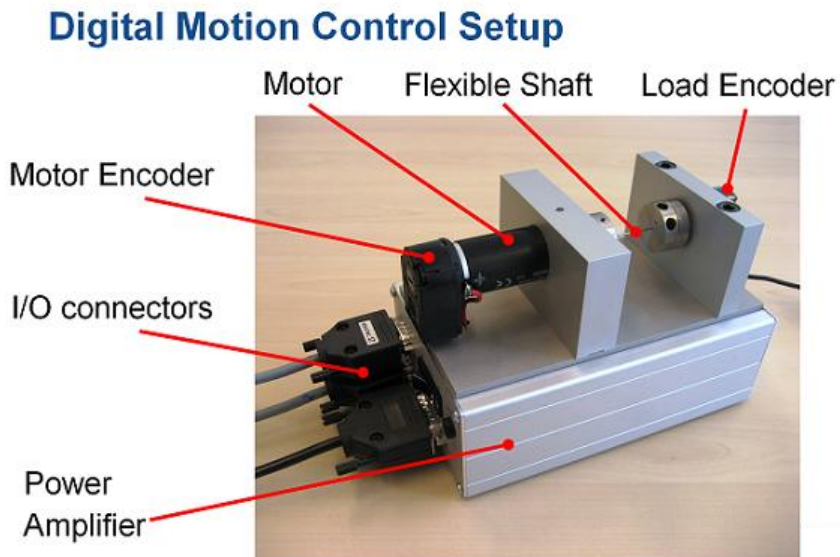


Figure 1: Digital Motion Control Hardware

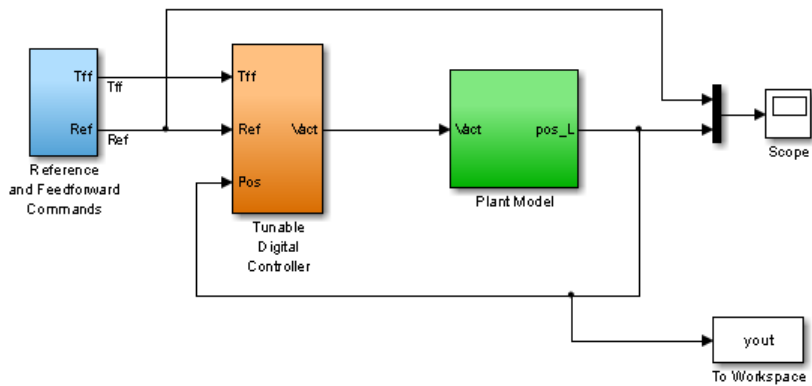
This device could be part of some production machine and is intended to move some load (a gripper, a tool, a nozzle, or anything else that you can imagine) from one angular position to another and back again. This task is part of the "production cycle" that has to be completed to create each product or batch of products.

The digital controller must be tuned to maximize the production speed of the machine without compromising accuracy and product quality. To do this,

we first model the control system in Simulink using a 4th-order model of the inertia and flexible shaft:

```
open_system('rct_dmc')
```

Digital Motion Control System



The "Tunable Digital Controller" consists of a gain in series with a lead/lag controller.

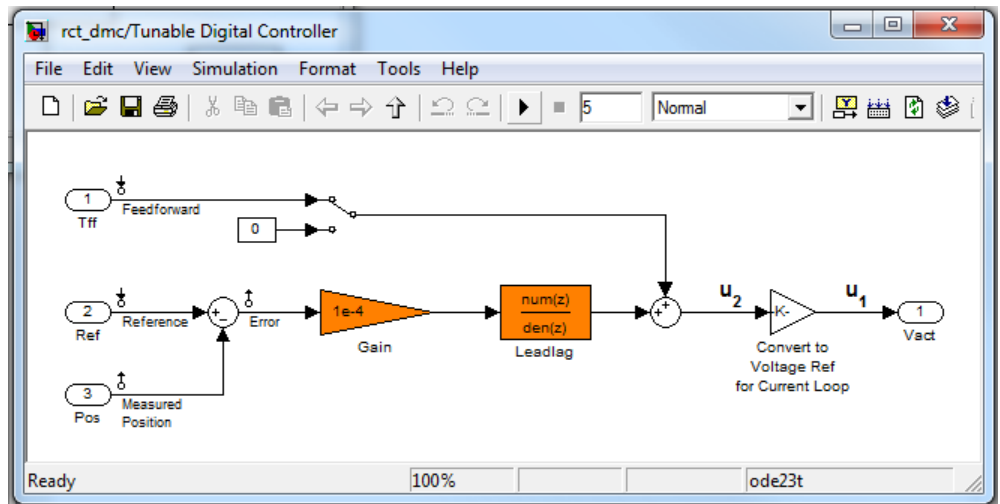
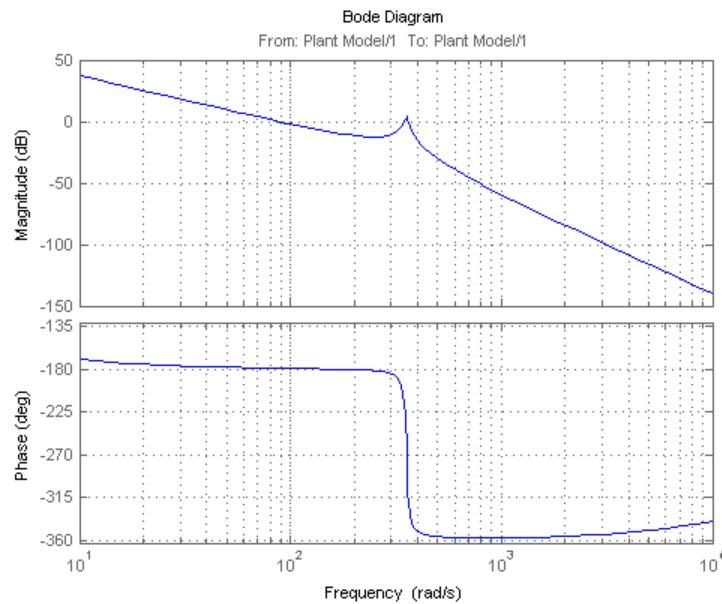


Figure 2: Digital Controller

Tuning is complicated by the presence of a flexible mode near 350 rad/s in the plant:

```
G = linearize('rct_dmc','rct_dmc/Plant Model');
bode(G,{10,1e4}), grid
```



Compensator Tuning

We are seeking a 0.5 second response time to a step command in angular position with minimum overshoot. This corresponds to a target bandwidth of approximately 5 rad/s. The `looptune` command offers a convenient way to tune fixed-structure compensators like the one in this application. To use `looptune`, first instantiate the `sITunable` interface to automatically acquire the control structure from Simulink. This requires specifying the tuned blocks and the control and measurement signals (to delimit the controller).

```
ST0 = sITunable('rct_dmc',{ 'Gain', 'Leadlag'});
ST0.addControl('Leadlag');
ST0.addMeasurement('Measured Position');
```

Next use `looptune` to tune the compensator parameters for the target gain crossover frequency of 5 rad/s:

```
ST1 = ST0.looptune(5);
```

```
Final: Peak gain = 0.975, Iterations = 21
```

A final value below or near 1 indicates success. Inspect the tuned values of the gain and lead/lag filter:

```
ST1.showBlockValue
```

```
Block "rct_dmc/Tunable Digital Controller/Gain" =
```

```
Value =
```

```

d =
      u1
y1  2.757e-06
```

```
Name: Gain
Static gain.
```

```
-----
Block "rct_dmc/Tunable Digital Controller/Leadlag" =
```

```
Value =
```

```

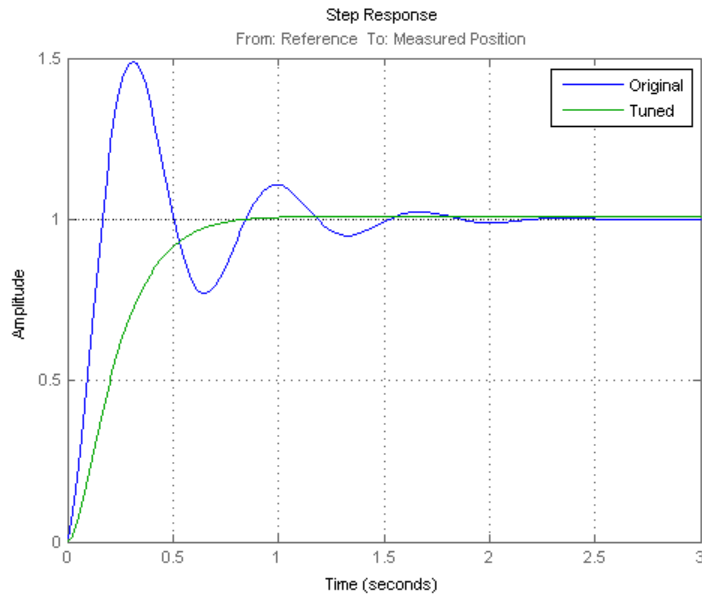
30.53 s + 59.01
-----
s + 18.97
```

```
Name: Leadlag
Continuous-time transfer function.
```

Design Validation

To validate the design, use the `slTunable` interface to quickly access the closed-loop transfer functions of interest and compare the responses before and after tuning.

```
T0 = ST0.getIOTransfer('Reference','Measured Position');
T1 = ST1.getIOTransfer('Reference','Measured Position');
step(T0,T1), grid
legend('Original','Tuned')
```



The tuned response has significantly less overshoot and satisfies the response time requirement. However these simulations are obtained using a continuous-time lead/lag compensator (`looptune` operates in continuous time) so we need to further validate the design in Simulink using a digital implementation of the lead/lag compensator. Use `writeBlockValue` to apply

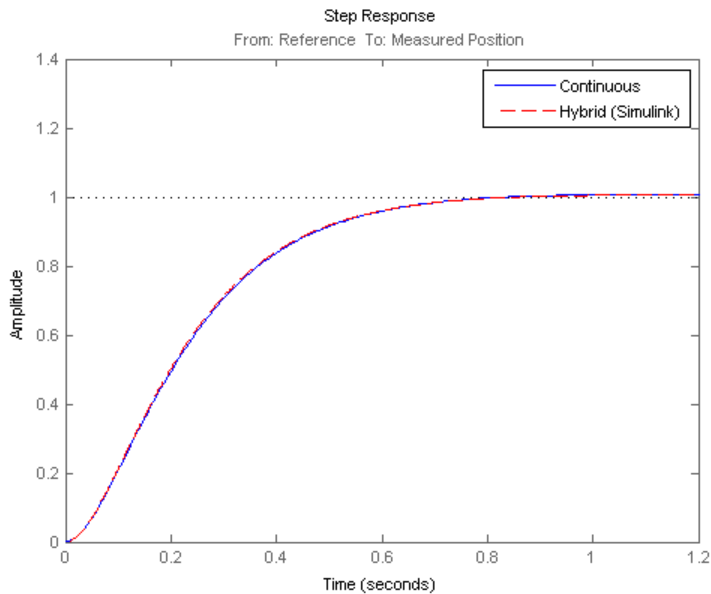
the tuned values to the Simulink model and automatically discretize the lead/lag compensator to the rate specified in Simulink.

```
ST1.writeBlockValue
```

You can now simulate the response of the continuous-time plant with the digital controller:

```
sim('rct_dmc'); % angular position logged in "yout" variable
t = yout.time;
y = yout.signals.values;
step(T1), hold, plot(t,y,'r--')
legend('Continuous','Hybrid (Simulink)')
```

Current plot held



The simulations closely match and the coefficients of the digital lead/lag can be read from the "Leadlag" block in Simulink.

Tuning an Additional Notch Filter

Next try to increase the control bandwidth from 5 to 50 rad/s. Because of the plant resonance near 350 rad/s, the lead/lag compensator is no longer sufficient to get adequate stability margins and small overshoot. One remedy is to add a notch filter as shown in Figure 3.

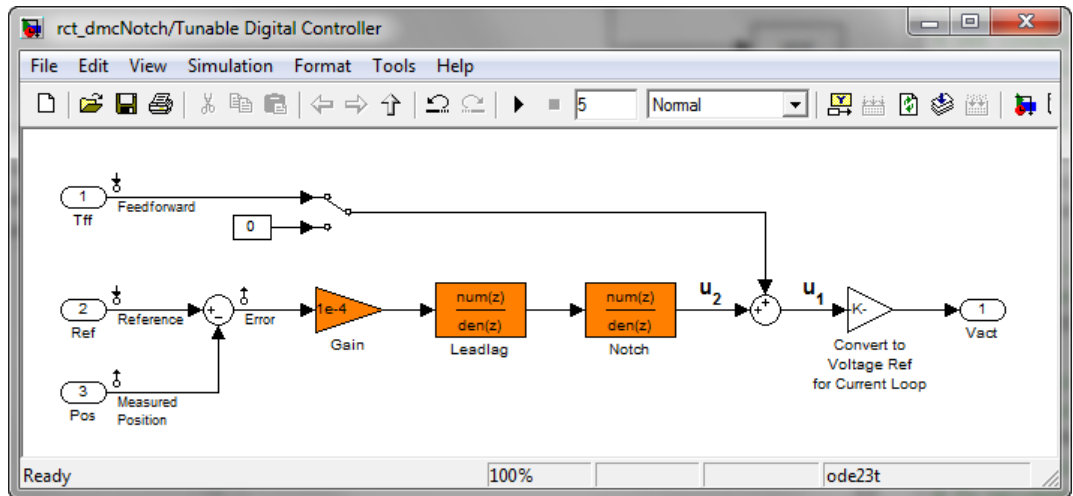


Figure 3: Digital Controller with Notch Filter

To tune this modified control architecture, specify the tunable blocks and controller I/Os as before:

```
ST0 = slTunable('rct_dmcNotch', {'Gain', 'Leadlag', 'Notch'});
ST0.addControl('Notch');
ST0.addMeasurement('Measured Position');
```

By default the "Notch" block is parameterized as any second-order transfer function. To retain the notch structure

$$N(s) = \frac{s^2 + 2\zeta_1\omega_n s + \omega_n^2}{s^2 + 2\zeta_2\omega_n s + \omega_n^2},$$

specify the coefficients ω_n , ζ_1 , ζ_2 as real parameters and create a parametric model N of the transfer function shown above:

```

wn = realp('wn',300);
zeta1 = realp('zeta1',1);    zeta1.Maximum = 1;    % zeta1 <= 1
zeta2 = realp('zeta2',1);    zeta2.Maximum = 1;    % zeta2 <= 1
N = tf([1 2*zeta1*wn wn^2],[1 2*zeta2*wn wn^2]); % tunable notch filter

```

Then associate this parametric notch model with the "Notch" block in the Simulink model. Because the control system is tuned in the continuous time, you can use a continuous-time parameterization of the notch filter even though the "Notch" block itself is discrete.

```
ST0.setBlockParam('Notch',N);
```

Next use `looptune` to jointly tune the "Gain", "Leadlag", and "Notch" blocks with a 50 rad/s target crossover frequency. To eliminate residual oscillations from the plant resonance, specify a target loop shape with a -40 dB/decade roll-off past 50 rad/s.

```

% Specify target loop shape with a few frequency points
Freqs = [5 50 500];
Gains = [10 1 0.01];
TLS = TuningGoal.LoopShape('Notch',frd(Gains,Freqs));

```

```
ST2 = ST0.looptune(TLS);
```

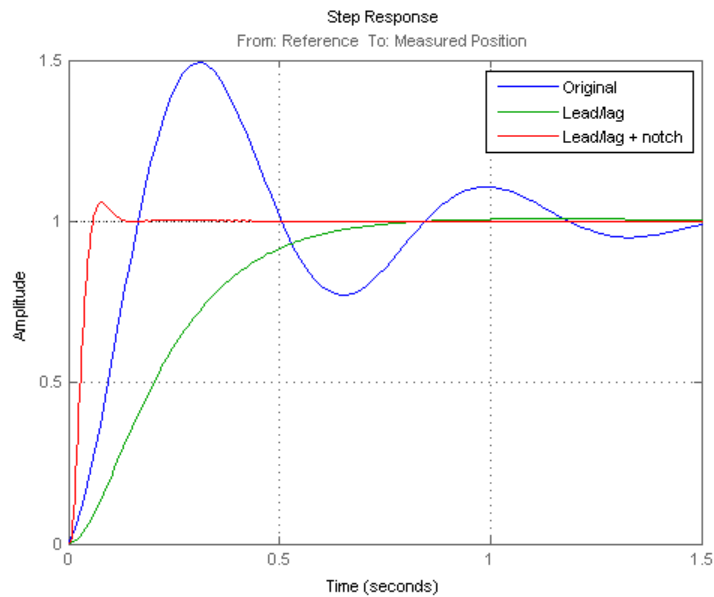
```
Final: Peak gain = 1.05, Iterations = 66
```

The final gain is close to 1, indicating that all requirements are met. Compare the closed-loop step response with the previous designs.

```

T2 = ST2.getIOTransfer('Reference','Measured Position');
clf
step(T0,T1,T2,1.5), grid
legend('Original','Lead/lag','Lead/lag + notch')

```

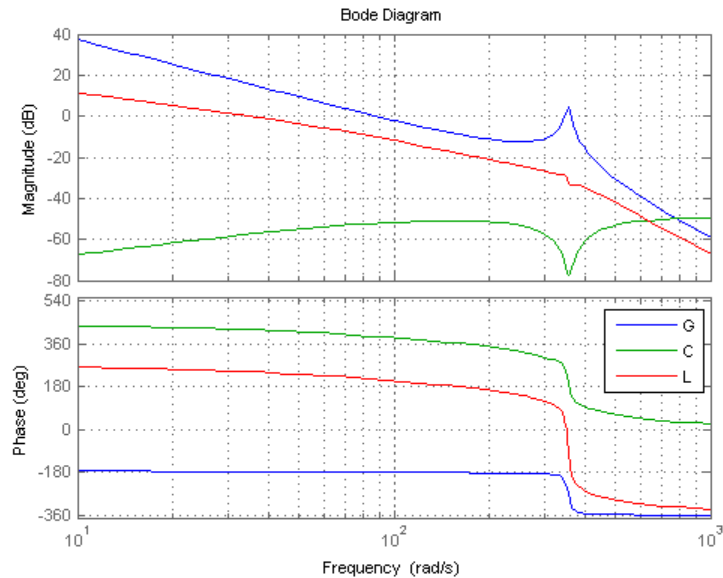



To verify that the notch filter performs as expected, evaluate the total compensator C and the open-loop response L and compare the Bode responses of G , C , L :

```
% Get tuned block values (in the order blocks are listed in ST2.TunedBlocks)
[g,LL,N] = ST2.getBlockValue();
C = N * LL * g;
```

```
L = ST2.getLoopTransfer('Notch', -1);
```

```
bode(G,C,L,{1e1,1e3}), grid
legend('G','C','L')
```



This Bode plot confirms that the plant resonance has been correctly "notched out."

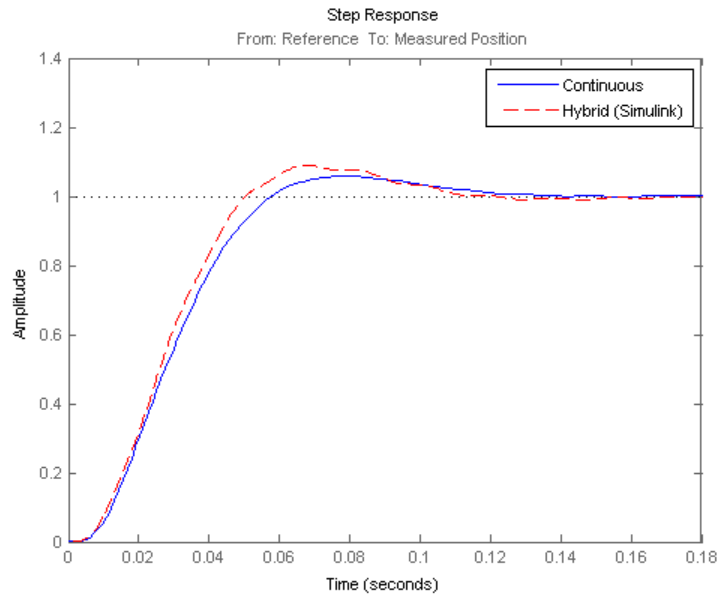
Discretizing the Notch Filter

Again use `writeBlockValue` to discretize the tuned lead/lag and notch filters and write their values back to Simulink. Compare the MATLAB and Simulink responses:

```
ST2.writeBlockValue
```

```
sim('rct_dmcNotch');
t = yout.time;
y = yout.signals.values;
step(T2), hold, plot(t,y,'r--')
legend('Continuous','Hybrid (Simulink)')
```

```
Current plot held
```



The Simulink response exhibits small residual oscillations. The notch filter discretization is the likely culprit because the notch frequency is close to the Nyquist frequency $\pi/0.002=1570$ rad/s. By default the notch is discretized using the ZOH method. Compare this with the Tustin method prewarped at the notch frequency:

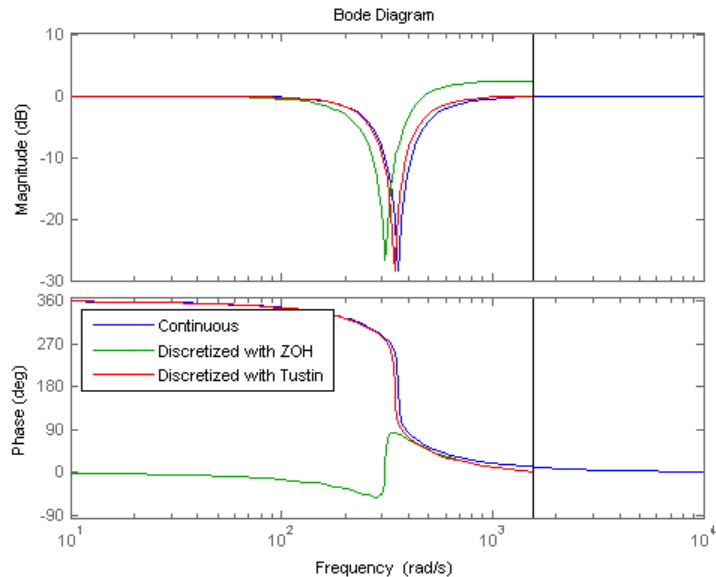
```

wn = damp(N); % natural frequency of the notch filter
Ts = 0.002; % sample time of discrete notch filter

Nd1 = c2d(N,Ts,'zoh');
Nd2 = c2d(N,Ts,'tustin',c2dOptions('PrewarpFrequency',wn(1)));

clf, bode(N,Nd1,Nd2)
legend('Continuous','Discretized with ZOH','Discretized with Tustin',...
      'Location','NorthWest')

```



The ZOH method has significant distortion and prewarped Tustin should be used instead. To do this, specify the desired rate conversion method for the notch filter block:

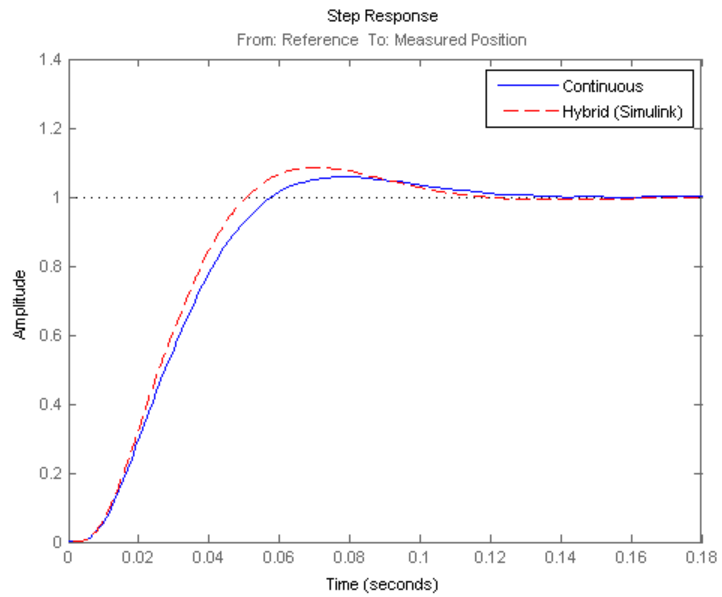
```
ST2.setBlockRateConversion('Notch','tustin',wn(1))
```

```
ST2.writeBlockValue
```

`writeBlockValue` now uses Tustin prewarped at the notch frequency to discretize the notch filter and write it back to Simulink. Verify that this gets rid of the oscillations.

```
sim('rct_dmcNotch');
t = yout.time;
y = yout.signals.values;
step(T2), hold, plot(t,y,'r--')
legend('Continuous','Hybrid (Simulink)')
```

Current plot held



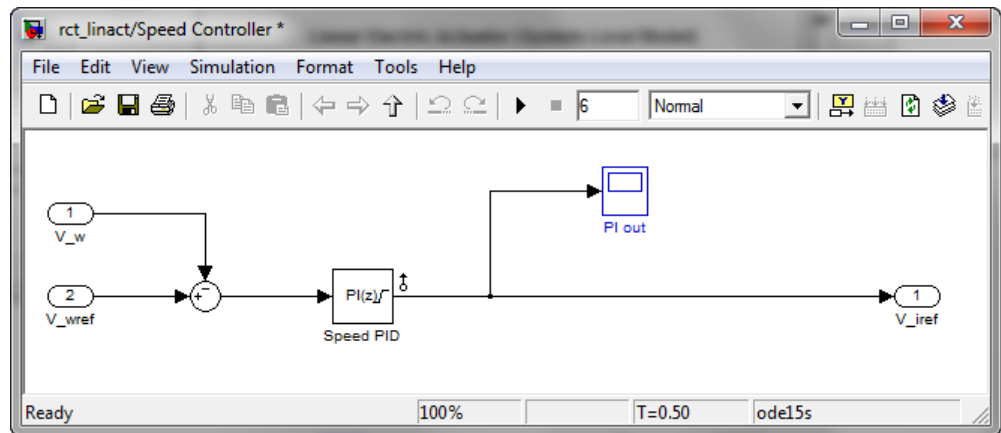
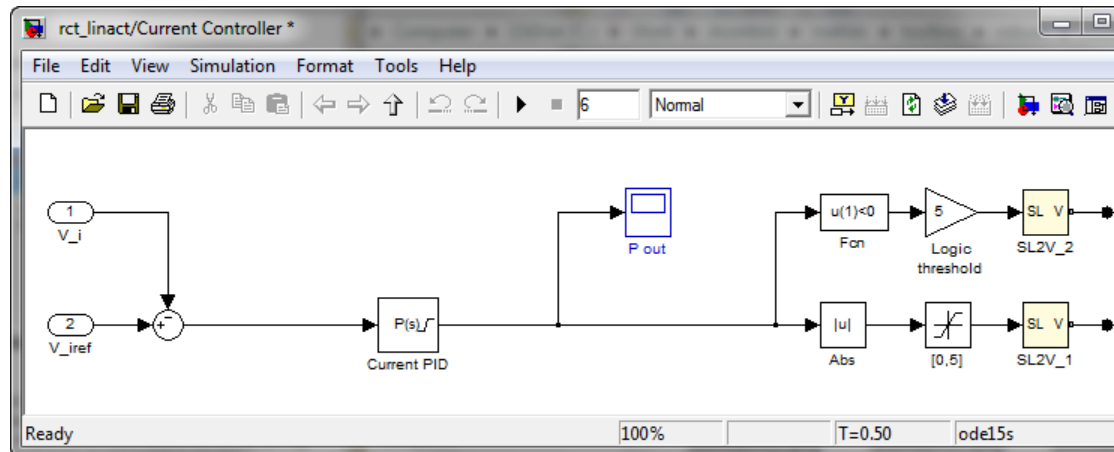


Figure 1: Current and Speed Controllers.

Note that the inner-loop (current) controller is a proportional gain while the outer-loop (speed) controller has proportional and integral actions. The output of both controllers is limited to plus/minus 5.

Design Specifications

We need to tune the proportional and integral gains to respond to a 2000 rpm speed demand in about 0.1 seconds with minimum overshoot. The initial gain settings in the model are $P=50$ and $PI(s)=0.2+0.1/s$ and the corresponding response is shown in Figure 2. This response is too slow and too sensitive to load disturbances.

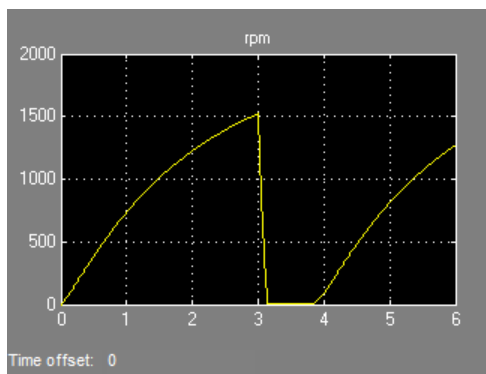


Figure 2: Untuned Response.

Control System Tuning

You can use `systemtune` to jointly tune both feedback loops. To set up the design, create an instance of the `s1Tunable` interface with the list of tuned blocks. All blocks and signals are specified by their names in the model. The model is linearized at $t=0.5$ to avoid discontinuities in some derivatives at $t=0$.

```
TunedBlocks = {'Current PID','Speed PID'};
op = findop('rct_linact',0.5); % linearize at t=0.5

% Create tuning interface
STO = s1Tunable('rct_linact',TunedBlocks,op);
```

The data structure `STO` contains a description of the control system and its tunable elements. Next specify that the DC motor should follow a 2000 rpm speed demand in 0.1 seconds:

```
TR = TuningGoal.Tracking('Speed Demand (rpm)','rpm',0.1);
```

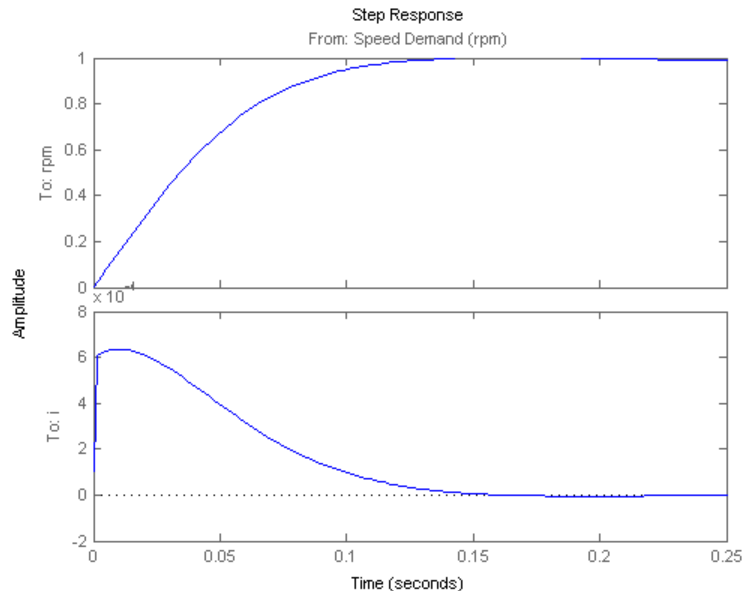

You can now tune the proportional and integral gains with `looptune`:

```
ST1 = systune(ST0,TR);
```

```
Final: Soft = 1.12, Hard = -Inf, Iterations = 25
```

This returns an updated description `ST1` containing the tuned gain values. To validate this design, plot the closed-loop response from speed demand to speed:

```
T1 = ST1.getIOTransfer('Speed Demand (rpm)', {'rpm', 'i'});
step(T1)
```



The response looks good in the linear domain so push the tuned gain values to Simulink and further validate the design in the nonlinear model.

```
ST1.writeBlockValue()
```

The nonlinear simulation results appear in Figure 3. The nonlinear behavior is far worse than the linear approximation, a discrepancy that can be traced to saturations in the inner loop (see Figure 4).

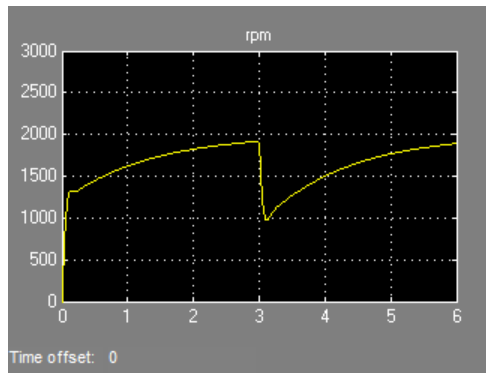


Figure 3: Nonlinear Simulation of Tuned Controller.

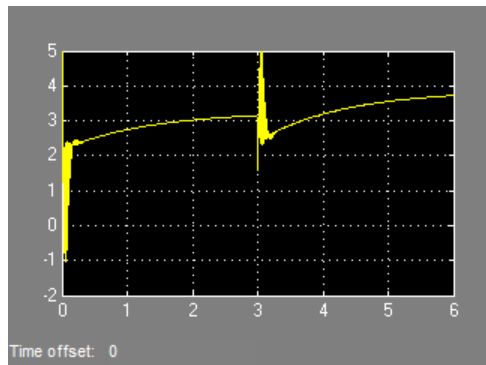


Figure 4: Current Controller Output (limited to plus/minus 5).

```
% Close model to discard this first set of tuned gains  
close_system('rct_linact',0)
```

Preventing Saturations

So far we have only specified a desired response time for the outer (speed) loop. This leaves `system` free to allocate the control effort between the inner and outer loops. Saturations in the inner loop suggest that the proportional gain is too high and that some rebalancing is needed. One possible remedy is

to explicitly limit the gain from the speed command to the outputs of the P and PI controllers. For a speed reference of 2000 rpm and saturation limits of plus/minus 5, the average gain should not exceed $5/2000 = 0.0025$. To be conservative, we can try to keep the gain from speed reference to controller outputs below 0.001. To do this, add two gain requirements and retune the controller gains with all three requirements in place.

```
% Mark the outputs of the "Current PID" and "Speed PID" blocks as control
% signals so that they can be referenced in the gain requirements
ST0 = slTunable('rct_linact',TunedBlocks,op);
ST0.addControl({'Current PID','Speed PID'})
```

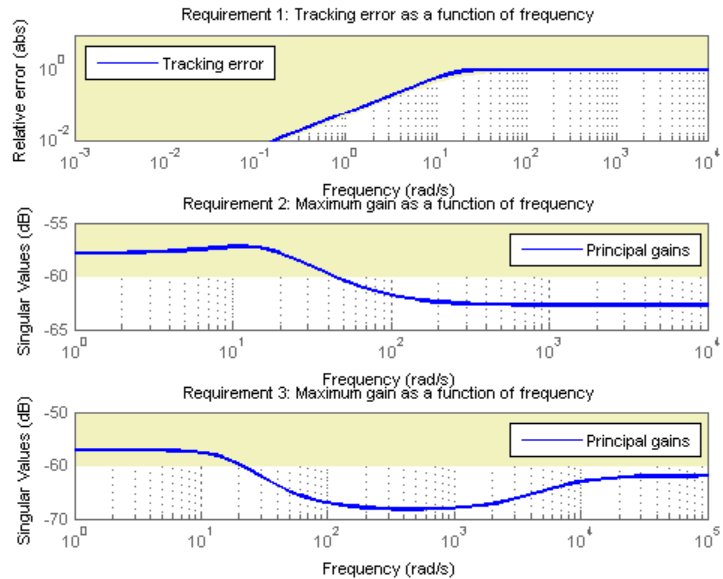
```
% Limit gain from speed demand to control signals to avoid saturation
MG1 = TuningGoal.Gain('Speed Demand (rpm)','Speed PID',0.001);
MG2 = TuningGoal.Gain('Speed Demand (rpm)','Current PID',0.001);
```

```
% Retune with these additional requirements
[ST2,~,~,info] = systune(ST0,[TR,MG1,MG2]);
```

```
Final: Soft = 1.39, Hard = -Inf, Iterations = 34
```

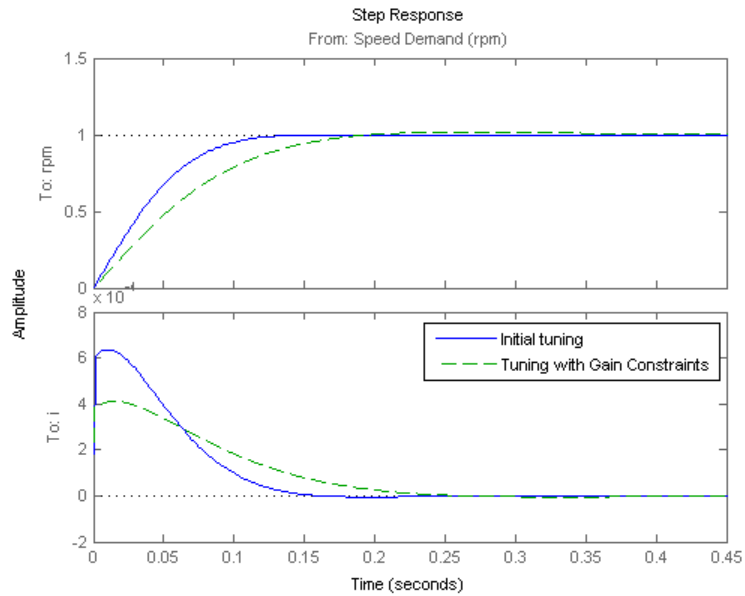
The final gain 1.39 indicates that the requirements are nearly but not exactly met (all requirements are met when the final gain is less than 1). Use `viewSpec` to inspect how the tuned controllers fare against each requirement.

```
viewSpec([TR,MG1,MG2],ST2,info)
```



Next compare the two designs in the linear domain.

```
T2 = ST2.getIOTransfer('Speed Demand (rpm)', {'rpm', 'i'});
clf, step(T1, 'b', T2, 'g--')
legend('Initial tuning', 'Tuning with Gain Constraints')
```



The second design is less aggressive but still meets the response time requirement. Finally, push the new tuned gain values to the Simulink model and simulate the response to a 2000 rpm speed demand and 500 N load disturbance. The simulation results appear in Figure 5 and the current controller output is shown in Figure 6.

```
ST2.writeBlockValue()
```

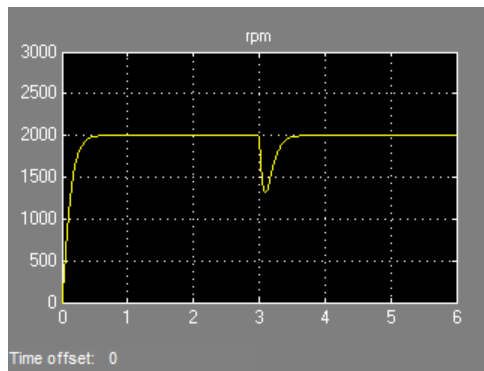


Figure 5: Nonlinear Response of Tuning with Gain Constraints.

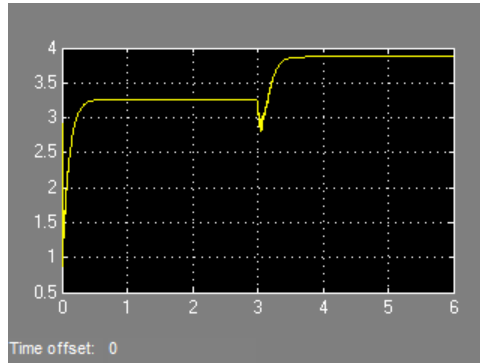


Figure 6: Current Controller Output.

The nonlinear responses are now satisfactory and the current loop is no longer saturating. The additional gain constraints have forced systune to re-distribute the control effort between the inner and outer loops so as to avoid saturation.

Active Vibration Control in Three-Story Building

This example uses systune to control seismic vibrations in a three-story building.

Background

This example considers an Active Mass Driver (AMD) control system for vibration isolation in a three-story experimental structure. This setup is used to assess control design techniques for increasing safety of civil engineering structures during earthquakes. The structure consists of three stories with an active mass driver on the top floor which is used to attenuate ground disturbances. This application is borrowed from "*Benchmark Problems in Structural Control: Part I - Active Mass Driver System*," B.F. Spencer Jr., S.J. Dyke, and H.S. Deoskar, *Earthquake Engineering and Structural Dynamics*, 27(11), 1998, pp. 1127-1139.

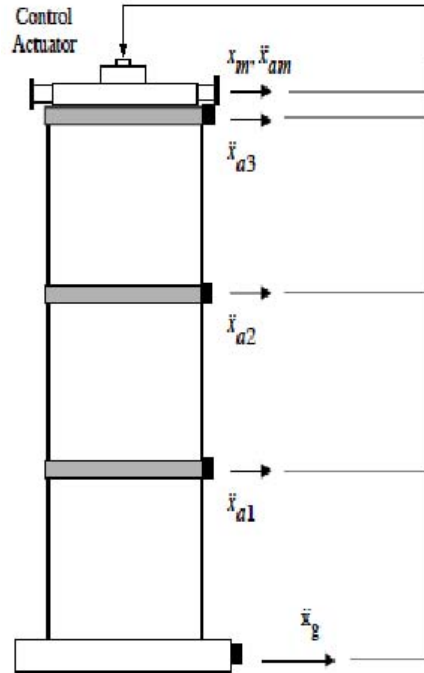


Figure 1: Active Mass Driver Control System

The plant P is a 28-state model with the following state variables:

- $x(i)$: displacement of i-th floor relative to the ground (cm)
- x_m : displacement of AMD relative to 3rd floor (cm)
- $xv(i)$: velocity of i-th floor relative to the ground (cm/s)
- xvm : velocity of AMD relative to the ground (cm/s)
- $xa(i)$: acceleration of i-th floor relative to the ground (g)
- xam : acceleration of AMD relative to the ground (g)

- $d(1)=x(1)$, $d(2)=x(2) - x(1)$, $d(3)=x(3) - x(2)$: inter-story drifts

The inputs are the ground acceleration xag (in g) and the control signal u . We use $1\ g = 981\ \text{cm/s}^2$.

```
load ThreeStoryData
size(P)
```

State-space model with 20 outputs, 2 inputs, and 28 states.

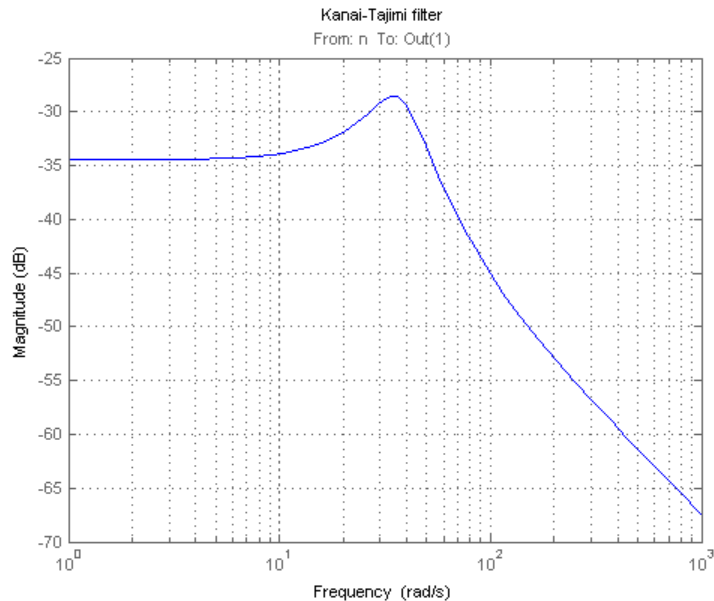
Model of Earthquake Acceleration

The earthquake acceleration is modeled as a white noise process filtered through a Kanai-Tajimi filter.

```
zg = 0.3;    wg = 37.3;
S0 = 0.03*zg/(pi*wg*(4*zg^2+1));
num = sqrt(S0)*[2*zg*wg wg^2];
den = [1 2*zg*wg wg^2];

F = sqrt(2*pi)*tf(num,den);
F.InputName = 'n';    % white noise input

bodemag(F), grid, title('Kanai-Tajimi filter')
```



Open-Loop Characteristics

The effect of an earthquake on the uncontrolled structure can be simulated by injecting a white noise input n into the plant-filter combination. You can also use `covar` to directly compute the standard deviations of the resulting inter-story drifts and accelerations.

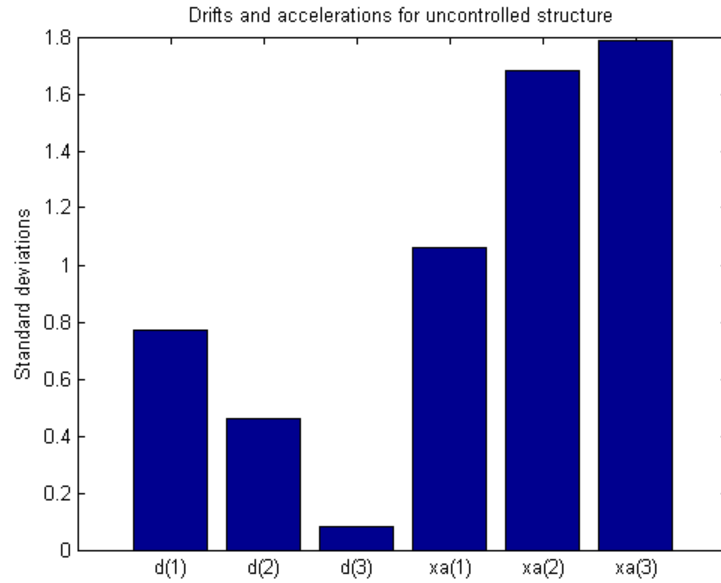
```
% Add Kanai-Tajimi filter to the plant
PF = P*append(F , 1);

% Standard deviations of open-loop drifts
CV = covar(PF('d','n'),1);
d0 = sqrt(diag(CV));

% Standard deviations of open-loop acceleration
CV = covar(PF('xa','n'),1);
xa0 = sqrt(diag(CV));

% Plot open-loop RMS values
```

```
clf, bar([d0 ; xa0])  
title('Drifts and accelerations for uncontrolled structure')  
ylabel('Standard deviations')  
set(gca, 'XTickLabel', {'d(1)', 'd(2)', 'd(3)', 'xa(1)', 'xa(2)', 'xa(3)'})
```



Control Structure and Design Requirements

The control structure is depicted in Figure 2.

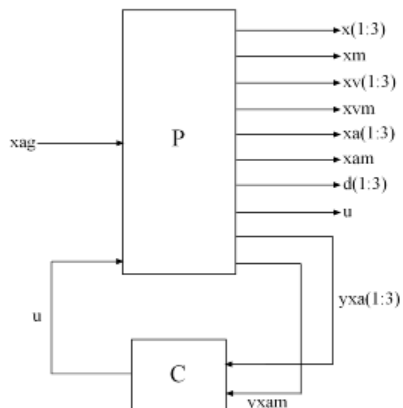


Figure 2: Control Structure

The controller C uses measurements y_{xa} and y_{xam} of x_a and x_{am} to generate the control signal u . Physically, the control u is an electrical current driving an hydraulic actuator that moves the masses of the AMD. The design requirements involve:

- Minimization of the inter-story drifts $d(i)$ and accelerations $x_a(i)$
- Hard constraints on control effort in terms of mass displacement x_m , mass acceleration x_{am} , and control effort u

All design requirements are assessed in terms of standard deviations of the corresponding signals. Use `TuningGoal.Variance` to express these requirements and scale each variable by its open-loop standard deviation to seek uniform relative improvement in all variables.

```
% Soft requirements on drifts and accelerations
Soft = [...
    TuningGoal.Variance('n','d(1)', d0(1)) ; ...
    TuningGoal.Variance('n','d(2)', d0(2)) ; ...
    TuningGoal.Variance('n','d(3)', d0(3)) ; ...
    TuningGoal.Variance('n','xa(1)', xa0(1)) ; ...
    TuningGoal.Variance('n','xa(2)', xa0(2)) ; ...
    TuningGoal.Variance('n','xa(3)', xa0(3))];
```

```
% Hard requirements on control effort
Hard = [...
    TuningGoal.Variance('n','xm', 3) ; ...
    TuningGoal.Variance('n','xam', 2) ; ...
    TuningGoal.Variance('n','u', 1)];
```

Controller Tuning

`systeme` lets you tune virtually any controller structure subject to these requirements. The controller complexity can be adjusted by trial-and-error, starting with sufficiently high order to gauge the limits of performance, then reducing the order until you observe a noticeable performance degradation. For this example, start with a 5th-order controller with no feedthrough term.

```
C = ltiblock.ss('C',5,1,4);
C.d.Value = 0;
C.d.Free = false; % Fix feedthrough to zero
```

Construct a tunable model `T0` of the closed-loop system of Figure 2 and tune the controller parameters with `systeme`.

```
% Build tunable closed-loop model
T0 = lft(PF,C);

% Tune controller parameters
[T,fSoft,gHard] = systeme(T0,Soft,Hard);
```

```
Final: Soft = 0.601, Hard = 0.99898, Iterations = 138
```

The summary indicates that we achieved an overall reduction of 40% in standard deviations (`Soft = 0.6`) while meeting all hard constraints (`Hard < 1`).

Validation

Compute the standard deviations of the drifts and accelerations for the controlled structure and compare with the uncontrolled results. The AMD control system yields significant reduction of both drift and acceleration.

```
% Standard deviations of closed-loop drifts
```

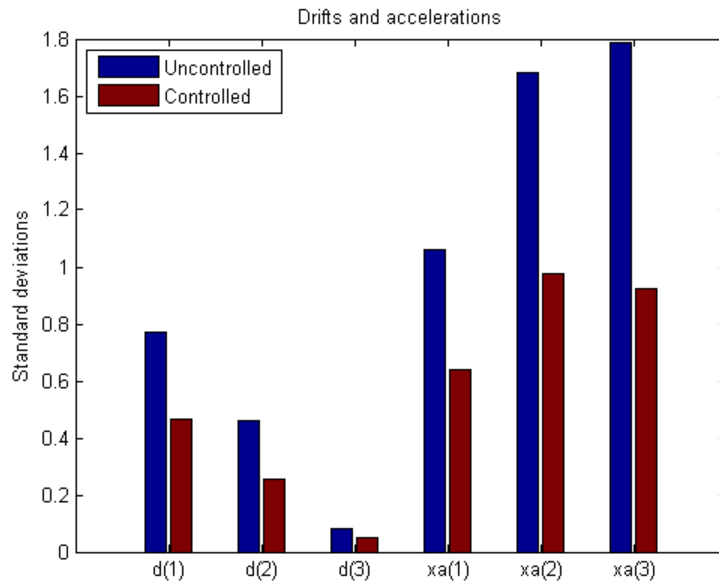
```

CV = covar(T('d','n'),1);
d = sqrt(diag(CV));

% Standard deviations of closed-loop acceleration
CV = covar(T('xa','n'),1);
xa = sqrt(diag(CV));

% Compare open- and closed-loop values
clf, bar([d0 d; xa0 xa])
title('Drifts and accelerations')
ylabel('Standard deviations')
set(gca,'XTickLabel',{'d(1)', 'd(2)', 'd(3)', 'xa(1)', 'xa(2)', 'xa(3)'})
legend('Uncontrolled', 'Controlled', 'location', 'NorthWest')

```



Simulate the response of the 3-story structure to an earthquake-like excitation in both open and closed loop. The earthquake acceleration is modeled as a white noise process colored by the Kanai-Tajimi filter.

```
% Sampled white noise process
```

```

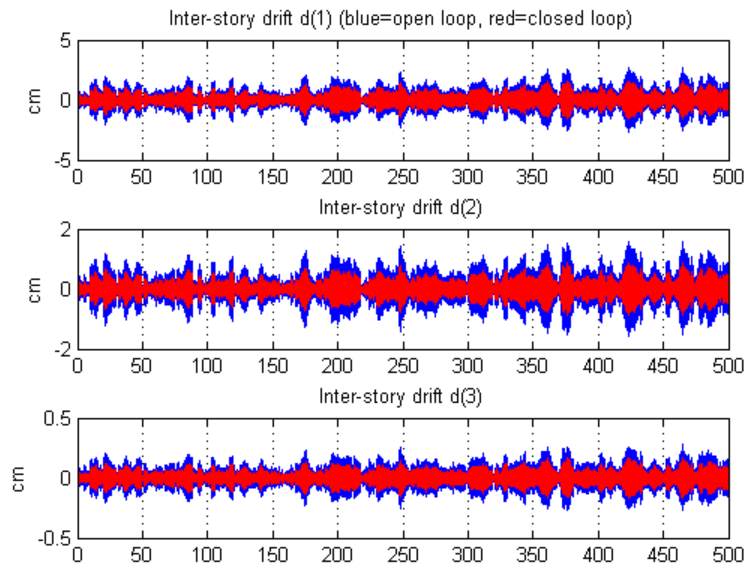
rng('default')
dt = 1e-3;
t = 0:dt:500;
n = randn(1,length(t))/sqrt(dt); % white noise signal

% Open-loop simulation
ysimOL = lsim(PF(:,1), n , t);

% Closed-loop simulation
ysimCL = lsim(T, n , t);

% Drifts
clf, subplot(311); plot(t,ysimOL(:,13),'b',t,ysimCL(:,13),'r'); grid;
title('Inter-story drift d(1) (blue=open loop, red=closed loop)'); ylabel('cm');
subplot(312); plot(t,ysimOL(:,14),'b',t,ysimCL(:,14),'r'); grid;
title('Inter-story drift d(2)'); ylabel('cm');
subplot(313); plot(t,ysimOL(:,15),'b',t,ysimCL(:,15),'r'); grid;
title('Inter-story drift d(3)'); ylabel('cm');

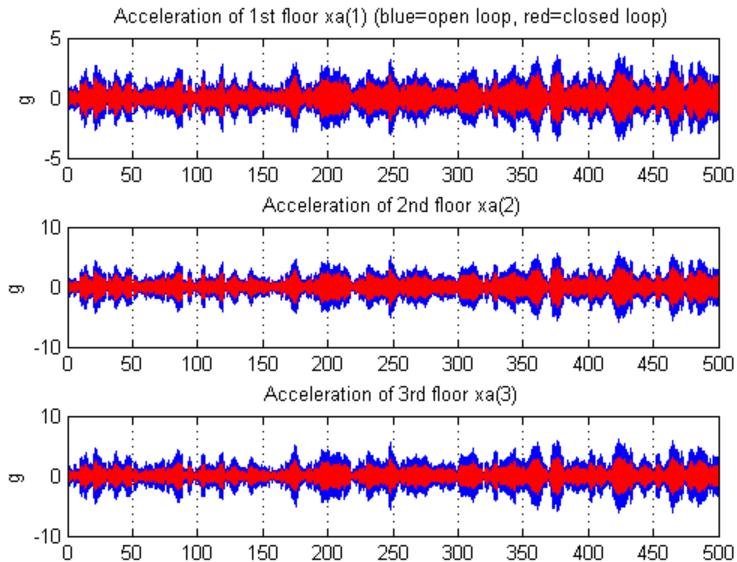
```



```

% Accelerations
clf, subplot(311); plot(t,ysimOL(:,9),'b',t,ysimCL(:,9),'r'); grid;
title('Acceleration of 1st floor xa(1) (blue=open loop, red=closed loop)');
subplot(312); plot(t,ysimOL(:,10),'b',t,ysimCL(:,10),'r'); grid;
title('Acceleration of 2nd floor xa(2)'); ylabel('g');
subplot(313); plot(t,ysimOL(:,11),'b',t,ysimCL(:,11),'r'); grid;
title('Acceleration of 3rd floor xa(3)'); ylabel('g');

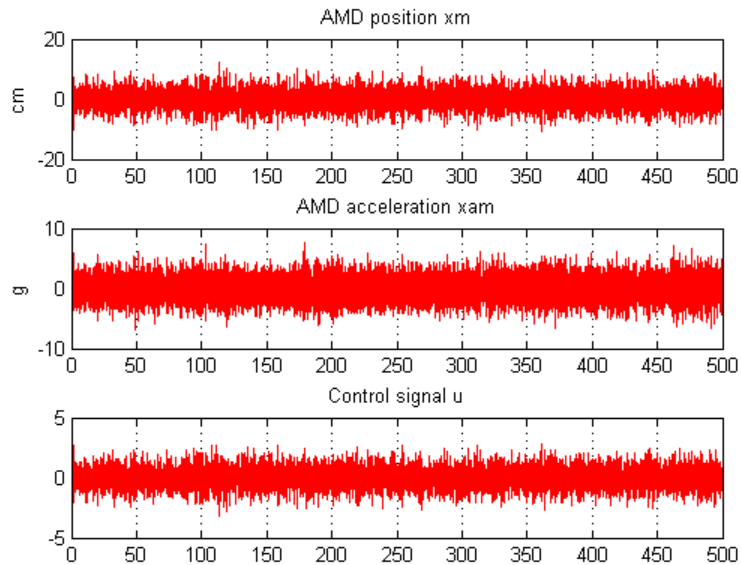
```



```

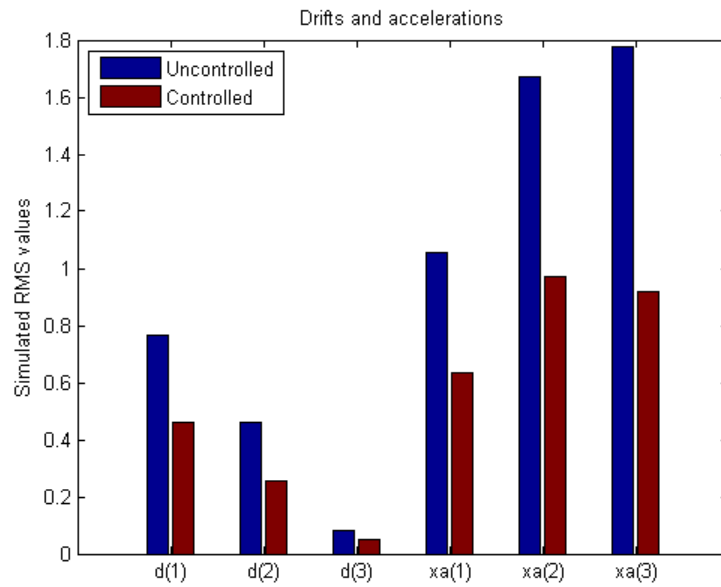
% Control variables
clf, subplot(311); plot(t,ysimCL(:,4),'r'); grid;
title('AMD position xm'); ylabel('cm');
subplot(312); plot(t,ysimCL(:,12),'r'); grid;
title('AMD acceleration xam'); ylabel('g');
subplot(313); plot(t,ysimCL(:,16),'r'); grid;
title('Control signal u');

```

Plot the root-mean-square (RMS) of the simulated signals for both the controlled and uncontrolled scenarios. Assuming ergodicity, the RMS performance can be estimated from a single sufficiently long simulation of the process and coincides with the standard deviations computed earlier. Indeed the RMS plot closely matches the standard deviation plot obtained earlier.

```
clf, bar([std(ysimOL(:,13:15)) std(ysimOL(:,9:11)) ; ...
         std(ysimCL(:,13:15)) std(ysimCL(:,9:11))])
title('Drifts and accelerations')
ylabel('Simulated RMS values')
set(gca,'XTickLabel',{'d(1)', 'd(2)', 'd(3)', 'xa(1)', 'xa(2)', 'xa(3)'})
legend('Uncontrolled', 'Controlled', 'location', 'NorthWest')
```



Overall, the controller achieves significant reduction of ground vibration both in terms of drift and acceleration for all stories while meeting the hard constraints on control effort and mass displacement.

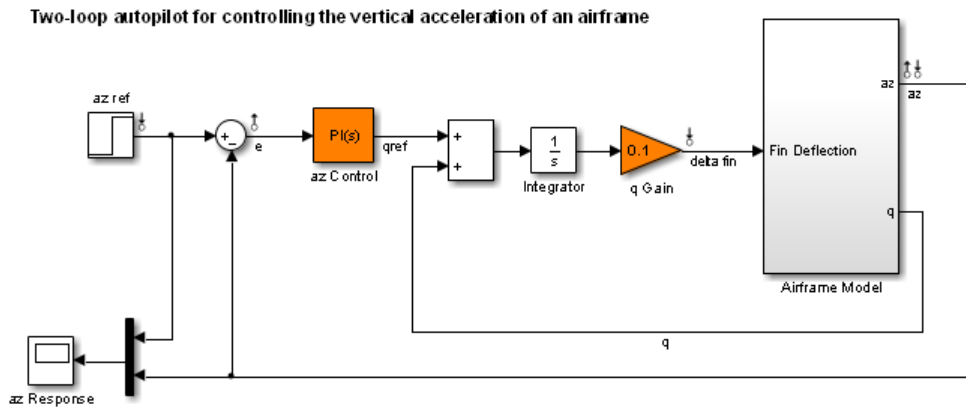
Tuning of a Two-Loop Autopilot

This example shows how to use Robust Control Toolbox™ to tune a two-loop autopilot controlling the pitch rate and vertical acceleration of an airframe.

Model of Airframe Autopilot

The airframe dynamics and the autopilot are modeled in Simulink. To open the model, type

```
open_system('rct_airframe1')
```



See `airframe_demo` to see how you can tune the autopilot gains with the `LOOP Tune` command.

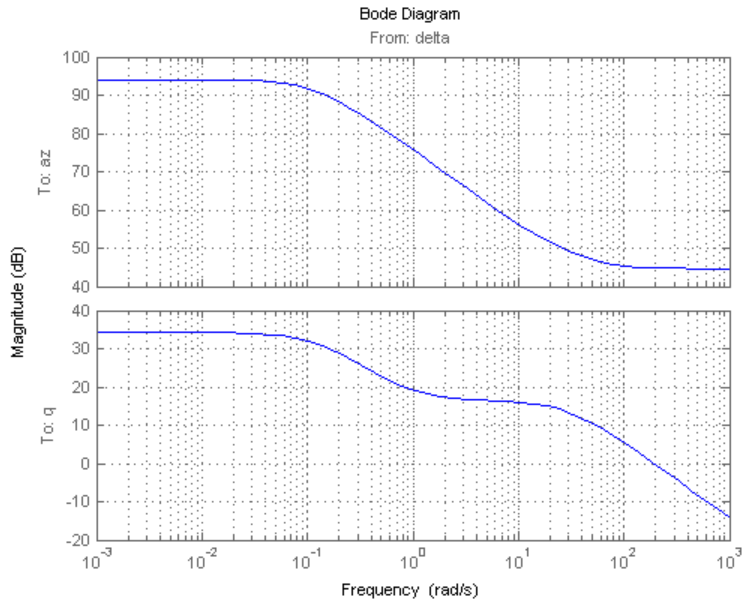
Copyright 2004-2010 The MathWorks, Inc.

The autopilot consists of two cascaded loops. The inner loop controls the pitch rate q , and the outer loop controls the vertical acceleration az in response to the pilot stick command az_{ref} . In this architecture, the tunable elements include the PI controller gains ("az Control" block) and the pitch-rate gain ("q Gain" block). The autopilot must be tuned to respond to a step command az_{ref} in about 1 second with minimal overshoot.

To analyze the airframe dynamics, linearize the "Airframe Model" block and plot the gains from the fin deflection δa to az and q :

```
G = linearize('rct_airframe1','rct_airframe1/Airframe Model');
G.InputName = 'delta';
G.OutputName = {'az','q'};

bodemag(G), grid
```



Note that the airframe model has an unstable pole:

```
pole(G)
```

ans =

```
0.1213
-29.4685
```

Frequency-Domain Tuning with LOOPTUNE

You can use the `looptune` function to automatically tune multi-loop control systems subject to basic requirements such as integral action, adequate stability margins, and desired bandwidth. To apply `looptune` to the autopilot model, create an instance of the `sITunable` interface and designate the Simulink blocks `az Control` and `q Gain` as tunable:

```
STO = sITunable('rct_airframe1',{'az Control','q Gain'});
```

Next outline the plant/controller boundary by specifying the control and measurement signals:

```
STO.addControl('delta fin');
STO.addMeasurement({'az','q'});
```

Finally, tune the control system parameters to meet the 1 second response time requirement. In the frequency domain, this roughly corresponds to a gain crossover frequency $\omega_c = 5$ rad/s for the open-loop response at the plant input "delta fin". Run a few optimizations from random starting points to increase the likelihood of finding a stabilizing controller for this unstable plant:

```
rng('default')
Options = looptuneOptions('RandomStart',10);
wc = 5;
[ST,gam,Info] = STO.looptune(wc,Options);
```

```
gam
```

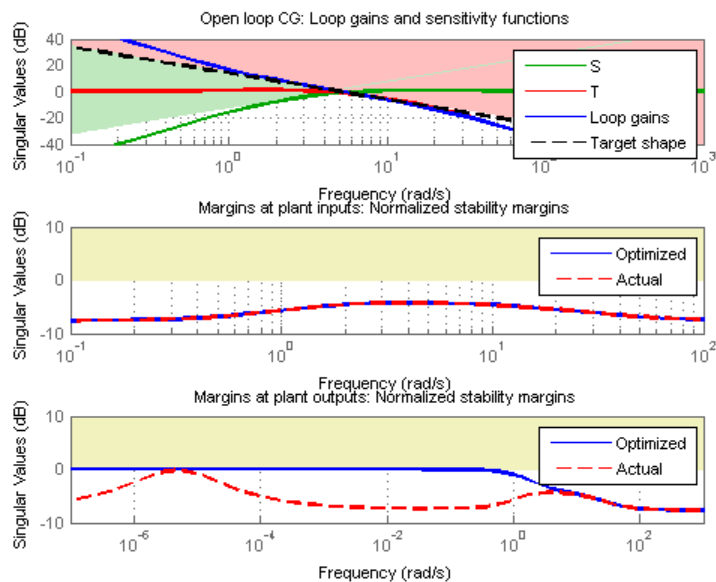
```
Final: Peak gain = 1, Iterations = 32
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Peak gain = 1, Iterations = 42
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Peak gain = 1, Iterations = 34
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0317)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
```

```
gam =
```

1.0002

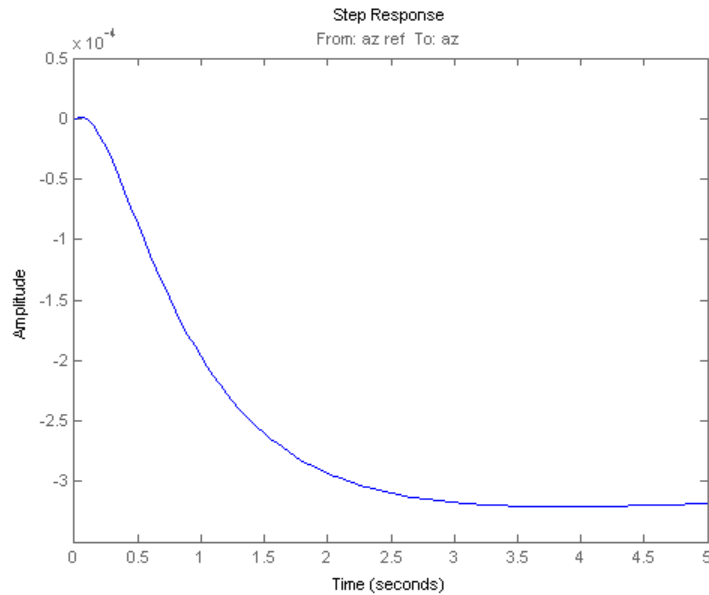
The final value of the success gauge `gam` is less than one, which means that all requirements were met. Confirm this by graphically validating the design.

`ST.loopview(Info)`



The first plot confirms that the open-loop response has integral action and the desired gain crossover frequency while the second plot shows that the MIMO stability margins are satisfactory (the blue curve should remain below the yellow bound). Next check the response from the step command `azref` to the vertical acceleration `az`:

```
T = ST.getIOTransfer('az ref','az');
clf
step(T,5)
```



The acceleration az does not track $azref$ despite the presence of an integrator in the loop. This is because the feedback loop acts on the two variables az and q and we have not specified which one should track $azref$.

Adding a Tracking Requirement

To remedy this issue, add an explicit requirement that az should follow the step command $azref$ with a 1 second response time. Also relax the gain crossover requirement to the interval $[3,12]$ to let the tuner find the appropriate gain crossover frequency.

```
rng(0)
TrackReq = TuningGoal.Tracking('az ref','az',1);
ST = ST0.looptune([3,12],TrackReq,Options);
```

```
Final: Peak gain = 62.1, Iterations = 38
```

```
Min decay rate 2.1e-07 is close to lower bound 1e-07
```

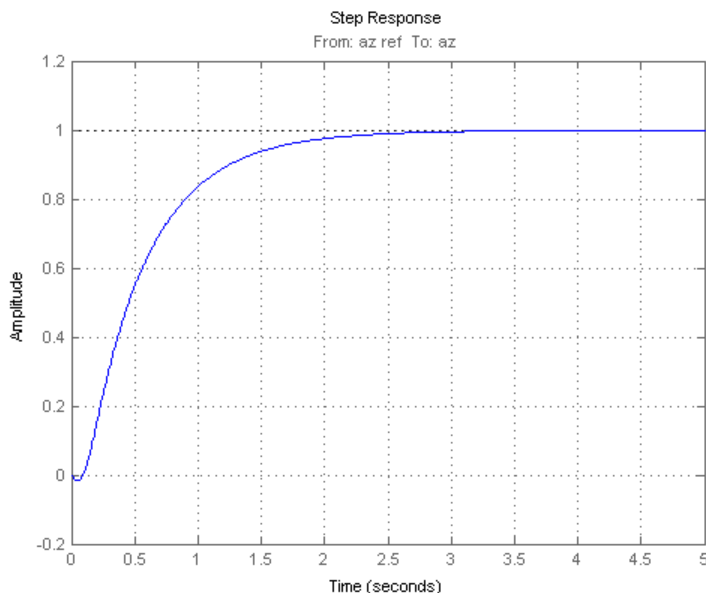
```
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
```

```
Final: Peak gain = 1.23, Iterations = 122
```

```
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Peak gain = 62.1, Iterations = 59
      Min decay rate 2.68e-07 is close to lower bound 1e-07
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0317)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0406)
Final: Failed to enforce closed-loop stability (spectral abscissa = 0.0405)
```

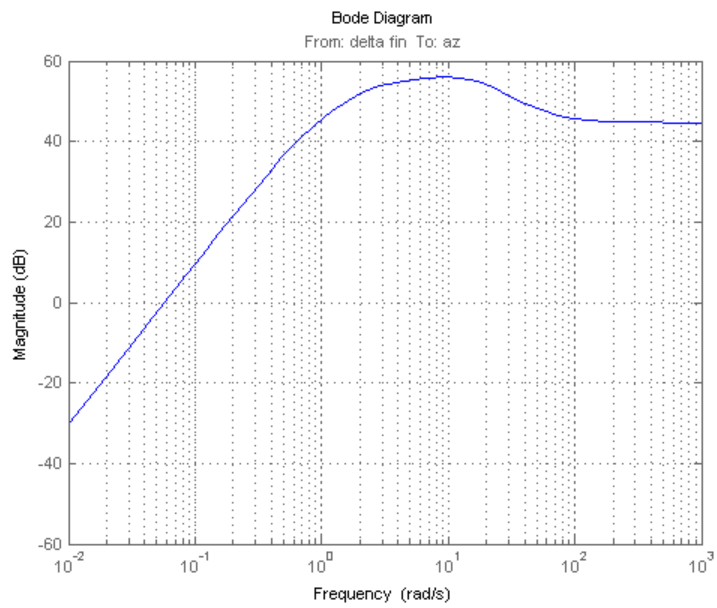
The step response from azref to az is now satisfactory:

```
Tr1 = ST.getIOTransfer('az ref','az');
step(Tr1,5), grid
```

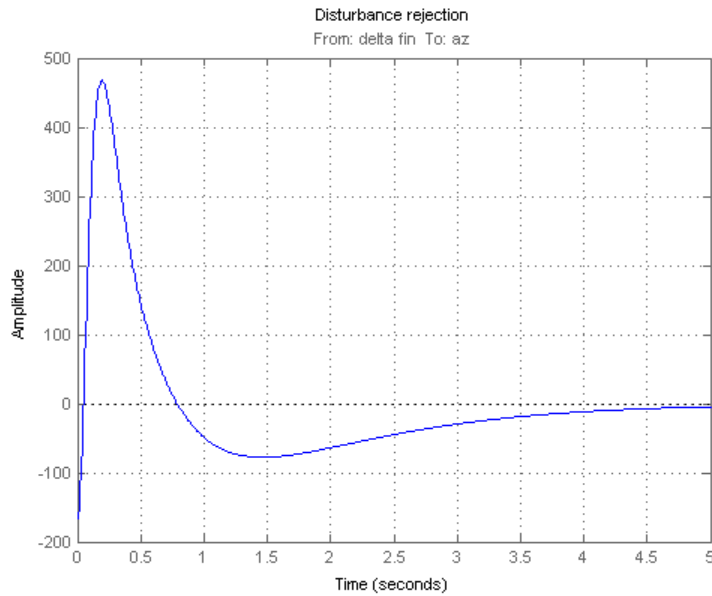


Also check the disturbance rejection characteristics by looking at the responses from a disturbance entering at the plant input

```
Td1 = ST.getIOTransfer('delta fin','az');
bodemag(Td1), grid
```

```
step(Td1,5), grid, title('Disturbance rejection')
```



Use `showBlockValue` to see the tuned values of the PI controller and inner-loop gain

`ST.showBlockValue()`

Block "rct_airframe1/az Control" =

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = 0.00165$, $K_i = 0.00166$

Name: az_Control

Continuous-time PI controller in parallel form.

```
-----  
Block "rct_airframe1/q Gain" =
```

```
Value =
```

```
    d =  
        u1  
    y1  1.985
```

```
Name: q_Gain  
Static gain.
```

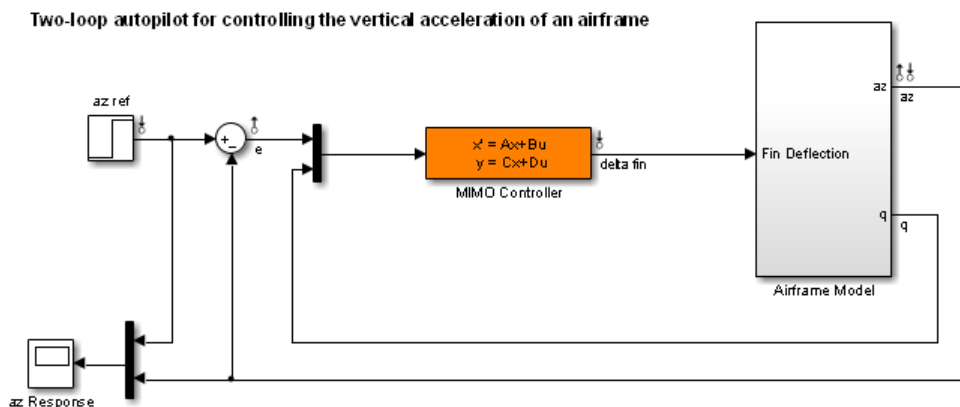
If this design is satisfactory, use `writeBlockValue` to apply the tuned values to the Simulink model and simulate the tuned controller in Simulink.

```
ST.writeBlockValue()
```

MIMO Design with SYSTUNE

Cascaded loops are commonly used for autopilots. Yet one may wonder how a single MIMO controller that uses both `az` and `q` to generate the actuator command `delta fin` would compare with the two-loop architecture. Trying new control architectures is easy with `systemtune` or `looptune`. For variety, we now use `systemtune` to tune the following MIMO architecture.

```
open_system('rct_airframe2')
```



As with `looptune`, first create an `sITunable` interface to facilitate tuning in Simulink.

```
STO = sITunable('rct_airframe2','MIMO Controller');
STO.addControl('delta fin');
```

Try a second-order MIMO controller with zero feedthrough from `e` to `delta fin`. To do this, create the desired controller parameterization and associate it with the "MIMO Controller" block using `setBlockParam`:

```
C0 = ltiblock.ss('C',2,1,2); % Second-order controller
C0.d.Value(1) = 0; C0.d.Free(1) = false; % Fix D(1) to zero
STO.setBlockParam('MIMO Controller',C0)
```

Next create adequate tuning requirements. Here we use the following four requirements:

- 1 Tracking:** `az` should respond in about 1 second to the `azref` command
- 2 Bandwidth and roll-off:** The loop gain at `delta fin` should roll off after 25 rad/s with a -20 dB/decade slope

3 Stability margins: The margins at $\delta \omega$ should exceed 7 dB and 45 degrees

4 Disturbance rejection: Limit the gain from $\delta \omega$ to az to ensure good disturbance rejection at low frequencies.

```
% Tracking
Req1 = TuningGoal.Tracking('az ref','az',1);

% Bandwidth and roll-off
Req2 = TuningGoal.Gain('delta fin','delta fin',tf(25,[1 0])); % roll-off

% Margins
Req3 = TuningGoal.Margins('delta fin',7,45);

% Disturbance rejection
% Use an FRD model to sketch the desired gain profile with a few points
Freqs = [0.02 2 200];
MaxGain = [2 200 200];
GainLimit = frd(MaxGain,Freqs);
Req4 = TuningGoal.Gain('delta fin','az',GainLimit);
```

You can now use `systemtune` to tune the controller parameters subject to these requirements.

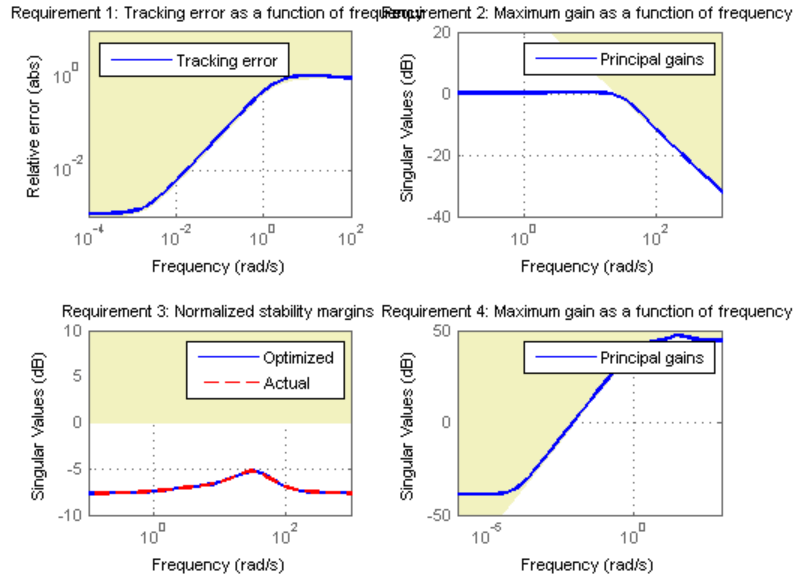
```
AllReqs = [Req1,Req2,Req3 Req4];
Opt = systemtuneOptions('RandomStart',3);

rng('default')
[ST,fSoft,~,Info] = ST0.systemtune(AllReqs,Opt);
```

```
Final: Soft = 1.5, Hard = -Inf, Iterations = 57
Final: Soft = 1.49, Hard = -Inf, Iterations = 93
Final: Soft = 1.15, Hard = -Inf, Iterations = 57
Final: Soft = 1.15, Hard = -Inf, Iterations = 82
```

The best design has an overall objective value close to 1, indicating that all four requirements are nearly met. Use `viewSpec` to inspect each requirement for the best design.

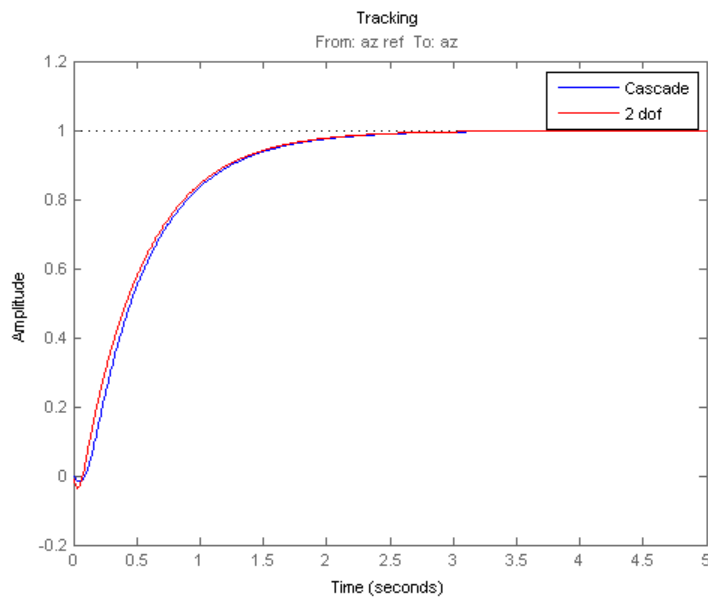
viewSpec(AllReqs,ST,Info)



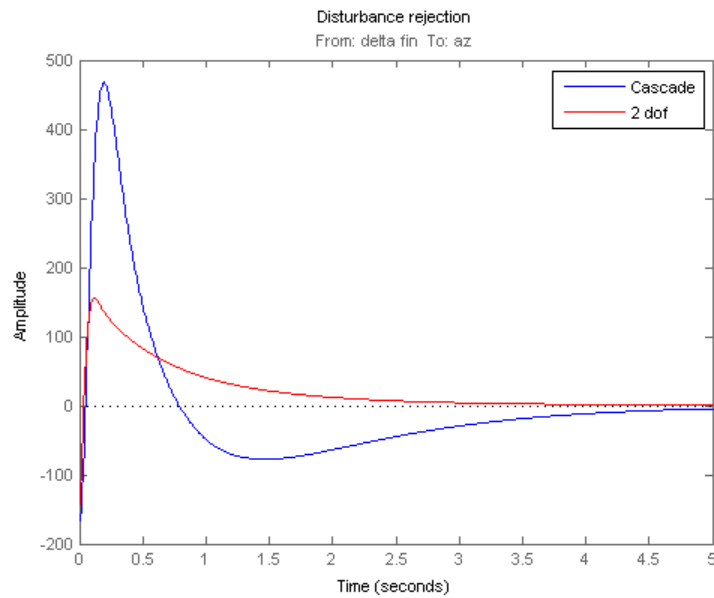
Compute the closed-loop responses and compare with the two-loop design.

```
T = ST.getIOTransfer({'az ref','delta fin'},'az');
```

```
clf
step(Tr1,'b',T(1),'r',5)
title('Tracking'), legend('Cascade','2 dof')
```



```
step(Td1,'b',T(2),'r',5)  
title('Disturbance rejection'), legend('Cascade','2 dof')
```



The tracking performance is similar but the second design has better disturbance rejection properties.

Control of a UAV Formation

This example uses `systune` to optimize a decentralized control law for a UAV formation. For more details, see *"A Model-Predictive Decentralized Control Scheme With Reduced Communication Requirement for Spacecraft Formation"*, J. Lavaei et al., IEEE Transactions of Control Systems Technology, 16(2), 2008.

Background

This example deals with the guidance of a leader-follower formation of 3 UAVs. The leader is UAV 1 and the followers are UAVs 2 and 3. The objective is to maintain prescribed velocities for all UAVs and prescribed distances along the x and y axes between UAVs 1 and 2, and UAVs 2 and 3. Each UAV has only access to partial information which leads to a decentralized control problem.



Model and Control Structure

We consider the motion of the formation in the horizontal (x,y) plane. The state of each UAV is described by:

- Leader (UAV 1): Two-entry vector δv_1 of speed errors of leader in x and y directions
- Followers (UAV 2 and UAV 3): Two-entry vectors $\delta d_2, \delta d_3$ of errors in x,y distances with predecessor and two-entry vectors $\delta v_2, \delta v_3$ of speed errors

Each UAV is controlled by a two-entry thrust vector $\delta u_i, i = 1, 2, 3$ (thrusts in x and y directions). The linearized dynamics for the entire formation are:

$$\dot{X} = AX + BU$$

where X is the overall state vector obtained by stacking $\delta v_1, \delta d_2, \delta v_2, \delta d_3, \delta v_3$ and U is the overall thrust vector obtained by stacking $\delta u_1, \delta u_2, \delta u_3$. Units are ft, ft/s, and ft/s².

The leader sets the target velocity for the entire formation and only worries about achieving this target velocity. Meanwhile, the followers must control both their velocity and their distance to their predecessor. Each follower only knows its own speed and its position relative to its predecessor, which leads to the decentralized control law:

$$\delta u_1 = K_1 \delta v_1, \quad \delta u_2 = K_2 \begin{pmatrix} \delta d_2 \\ \delta v_2 \end{pmatrix}, \quad \delta u_3 = K_2 \begin{pmatrix} \delta d_3 \\ \delta v_3 \end{pmatrix}$$

where K_1 is a 2-by-2 gain matrix and K_2 is a 2-by-4 gain matrix. Note that we use the same gain K_2 for both followers since they are interchangeable UAVs. Overall, this amounts to state-feedback control with the diagonal structure:

$$U = KX = \begin{pmatrix} K_1 & 0 & 0 \\ 0 & K_2 & 0 \\ 0 & 0 & K_2 \end{pmatrix} X.$$

Load the A, B matrices of the linearized dynamics and parameterized the controller gains.

```
load FormationFlightData
```

```
K1 = ltiblock.gain('K1', 2,2);
K2 = ltiblock.gain('K2', 2,4);
```

The closed-loop system has equations

$$\dot{X} = AX + BU + W, \quad U = KX$$

and is depicted in Figure 1. The input vector W models the speed and velocity disturbances.

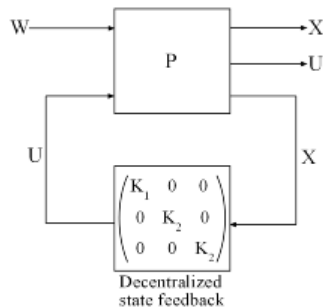


Figure 1: Closed-Loop System

Construct a tunable model $T0$ of the closed-loop system. This model depends on the tunable gains K_1, K_2 . Add a loopswitch block at the plant input U so that you can specify and analyze the gain and phase margins of the feedback loop.

```
% Plant P
nx = size(A,1); % number of states
nu = size(B,2); % number of controls
BP = [eye(nx) , B];
CP = [eye(nx) ; zeros(nu,nx) ; eye(nx)];
DP = [zeros(nx,nx+nu) ; zeros(nu,nx) eye(nu) ; zeros(nx,nx+nu)];
P = ss(A, BP, CP, DP);

% Loop switch for open-loop analysis/requirement
LSU = loopswitch('U',nu);
```

```
% Tunable closed-loop model
T0 = lft( P , LSU * blkdiag(K1,K2,K2) );
T0.InputName = 'W';
T0.OutputName = {'X(1)' 'X(2)' 'X(3)' 'X(4)' 'X(5)' 'X(6)' 'X(7)'...
                 'X(8)' 'X(9)' 'X(10)' 'U(1)' 'U(2)' 'U(3)' 'U(4)' 'U(5)' 'U(6)'};
```

Design Requirements

The feedback gains K_1, K_2 must be tuned to ensure:

- 1 Stability of the formation flight
- 2 Appropriate decay of the distances and speeds to their desired values. The reference is zero here since we are working with error dynamics.
- 3 Reasonable thrust levels
- 4 Adequate gain and phase margins at the plant inputs.

Use a quadratic LQG-like cost of the form

$$J = \lim_{T \rightarrow \infty} E \left(\frac{1}{T} \int_0^T (X^T Q X + U^T R U) dt \right)$$

to capture the first three requirements. The value J^2 is just the variance of the weighted output signal

$$Z = \begin{pmatrix} Q^{1/2} X \\ R^{1/2} U \end{pmatrix}$$

for a unit-variance white noise input W (see Figure 1). Use $Q = I$ and $R = I$ for the initial tuning.

```
Q = eye(nx);
R = eye(nu);
Req1 = TuningGoal.WeightedVariance('W',T0.OutputName,...
                                   blkdiag(sqrtm(Q),sqrtm(R)),[]);
```

For the last requirement, specify a minimum of 10 dB and 45 degrees of gain and phase margins at the plant inputs.

```
Req2 = TuningGoal.Margins('U',10,45);
```

Tuning of Decentralized Controller

Use `systune` to tune the gains K_1, K_2 subject to the two requirements defined above.

```
rng('default')
T = systune(T0,Req1,Req2,systuneOptions('RandomStart',2));

Final: Failed to enforce closed-loop stability (spectral abscissa = 0)
Final: Soft = 4.42, Hard = 0.84135, Iterations = 63
Final: Soft = 4.42, Hard = 0.77999, Iterations = 48
```

The stability margins requirement (hard constraint) is satisfied since its value is less than 1. The optimized value of J is about 4.4. Confirm the stability margins by computing the negative-feedback open-loop response L at the plant inputs and using `loopmargin` to compute the corresponding disk margins.

```
L = getLoopTransfer(T,'U',-1); % negative-feedback loop transfer
[~,~,MM] = loopmargin(L);
```

```
% Gain margins
mag2db(MM.GainMargin)
```

```
ans =

    -14.6551    14.6551
```

```
% Phase margins
MM.PhaseMargin
```

```
ans =

    -69.0341    69.0341
```

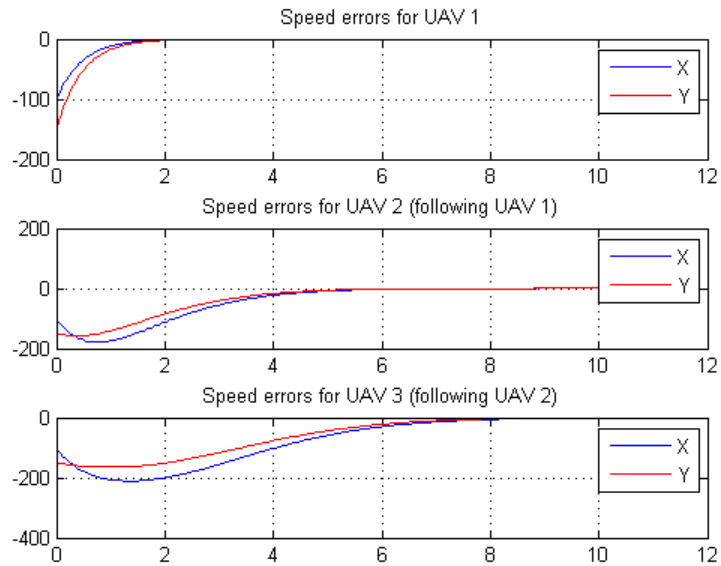
The loopmargin results indicate that the tuned control system can sustain concurrent gain and phase variations in excess of ± 10 dB and ± 60 degrees in all feedback channels U without going unstable.

Closed-Loop Simulation

Simulate how the UAV formation responds to initial errors of 100 ft/s in x-speed, 150 ft/s in y-speed, 400 ft in x-distance, and 300 ft in y-distance.

```
X0 = -[100 150 400 300 100 150 400 300 100 150]'; % initial errors
[X,t] = initial(T(1:nx,:),X0);

% Plot speed errors for leader and followers
clf
subplot(311), plot(t, X(:,1), 'b', t, X(:,2), 'r'), grid
title('Speed errors for UAV 1'); legend('X','Y')
subplot(312), plot(t, X(:,5), 'b', t, X(:,6), 'r'), grid
title('Speed errors for UAV 2 (following UAV 1)'); legend('X','Y')
subplot(313), plot(t, X(:,9), 'b', t, X(:,10), 'r'), grid
title('Speed errors for UAV 3 (following UAV 2)'); legend('X','Y')
```



```
% Plot distance errors
```

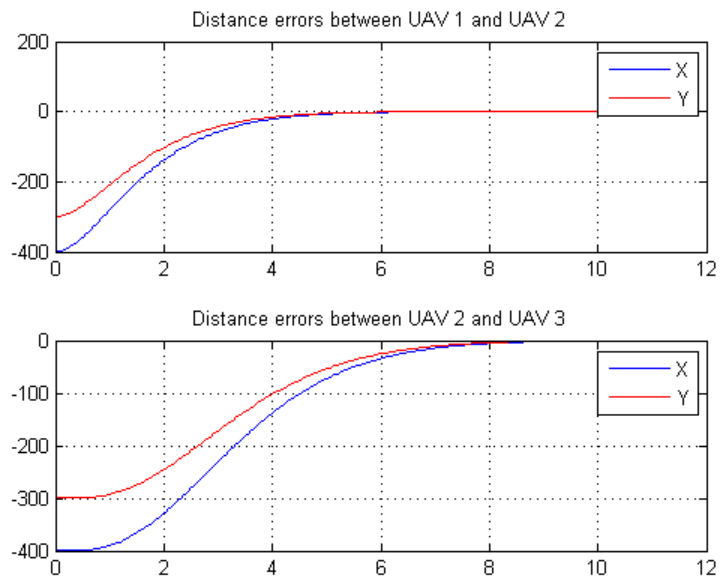
```
clf
```

```
subplot(211), plot(t, X(:,3), 'b', t, X(:,4), 'r'), grid
```

```
title('Distance errors between UAV 1 and UAV 2'); legend('X','Y')
```

```
subplot(212), plot(t, X(:,7), 'b', t, X(:,8), 'r'), grid
```

```
title('Distance errors between UAV 2 and UAV 3'); legend('X','Y')
```



Note how the follower UAVs initially accelerate to reduce the distance errors and then slow down to keep a constant distance with their predecessor.

Multi-Loop Controller for the Westland Lynx Helicopter

This example shows how to use Robust Control Toolbox™ to tune a multi-loop controller for a rotorcraft. We thank Professor Pierre Apkarian from ONERA for providing the example material.

Helicopter Model

This example uses an 8-state model of the Westland Lynx helicopter at the hovering trim condition. The state vector $x = [u, w, q, \text{theta}, v, p, \text{phi}, r]$ consists of

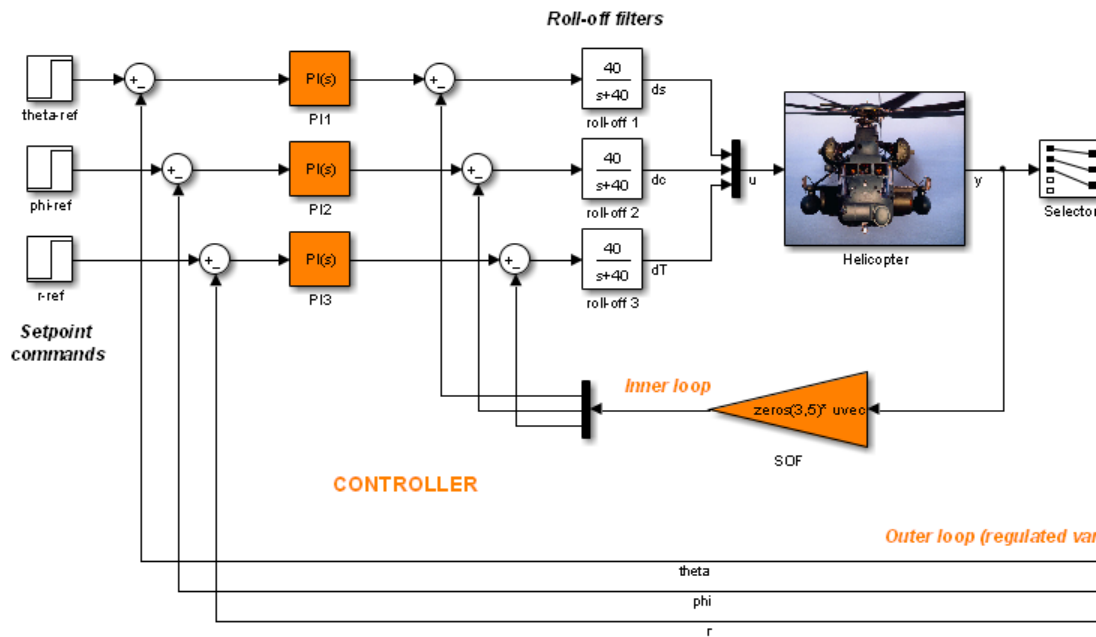
- Longitudinal velocity u (m/s)
- Lateral velocity v (m/s)
- Normal velocity w (m/s)
- Pitch angle theta (deg)
- Roll angle phi (deg)
- Roll rate p (deg/s)
- Pitch rate q (deg/s)
- Yaw rate r (deg/s).

The controller generates commands d_s, d_c, d_T in degrees for the longitudinal cyclic, lateral cyclic, and tail rotor collective using measurements of theta , phi , p , q , and r . For a detailed description of the model, see "Helicopter H_∞ Control Design with Robust Flying Quality," *Aerospace Science & Technology*, 7 (2003), pp 159-169.

Control Architecture

The following Simulink model depicts the control architecture:

```
open_system('rct_helico')
```



The control system consists of two feedback loops. The inner loop (static output feedback) provides stability augmentation and decoupling. The outer loop (PI controllers) provides the desired setpoint tracking performance. The main control objectives are as follows:

- Track setpoint changes in θ , ϕ , and r with zero steady-state error, settling times of about 2 seconds, minimal overshoot, and minimal cross-coupling
- Limit the control bandwidth to guard against neglected high-frequency rotor dynamics and measurement noise
- Provide strong multivariable gain and phase margins (robustness to simultaneous gain/phase variations at the plant inputs and outputs, see `loopmargin` for details).

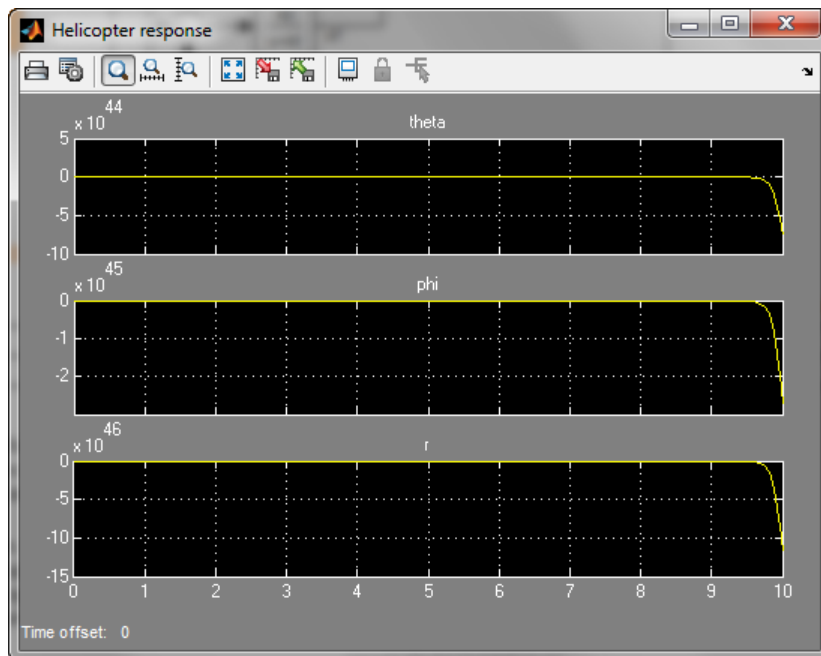
We use lowpass filters with cutoff at 40 rad/s to partially enforce the second objective.

Controller Tuning

You can jointly tune the inner and outer loops with the `looptune` command. This command only requires models of the plant and controller along with the desired bandwidth (which is function of the desired response time). When the control system is modeled in Simulink, you can use the `sITunable` interface to quickly set up the tuning task. Create an instance of this interface with the list of blocks to be tuned.

```
STO = sITunable('rct_helico',{ 'PI1', 'PI2', 'PI3', 'SOF' });
```

Each tunable block is automatically parameterized according to its type and initialized with its value in the Simulink model ($1 + 1/s$ for the PI controllers and zero for the static output-feedback gain). Simulating the model shows that the control system is unstable for these initial values:



Specify the I/O signals of interest for setpoint tracking.

```
STO.addIO({'theta-ref','phi-ref','r-ref'},'in') % setpoint commands
STO.addIO({'theta','phi','r'},'out') % corresponding outputs
```

Also identify the plant inputs and outputs (control and measurement signals) where the stability margin are measured.

```
STO.addControl('u');
STO.addMeasurement('y');
```

Finally, capture the design requirements using `TuningGoal` objects. We use the following requirements for this example:

- **Tracking requirement:** The outputs `theta`, `phi`, `r` must track the commands `theta_ref`, `phi_ref`, `r_ref` with a 2-second response time
- **Stability margins:** The multivariable gain and phase margins at the plant inputs `u` and plant outputs `y` must be at least 5 dB and 40 degrees
- **Fast dynamics:** The magnitude of the closed-loop poles must not exceed 20 to prevent fast dynamics and jerky transients

```
% Tracking
TrackReq = TuningGoal.Tracking({'theta-ref','phi-ref','r-ref'},...
    {'theta','phi','r'},2,1e-2);

% Gain and phase margins at plant inputs and outputs
MarginReq1 = TuningGoal.Margins('u',5,40);
MarginReq2 = TuningGoal.Margins('y',5,40);

% Limit on fast dynamics
PoleReq = TuningGoal.Poles();
PoleReq.MaxFrequency = 20;
```

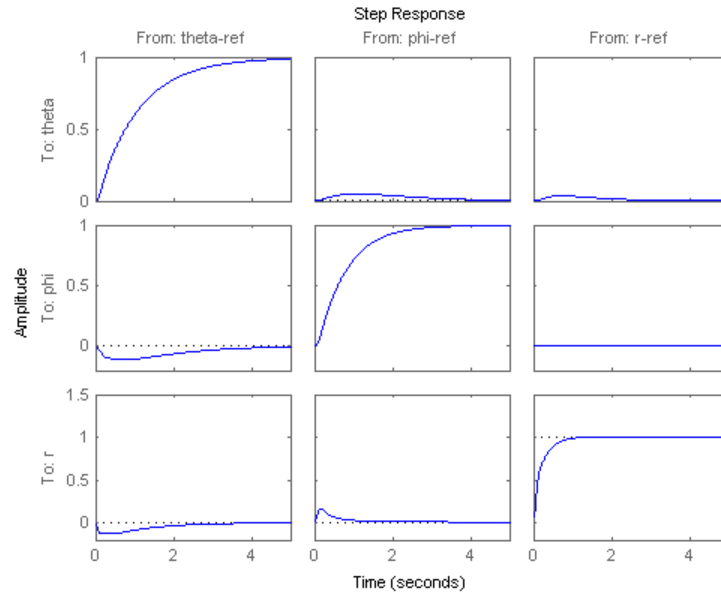
You can now use `systemtune` to jointly tune all controller parameters. This returns the tuned version `ST1` of the control system `STO`.

```
AllReqs = [TrackReq,MarginReq1,MarginReq2,PoleReq];
[ST1,fSoft,~,Info] = STO.systemtune(AllReqs);
```

```
Final: Soft = 1.15, Hard = -Inf, Iterations = 49
```

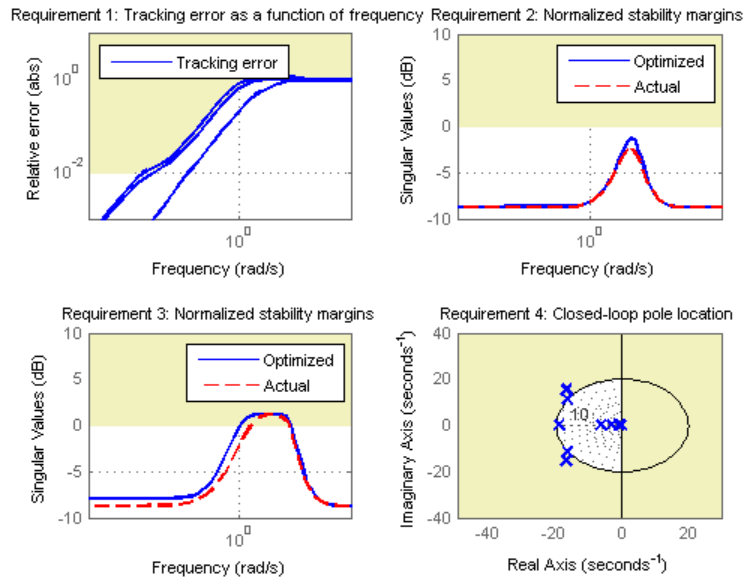
The final value is close to 1 so the requirements are nearly met. Plot the tuned responses to step commands in theta, phi, r:

```
T1 = ST1.getIOTransfer({'theta-ref','phi-ref','r-ref'},{'theta','phi','r'})
step(T1,5)
```



The response time is about two seconds with no overshoot and little cross-coupling. You can use `viewSpec` for a more thorough validation of each requirement, including a visual assessment of the multivariable stability margins (see `loopmargin` for details):

```
viewSpec(AllReqs,ST1,Info)
```



Inspect the tuned values of the PI controllers and static output-feedback gain.

```
ST1.showTunable()
```

```
Block "rct_helico/PI1" =
```

```
Value =
```

$$K_p + K_i \frac{1}{s}$$

```
with Kp = 0.504, Ki = 1.92
```

```
Name: PI1
```

```
Continuous-time PI controller in parallel form.
```

```
-----
```

Block "rct_helico/PI2" =

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = -0.0557$, $K_i = -1.88$

Name: PI2

Continuous-time PI controller in parallel form.

Block "rct_helico/PI3" =

Value =

$$K_p + K_i * \frac{1}{s}$$

with $K_p = -0.515$, $K_i = -4.69$

Name: PI3

Continuous-time PI controller in parallel form.

Block "rct_helico/SOF" =

Value =

$$d = \begin{array}{rcccccc} & u1 & u2 & u3 & u4 & u5 \\ y1 & 2.1 & -0.2376 & -0.09408 & 0.5177 & 0.003496 \\ y2 & -0.4855 & -1.514 & 0.008369 & -0.07761 & -0.1078 \\ y3 & -0.9372 & 0.48 & -0.9289 & -0.07711 & 0.2312 \end{array}$$

Name: SOF
Static gain.

Benefit of the Inner Loop

You may wonder whether the static output feedback is necessary and whether PID controllers aren't enough to control the helicopter. This question is easily answered by re-tuning the controller with the inner loop open. First break the inner loop by adding a loop opening after the SOF block:

```
STO.addOpening('SOF')
```

Then remove the SOF block from the tunable block list and re-parameterize the PI blocks as full-blown PIDs with the correct loop signs (as inferred from the first design).

```
PID = pid(0,0.001,0.001,.01); % initial guess for PID controllers
```

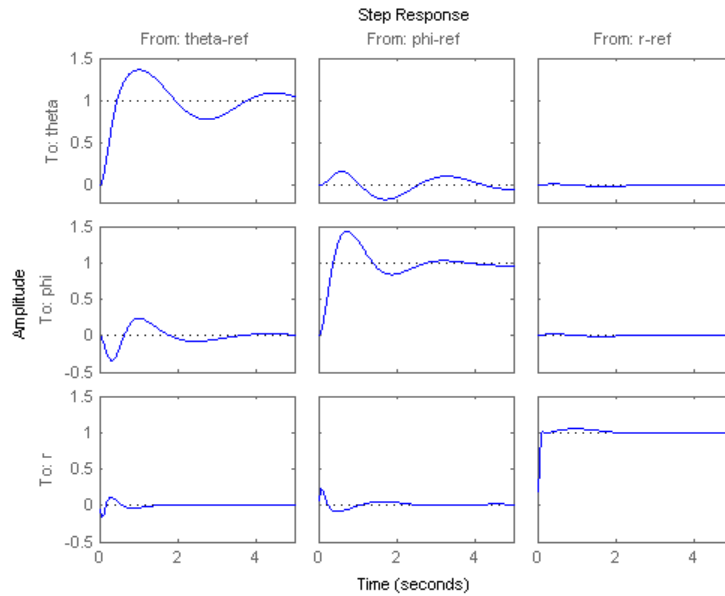
```
STO.TunedBlocks = STO.TunedBlocks(1:3);  
STO.setBlockParam('PI1',ltiblock.pid('C1',PID));  
STO.setBlockParam('PI2',ltiblock.pid('C2',-PID));  
STO.setBlockParam('PI3',ltiblock.pid('C3',-PID));
```

Re-tune the three PID controllers and plot the closed-loop step responses.

```
[ST2,fSoft,~,Info] = STO.systune(AllReqs);
```

```
Final: Soft = 2.07, Hard = -Inf, Iterations = 95
```

```
T2 = ST2.getIOTransfer({'theta-ref','phi-ref','r-ref'},{'theta','phi','r'})  
clf, step(T2,5)
```

The final value is no longer close to 1 and the step responses confirm some performance degradation with regard to settling time, overshoot, and decoupling. This suggests that the inner loop has an important stabilizing effect that should be preserved.

Fixed-Structure Autopilot for a Passenger Jet

This example shows how to use Robust Control Toolbox™ to tune the standard configuration of a longitudinal autopilot. We thank Professor D. Alazard from Institut Supérieur de l'Aéronautique et de l'Espace for providing the aircraft model and Professor Pierre Apkarian from ONERA for developing the example.

Aircraft Model and Autopilot Configuration

The longitudinal autopilot for a supersonic passenger jet flying at Mach 0.7 and 5000 ft is depicted in Figure 1. The autopilot main purpose is to follow vertical acceleration commands N_{ze} issued by the pilot. The feedback structure consists of an inner loop controlling the pitch rate q and an outer loop controlling the vertical acceleration N_z . The autopilot also includes a feedforward component and a reference model $G_{ref}(s)$ that specifies the desired response to a step command N_{ze} . Finally, the second-order roll-off filter

$$F_{ro}(s) = \frac{\omega_n^2}{s^2 + 2\zeta\omega_n s + \omega_n^2}$$

is used to attenuate noise and limit the control bandwidth as a safeguard against unmodeled dynamics. The tunable components are highlighted in orange.

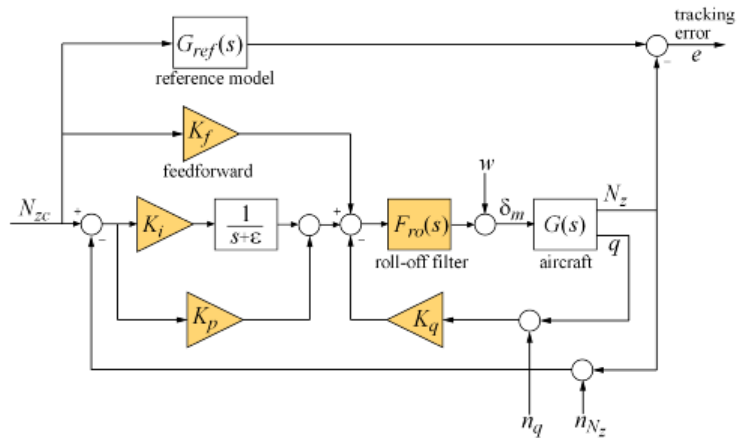
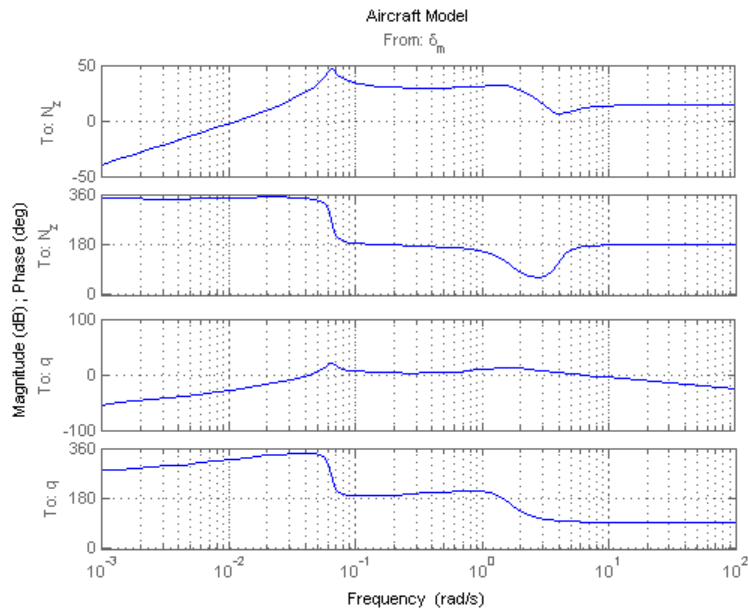


Figure 1: Longitudinal Autopilot Configuration.

The aircraft model $G(s)$ is a 5-state model, the state variables being the aerodynamic speed V (m/s), the climb angle γ (rad), the angle of attack α (rad), the pitch rate q (rad/s), and the altitude H (m). The elevator deflection δ_m (rad) is used to control the vertical load factor N_z . The open-loop dynamics include the α oscillation with frequency and damping ratio $\omega_n = 1.7$ (rad/s) and $\zeta = 0.33$, the phugoid mode $\omega_n = 0.64$ (rad/s) and $\zeta = 0.06$, and the slow altitude mode $\lambda = -0.0026$.

```
load ConcordeData G
bode(G,{1e-3,1e2}), grid
title('Aircraft Model')
```

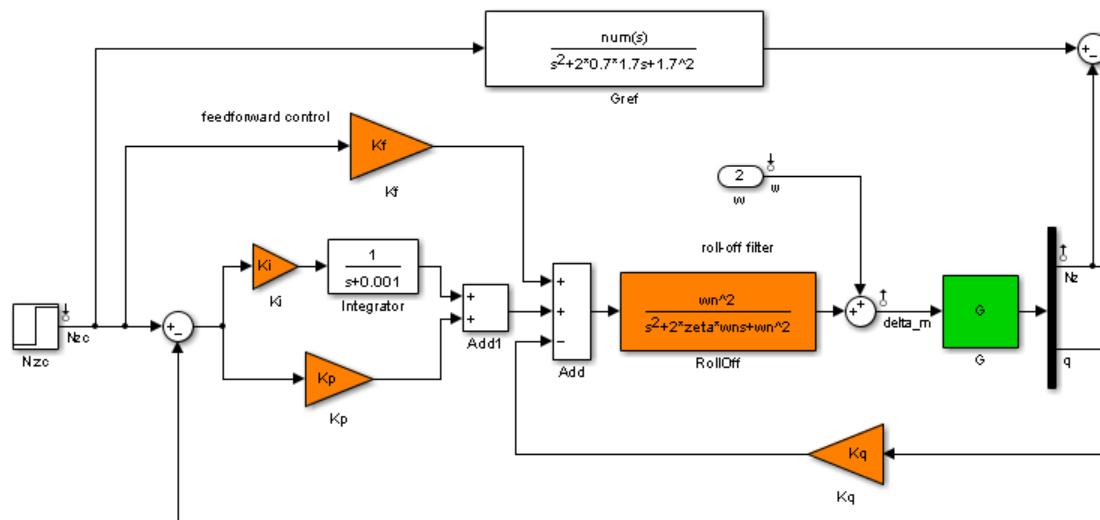


Note the zero at the origin in $G(s)$. Because of this zero, we cannot achieve zero steady-state error and must instead focus on the transient response to acceleration commands. Note that acceleration commands are transient in nature so steady-state behavior is not a concern. This zero at the origin also precludes pure integral action so we use a pseudo-integrator $1/(s + \epsilon)$ with $\epsilon = 0.001$.

Tuning Setup

When the control system is modeled in Simulink, you can use the `sITunable` interface to quickly set up the tuning task. Open the Simulink model of the autopilot.

```
open_system('rct_concorde')
```



Longitudinal autopilot for a passenger jet.

You can tune this autopilot with the SYSTUNE command, see `concorde_demo` for details

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Configure the `sITunable` interface by listing the tuned blocks in the Simulink model (highlighted in orange).

```
ST0 = sITunable('rct_concorde', {'Ki', 'Kp', 'Kq', 'Kf', 'RollOff'});
```

This automatically picks all linearization points in the model as analysis I/Os.

ST0.IOs

ans =

```
'rct_concorde/Nzc/1[Nzc]'      'in'
'rct_concorde/noise/1[n]'     'in'
'rct_concorde/w/1[w]'        'in'
'rct_concorde/Demux/1[Nz]'    'out'
'rct_concorde/Sum2/1[e]'     'out'
```

```
'rct_concorde/Sum5/1[delta_m]'      'out'
```

This also parameterizes each tuned block and initializes the block parameters based on their values in the Simulink model. Note that the four gains K_i, K_p, K_q, K_f are initialized to zero in this example. By default the roll-off filter $F_{ro}(s)$ is parameterized as a generic second-order transfer function. To parameterize it as

$$F_{ro}(s) = \frac{\omega_n^2}{s^2 + 2\zeta\omega_n s + \omega_n^2},$$

create real parameters ζ, ω_n , build the transfer function shown above, and associate it with the RollOff block.

```
wn = realp('wn', 3);           % natural frequency
zeta = realp('zeta', 0.8);     % damping
Fro = tf(wn^2, [1 2*zeta*wn wn^2]); % parametric transfer function
```

```
STO.setBlockParam('RollOff', Fro) % use Fro to parameterize "RollOff" block
```

Design Requirements

The autopilot must be tuned to satisfy three main design requirements:

1. **Setpoint tracking:** The response N_z to the command N_{zc} should closely match the response of the reference model:

$$G_{ref}(s) = \frac{1.7^2}{s^2 + 2 \times 0.7 \times 1.7s + 1.7^2}.$$

This reference model specifies a well-damped response with a 2 second settling time.

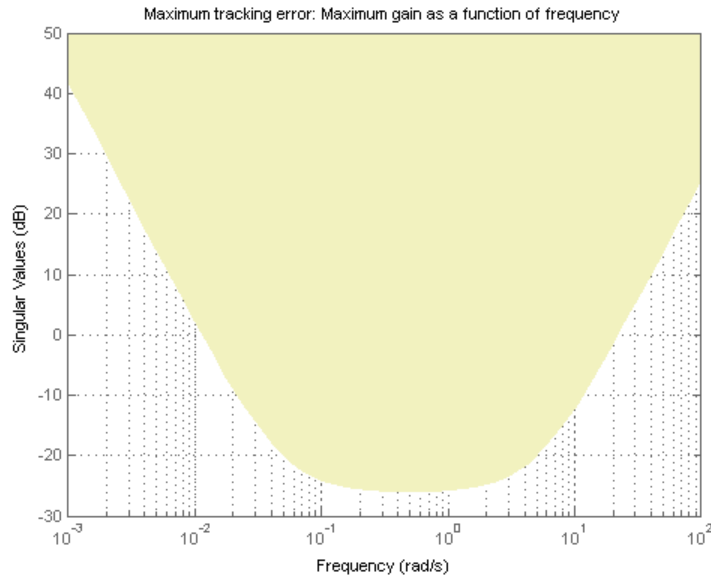
2. **High-frequency roll-off:** The closed-loop response from the noise signals to δ_m should roll off past 8 rad/s with a slope of at least -40 dB/decade.
3. **Stability margins:** The stability margins at the plant input δ_m should be at least 7 dB and 45 degrees.

For setpoint tracking, we require that the gain of the closed-loop transfer from the command N_{zc} to the tracking error e be small in the frequency band $[0.05, 5]$ rad/s (recall that we cannot drive the steady-state error to zero because of the plant zero at $s=0$). Using a few frequency points, sketch the maximum tracking error as a function of frequency and use it to limit the gain from N_{zc} to e .

```
Freqs = [0.005 0.05 5 50];
Gains = [5 0.05 0.05 5];
Req1 = TuningGoal.Gain('Nzc', 'e', frd(Gains, Freqs));
Req1.Name = 'Maximum tracking error';
```

The `TuningGoal.Gain` constructor automatically turns the maximum error sketch into a smooth weighting function. Use `viewSpec` to graphically verify the desired error profile.

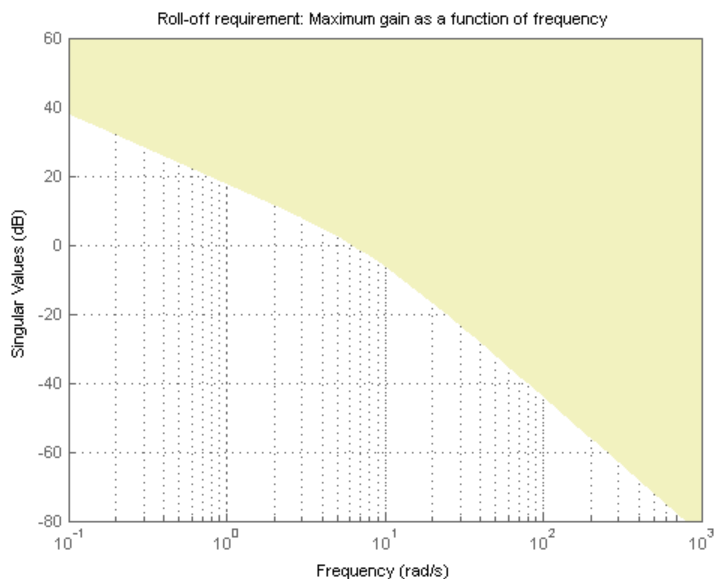
```
viewSpec(Req1)
```



Repeat the same process to limit the high-frequency gain from the noise inputs to δ_m and enforce a -40 dB/decade slope in the frequency band from 8 to 800 rad/s

```
Freqs = [0.8 8 800];
Gains = [10 1 1e-4];
Req2 = TuningGoal.Gain('n','delta_m',frd(Gains,Freqs));
Req2.Name = 'Roll-off requirement';
```

```
viewSpec(Req2)
```



Finally, register the plant input δ_m as a site for open-loop analysis and use `TuningGoal.Margins` to capture the stability margin requirement.

```
ST0.addSwitch('delta_m')

Req3 = TuningGoal.Margins('delta_m',7,45);
```

Autopilot Tuning

We are now ready to tune the autopilot parameters with `sysune`. This command takes the untuned configuration `ST0` and the three design requirements and returns the tuned version `ST` of `ST0`. All requirements are satisfied when the final value is less than one.

```
[ST,fSoft,~,Info] = sysune(ST0,[Req1 Req2 Req3]);
```

```
Final: Soft = 0.965, Hard = -Inf, Iterations = 49
```

Use `showTunable` to see the tuned block values.

```
ST.showTunable()
```

```
Block "rct_concorde/Ki" =
```

```
Value =
```

```

d =
      u1
y1  -0.03008
```

```
Name: Ki
Static gain.
```

```
-----
Block "rct_concorde/Kp" =
```

```
Value =
```

```

d =
      u1
y1  -0.009661
```

```
Name: Kp
Static gain.
```

Block "rct_concorde/Kq" =

Value =

$$d = \begin{matrix} & u1 \\ y1 & -0.2889 \end{matrix}$$

Name: Kq
Static gain.

Block "rct_concorde/Kf" =

Value =

$$d = \begin{matrix} & u1 \\ y1 & -0.02291 \end{matrix}$$

Name: Kf
Static gain.

Block "rct_concorde/RollOff" =

Value =

$$a = \begin{matrix} & x1 & x2 \\ x1 & -4.996 & -23.31 \\ x2 & 1 & 0 \end{matrix}$$
$$b = \begin{matrix} & u1 \\ x1 & 1 \\ x2 & 0 \end{matrix}$$

```

c =
      x1      x2
y1      0 23.31

d =
      u1
y1      0

```

Continuous-time state-space model.

To get the tuned value of $F_{ro}(s)$, use `getBlockValue` to evaluate `Fro` for the tuned parameter values in `ST`:

```

Fro = ST.getBlockValue('RollOff');
tf(Fro)

```

```

ans =

      23.31
-----
s^2 + 4.996 s + 23.31

```

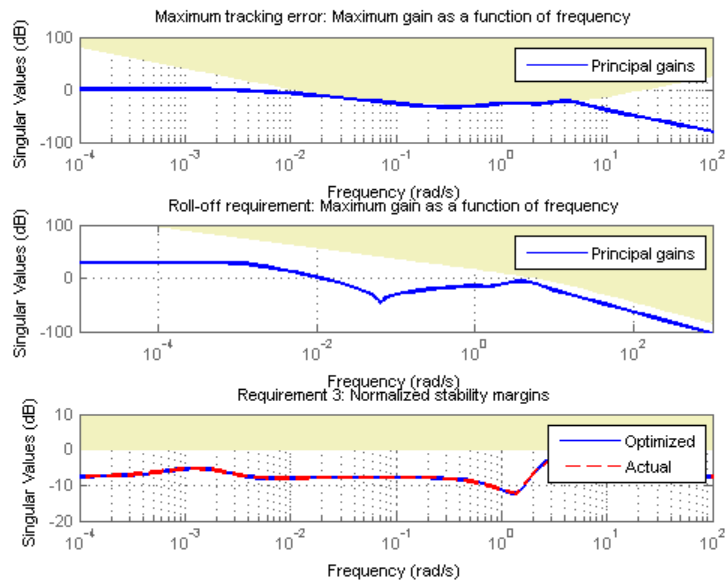
Continuous-time transfer function.

Finally, use `viewSpec` to graphically verify that all requirements are satisfied.

```

viewSpec([Req1 Req2 Req3],ST)

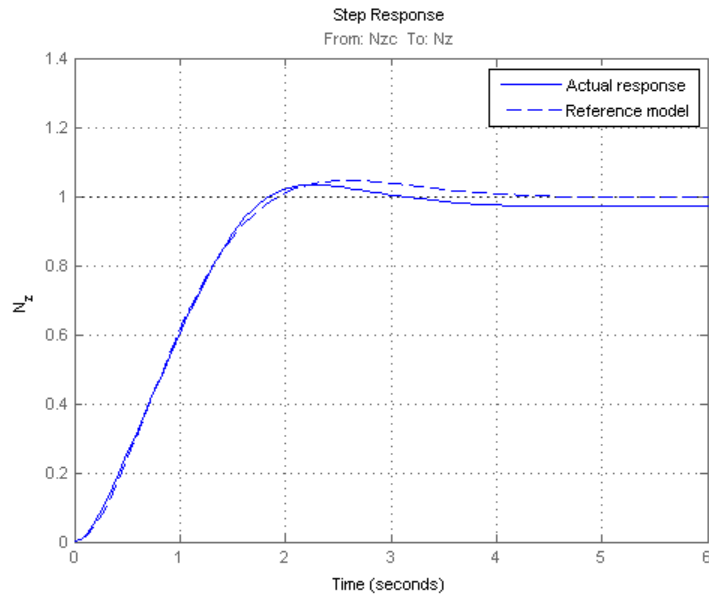
```



Closed-Loop Simulations

We now verify that the tuned autopilot satisfies the design requirements. First compare the step response of N_z with the step response of the reference model $G_{ref}(s)$. Again use `getIOTransfer` to compute the tuned closed-loop transfer from N_{zc} to N_z :

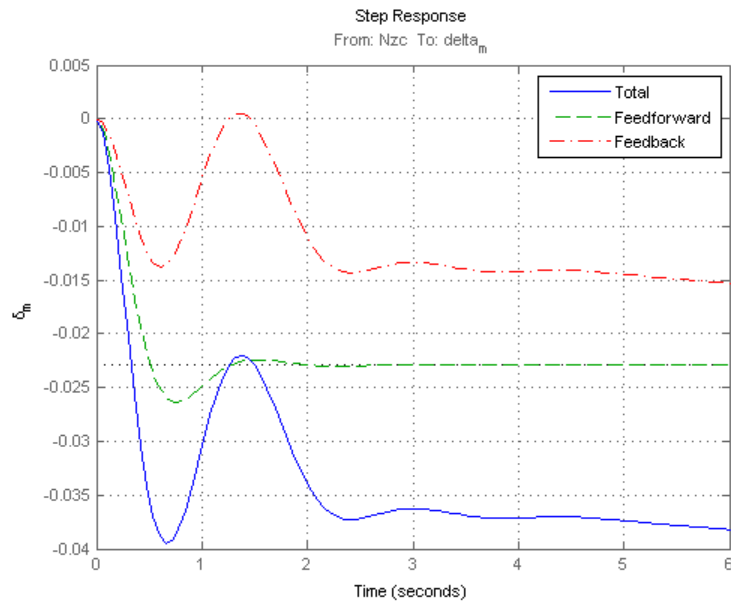
```
Gref = tf(1.7^2,[1 2*0.7*1.7 1.7^2]); % reference model
T = ST.getIOTransfer('Nzc','Nz'); % transfer Nzc -> Nz
clf, step(T,'b',Gref,'b--',6), grid,
ylabel('N_z'), legend('Actual response','Reference model')
```



Also plot the deflection δ_m and the respective contributions of the feedforward and feedback paths:

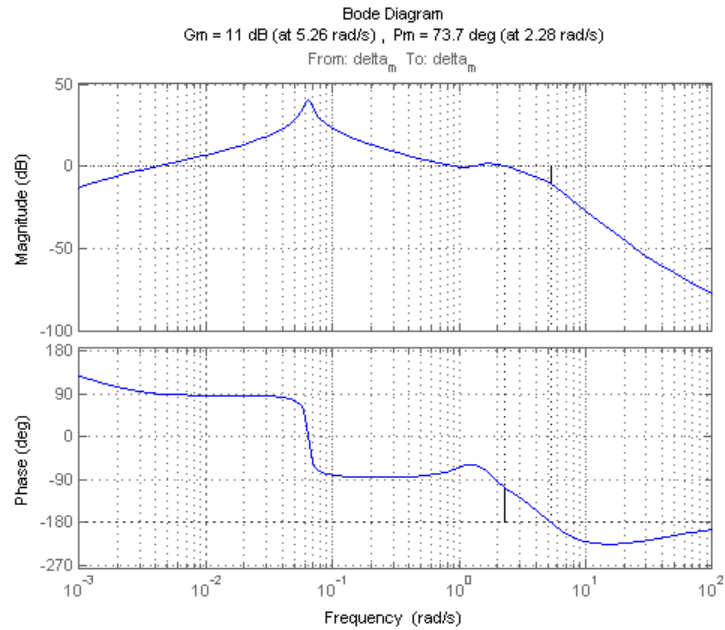
```
T = ST.getIOTransfer('Nzc','delta_m'); % transfer Nz_c -> delta_m
Kf = ST.getBlockValue('Kf');          % tuned value of Kf
Tff = Fro*Kf;                        % feedforward contribution to delta_m
```

```
step(T,'b',Tff,'g--',T-Tff,'r-.',6), grid
ylabel('\delta_m'), legend('Total','Feedforward','Feedback')
```



Finally, check the roll-off and stability margin requirements by computing the open-loop response at δ_m .

```
OL = ST.getLoopTransfer('delta_m',-1); % negative-feedback loop transfer
margin(OL); grid; set(gca,'XLim',[1e-3,1e2])
```



The Bode plot confirms a roll-off of -40 dB/decade past 8 rad/s and indicates gain and phase margins in excess of 10 dB and 70 degrees.

Reliable Control of a Fighter Jet

This example shows how to tune a fixed-structure controller for multiple operating modes of the plant.

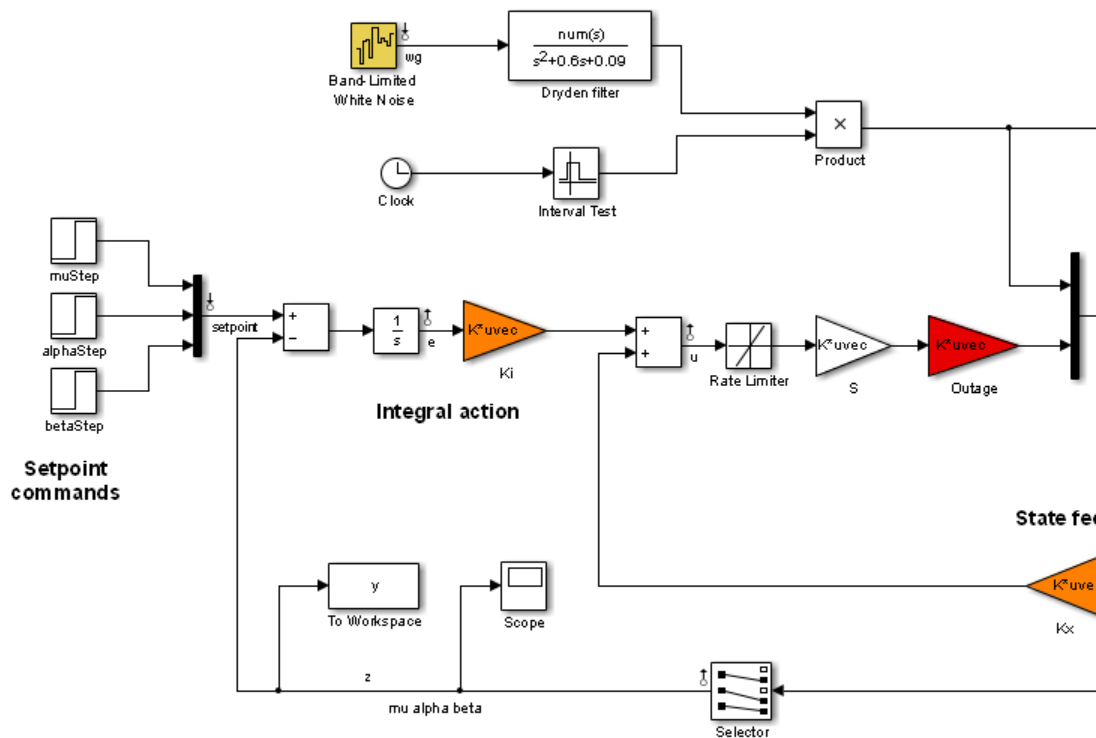
Background

This example deals with reliable flight control of an F16 aircraft undergoing outages in the elevator and aileron actuators. The flight control system is required to maintain stability and adequate performance in both nominal operation and degraded conditions where some actuators are no longer effective due to control surface impairment. Wind gusts must be alleviated in all conditions. This application is often called *reliable control* as aircraft safety must be maintained in extreme flight conditions.

Aircraft Model

The control system is modeled in Simulink.

```
open_system('rct_reliableF16')
```

The aircraft is modeled as a 6th-order state-space system with the following state variables (units are m/s for velocities and deg/s for angular rates):

- u : x-body axis velocity
- w : z-body axis velocity
- q : pitch rate
- v : y-body axis velocity
- p : roll rate
- r : yaw rate

The state vector is available for control as well as the flight-path bank angle rate μ (deg/s), the angle of attack α (deg), and the sideslip angle β

(deg). The control inputs are the deflections of the right elevator, left elevator, right aileron, left aileron, and rudder. All deflections are in degrees. Elevators are grouped symmetrically to generate the angle of attack. Ailerons are grouped anti-symmetrically to generate roll motion. This leads to 3 control actions as shown in the Simulink model.

The controller consists of state-feedback control in the inner loop and MIMO integral action in the outer loop. The gain matrices K_i and K_x are 3-by-3 and 3-by-6, respectively, so the controller has 27 tunable parameters.

Actuator Failures

We use a 9x5 matrix to encode the nominal mode and various actuator failure modes. Each row corresponds to one flight condition, a zero indicating outage of the corresponding deflection surface.

```
OutageCases = [...  
    1 1 1 1 1; ... % nominal operational mode  
    0 1 1 1 1; ... % right elevator outage  
    1 0 1 1 1; ... % left elevator outage  
    1 1 0 1 1; ... % right aileron outage  
    1 1 1 0 1; ... % left aileron outage  
    1 0 0 1 1; ... % left elevator and right aileron outage  
    0 1 0 1 1; ... % right elevator and right aileron outage  
    0 1 1 0 1; ... % right elevator and left aileron outage  
    1 0 1 0 1; ... % left elevator and left aileron outage  
];
```

Design Requirements

The controller should:

- 1 Provide good tracking performance in μ , α , and β in nominal operating mode with adequate decoupling of the three axes
- 2 Maintain performance in the presence of wind gust of 5 m/s
- 3 Limit stability and performance degradation in the face of actuator outage.

To express the first requirement, you can use an LQG-like cost function that penalizes the integrated tracking error e and the control effort u :

$$J = \lim_{T \rightarrow \infty} E \left(\frac{1}{T} \int_0^T \|W_e e\|^2 + \|W_u u\|^2 dt \right).$$

The diagonal weights W_e and W_u are the main tuning knobs for trading responsiveness and control effort and emphasizing some channels over others. Use the `WeightedVariance` requirement to express this cost function, and use a less stringent performance weight W_e for the outage scenarios.

```
% Nominal tracking requirement
We = diag([20 30 20]); Wu = eye(3);
SoftNom = TuningGoal.WeightedVariance('setpoint',{ 'e','u'}, blkdiag(We,Wu),
SoftNom.Models = 1; % nominal model
```

```
% Tracking requirement for outage conditions
We = diag([8 12 8]); Wu = eye(3);
SoftOut = TuningGoal.WeightedVariance('setpoint',{ 'e','u'}, blkdiag(We,Wu),
SoftOut.Models = 2:9; % outage scenarios
```

For wind gust alleviation, limit the variance of the error signal e due to the white noise wg driving the wind gust model. Again use a less stringent requirement for the outage scenarios.

```
% Nominal gust alleviation requirement
HardNom = TuningGoal.Variance('wg','e',0.01);
HardNom.Models = 1;
```

```
% Gust alleviation requirement for outage conditions
HardOut = TuningGoal.Variance('wg','e',0.03);
HardOut.Models = 2:9;
```

Controller Tuning for Nominal Flight

Set the wind gust speed to 5 m/s and initialize the tunable state-feedback and integrators gains of the controller.

```
GustSpeed = 5;
Ki = eye(3);
Kx = zeros(3,6);
```

Use the `sITunable` interface to acquire a closed-loop model for each of the nine flight conditions. The resulting generalized state-space array `T0` contains nine tunable models parameterized by the gains `Ki`, `Kx`.

```
clear T0
for k = 9:-1:1
    outage = OutageCases(k,:) ;
    ST = sITunable('rct_reliableF16',{'Ki','Kx'}) ;
    T0(:,:,k) = ST.getIOTransfer({'setpoint';'wg'},{'e';'u'}) ;
end
```

`T0`

`T0 =`

```
9x1 array of generalized continuous-time state-space models.
Each model has 6 outputs, 4 inputs, 11 states, and the following blocks:
  Ki: Parametric 3x3 gain, 1 occurrences.
  Kx: Parametric 3x6 gain, 1 occurrences.
```

Type "`ss(T0)`" to see the current value, "`get(T0)`" to see all properties, and

Use `system` to tune the controller gains for the first model (nominal flight condition) subject to the nominal requirements. Treat the wind gust alleviation as a hard constraint.

```
[T,fSoft,gHard] = systune(T0(:,:,1),SoftNom,HardNom);
```

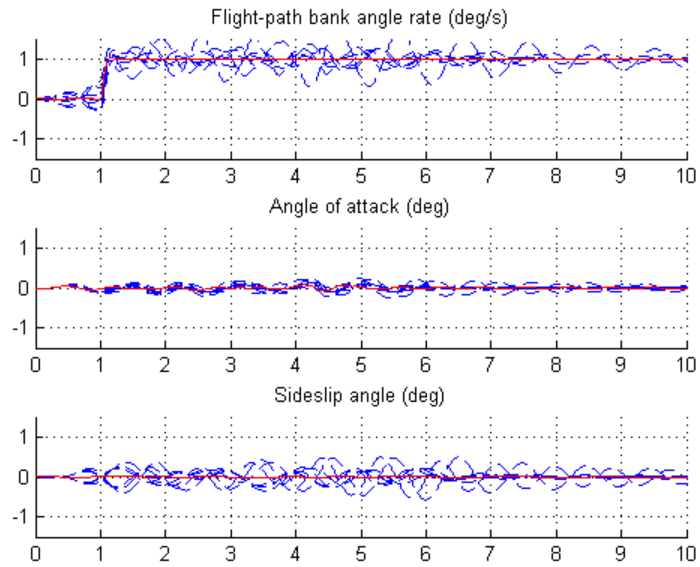
```
Final: Soft = 23.2, Hard = 0.99652, Iterations = 250
```

Retrieve the gain values and simulate the responses to step commands in `mu`, `alpha`, `beta` for the nominal and degraded flight conditions. All simulations include wind gust effects, and the red curve is the nominal response.

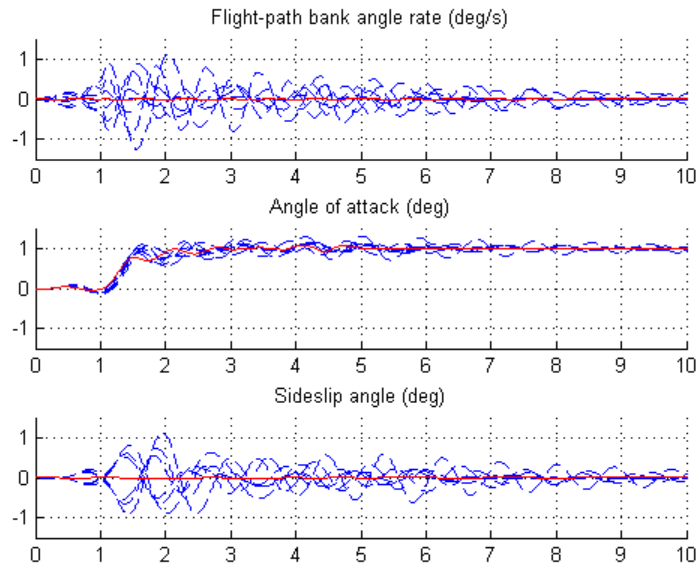
```
Ki = getBlockValue(T, 'Ki'); Ki = Ki.d;
Kx = getBlockValue(T, 'Kx'); Kx = Kx.d;
```

```
% Bank-angle setpoint simulation
```

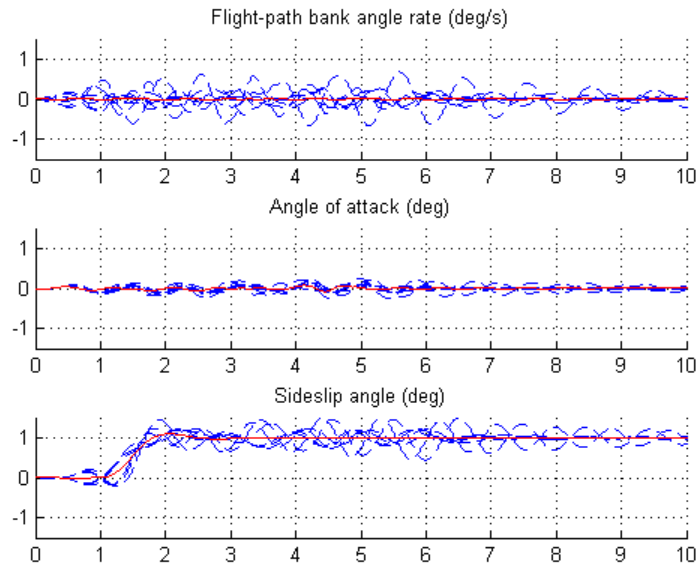
```
plotResponses(OutageCases,1,0,0);
```



```
% Angle-of-attack setpoint simulation  
plotResponses(OutageCases,0,1,0);
```



```
% Sideslip-angle setpoint simulation  
plotResponses(OutageCases,0,0,1);
```



The nominal responses are good but the deterioration in performance is unacceptable when faced with actuator outage.

Controller Tuning for Reliable Flight

To improve reliability, retune the controller gains to meet the nominal requirement for the nominal plant (first model) as well as the relaxed requirements for all eight outage scenarios. Note that `systune` now takes an array of tunable models and tunes the gains against this set of plant models.

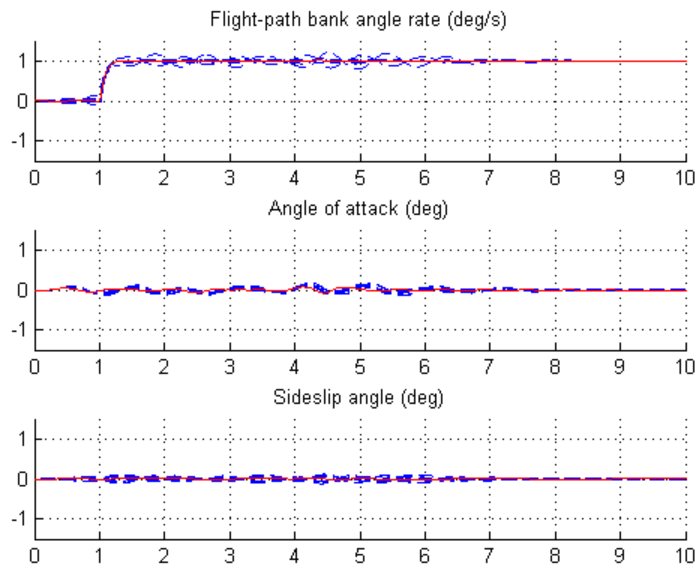
```
[T,fSoft,gHard] = systune(T0,[SoftNom;SoftOut],[HardNom;HardOut]);
```

```
Final: Soft = 25.9, Hard = 0.99758, Iterations = 358
```

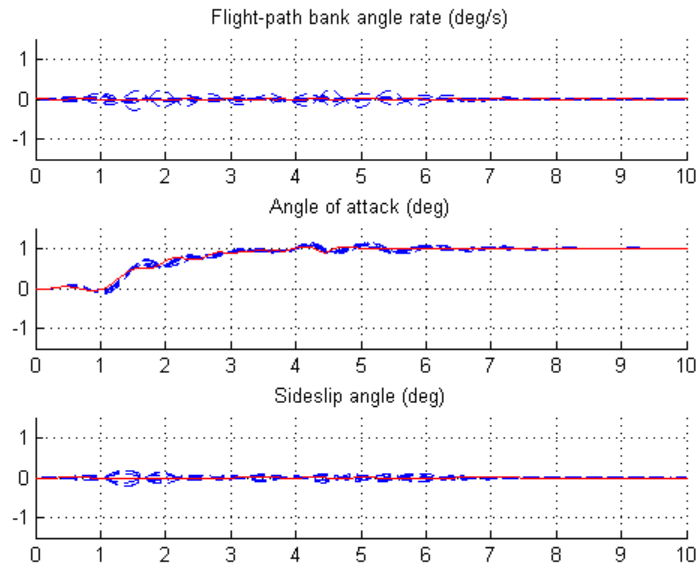
The optimal performance (square root of LQG cost J) is only slightly worse than for the nominal tuning (26 vs. 23). Retrieve the gain values and rerun the simulations (red curve is the nominal response).

```
Ki = getBlockValue(T, 'Ki'); Ki = Ki.d;
Kx = getBlockValue(T, 'Kx'); Kx = Kx.d;
```

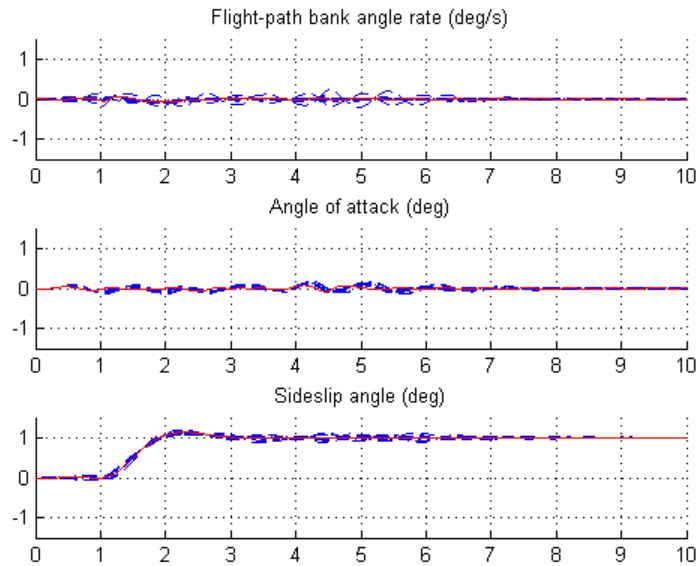
```
% Bank-angle setpoint simulation  
plotResponses(OutageCases,1,0,0);
```



```
% Angle-of-attack setpoint simulation  
plotResponses(OutageCases,0,1,0);
```

```
% Sideslip-angle setpoint simulation  
plotResponses(OutageCases,0,0,1);
```



The controller now provides reliable performance in all flight conditions. The design could be further refined by adding specifications such as minimum stability margins and gain limits to avoid actuator rate saturation.

Fixed-Structure H-infinity Synthesis with HINFSTRUCT

This example uses the `hinfstruct` command to tune a fixed-structure controller subject to H_∞ constraints.

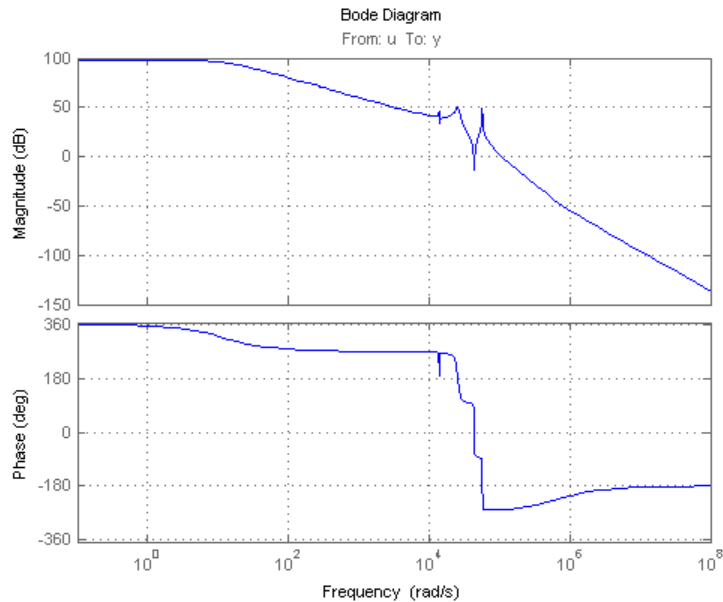
Introduction

The `hinfstruct` command extends classical H_∞ synthesis (see `hifsyn`) to fixed-structure control systems. This command is meant for users already comfortable with the `hifsyn` workflow. If you are unfamiliar with H_∞ synthesis or find augmented plants and weighting functions intimidating, use `sys tune` and `looptune` instead. See "*Tuning Control Systems with SYSTUNE*" for the `sys tune` counterpart of this example.

Plant Model

This example uses a 9th-order model of the head-disk assembly (HDA) in a hard-disk drive. This model captures the first few flexible modes in the HDA.

```
load hinfstruct_demo G
bode(G), grid
```



We use the feedback loop shown below to position the head on the correct track. This control structure consists of a PI controller and a low-pass filter in the return path. The head position y should track a step change r with a response time of about one millisecond, little or no overshoot, and no steady-state error.

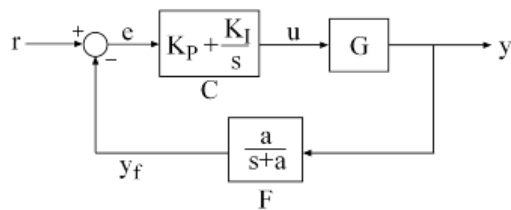


Figure 1: Control Structure

Tunable Elements

There are two tunable elements in the control structure of Figure 1: the PI controller $C(s)$ and the low-pass filter

$$F(s) = \frac{a}{s + a}.$$

Use the `ltiblock.pid` class to parameterize the PI block and specify the filter $F(s)$ as a transfer function depending on a tunable real parameter a .

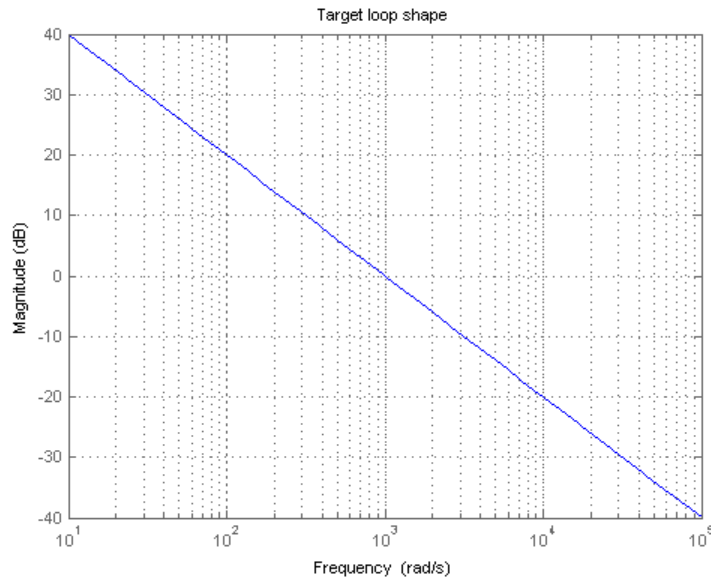
```
C0 = ltiblock.pid('C','pi'); % tunable PI

a = realp('a',1); % filter coefficient
F0 = tf(a,[1 a]); % filter parameterized by a
```

Loop Shaping Design

Loop shaping is a frequency-domain technique for enforcing requirements on response speed, control bandwidth, roll-off, and steady state error. The idea is to specify a target gain profile or "loop shape" for the open-loop response $L(s) = F(s)G(s)C(s)$. A reasonable loop shape for this application should have integral action and a crossover frequency of about 1000 rad/s (the reciprocal of the desired response time of 0.001 seconds). This suggests the following loop shape:

```
wc = 1000; % target crossover
s = tf('s');
LS = (1+0.001*s/wc)/(0.001+s/wc);
bodemag(LS,{1e1,1e5}), grid, title('Target loop shape')
```



Note that we chose a bi-proper, bi-stable realization to avoid technical difficulties with marginally stable poles and improper inverses. In order to tune $C(s)$ and $F(s)$ with `hinfstruct`, we must turn this target loop shape into constraints on the closed-loop gains. A systematic way to go about this is to instrument the feedback loop as follows:

- Add a measurement noise signal n
- Use the target loop shape LS and its reciprocal to filter the error signal e and the white noise source nw .

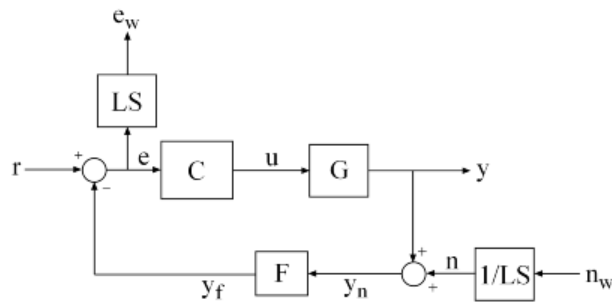


Figure 2: Closed-Loop Formulation

If $T(s)$ denotes the closed-loop transfer function from (r, n_w) to (y, e_w) , the gain constraint

$$\|T\|_{\infty} < 1$$

secures the following desirable properties:

- At low frequency ($\omega < \omega_c$), the open-loop gain stays above the gain specified by the target loop shape LS
- At high frequency ($\omega > \omega_c$), the open-loop gain stays below the gain specified by LS
- The closed-loop system has adequate stability margins
- The closed-loop step response has small overshoot.

We can therefore focus on tuning $C(s)$ and $F(s)$ to enforce $\|T\|_{\infty} < 1$.

Specifying the Control Structure in MATLAB

In MATLAB, you can use the `connect` command to model $T(s)$ by connecting the fixed and tunable components according to the block diagram of Figure 2:

```
% Label the block I/Os
```

```

Wn = 1/LS; Wn.u = 'nw'; Wn.y = 'n';
We = LS; We.u = 'e'; We.y = 'ew';
CO.u = 'e'; CO.y = 'u';
FO.u = 'yn'; FO.y = 'yf';

% Specify summing junctions
Sum1 = sumblk('e = r - yf');
Sum2 = sumblk('yn = y + n');

% Connect the blocks together
T0 = connect(G,Wn,We,CO,FO,Sum1,Sum2,{'r','nw'},{'y','ew'});

```

These commands construct a generalized state-space model T_0 of $T(s)$. This model depends on the tunable blocks C and a :

T_0 .Blocks

```

ans =

      C: [1x1 ltiblock.pid]
      a: [1x1 realp]

```

Note that T_0 captures the following "Standard Form" of the block diagram of Figure 2 where the tunable components C, F are separated from the fixed dynamics.

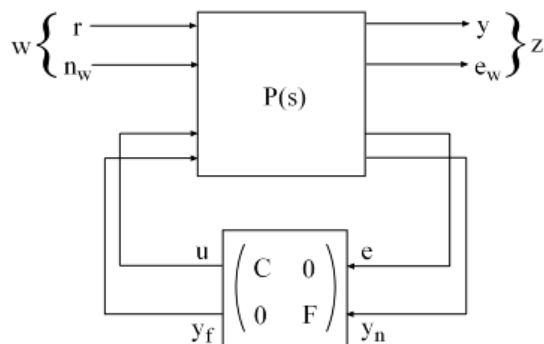


Figure 3: Standard Form for Disk-Drive Loop Shaping

Tuning the Controller Gains

We are now ready to use `hinfstruct` to tune the PI controller C and filter F for the control architecture of Figure 1. To mitigate the risk of local minima, run three optimizations, two of which are started from randomized initial values for C_0 and F_0 :

```
rng('default')
opt = hinfstructOptions('Display','final','RandomStart',4);
T = hinfstruct(T0,opt);
```

```
Final: Peak gain = 1.56, Iterations = 123
Final: Peak gain = 597, Iterations = 199
      Min decay rate 1e-07 is close to lower bound 1e-07
Final: Peak gain = 1.56, Iterations = 131
Final: Peak gain = 1.56, Iterations = 129
Final: Peak gain = 1.56, Iterations = 113
```

The best closed-loop gain is 1.56, so the constraint $\|T\|_{\infty} < 1$ is nearly satisfied. The `hinfstruct` command returns the tuned closed-loop transfer $T(s)$. Use `showTunable` to see the tuned values of C and the filter coefficient α :

```
showTunable(T)
```

```
C =
```

$$K_p + K_i * \frac{1}{s}$$

```
with Kp = 0.000846, Ki = 0.0103
```

```
Name: C
```

```
Continuous-time PI controller in parallel form.
```

```
-----
```

```
a = 5.49e+03
```

Use `getBlockValue` to get the tuned value of $C(s)$ and use `getValue` to evaluate the filter $F(s)$ for the tuned value of a :

```
C = getBlockValue(T, 'C');  
F = getValue(F0, T.Blocks); % propagate tuned parameters from T to F
```

```
tf(F)
```

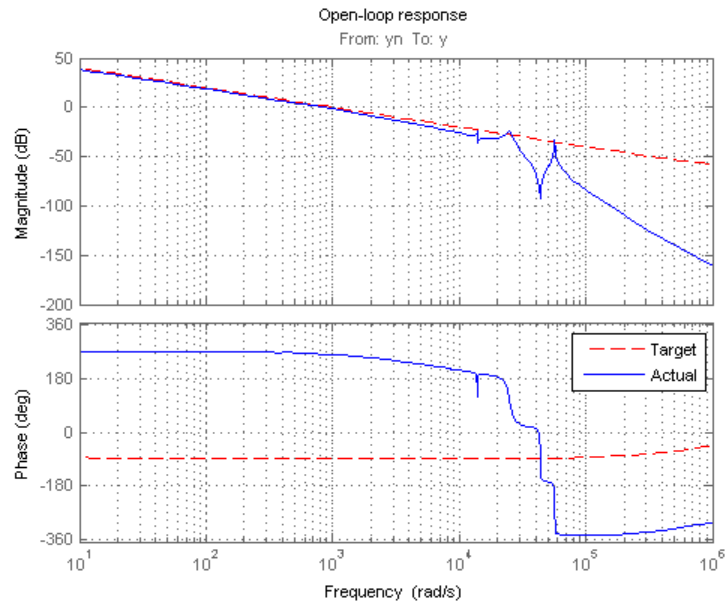
```
ans =
```

```
From input "yn" to output "yf":  
  5486  
-----  
s + 5486
```

Continuous-time transfer function.

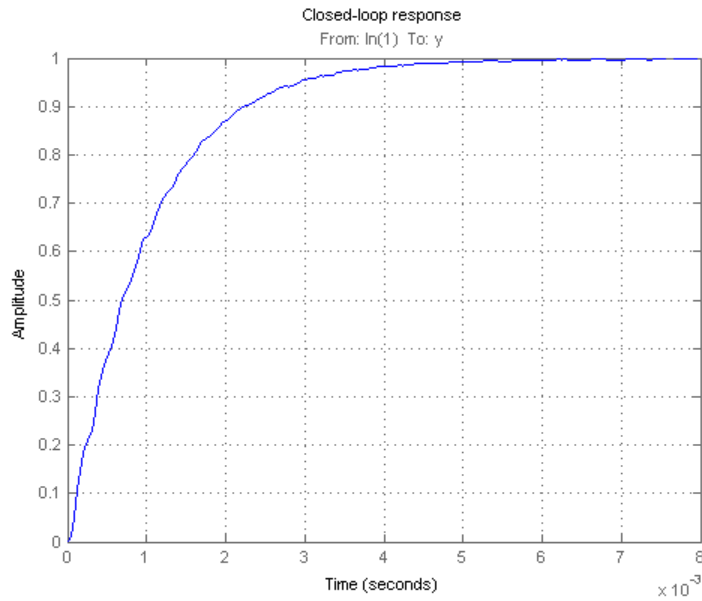
To validate the design, plot the open-loop response $L=F*G*C$ and compare with the target loop shape LS:

```
bode(LS, 'r--', G*C*F, 'b', {1e1, 1e6}), grid,  
title('Open-loop response'), legend('Target', 'Actual')
```



The 0dB crossover frequency and overall loop shape are as expected. The stability margins can be read off the plot by right-clicking and selecting the **Characteristics** menu. This design has 24dB gain margin and 81 degrees phase margin. Plot the closed-loop step response from reference r to position y :

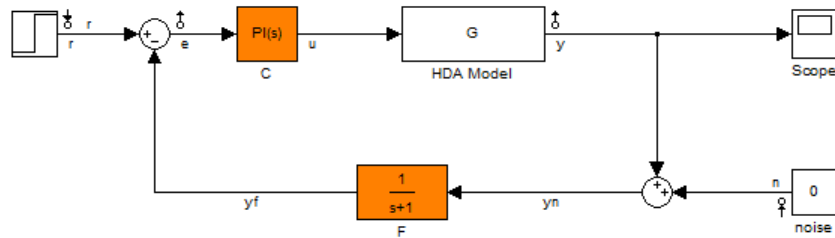
```
step(feedback(G*C,F)), grid, title('Closed-loop response')
```



While the response has no overshoot, there is some residual wobble due to the first resonant peaks in G . You might consider adding a notch filter in the forward path to remove the influence of these modes.

Tuning the Controller Gains from Simulink

Suppose you used this Simulink model to represent the control structure. If you have Simulink Control Design installed, you can tune the controller gains from this Simulink model as follows. First mark the signals r, e, y, n as linearization I/Os:



See `hinfstruct_demo` to see how you can tune the PI gains and the filter coefficient with the `HINFSTRUCT` command.

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Then create an instance of the `slTunable` interface and mark the Simulink blocks `C` and `F` as tunable:

```
ST = slTunable('rct_diskdrive',{'C','F'});
```

Since the filter $F(s)$ has a special structure, explicitly specify how to parameterize the `F` block:

```
a = realp('a',1); % filter coefficient
ST.setBlockParam('F',tf(a,[1 a]));
```

Finally, use the `getIOTransfer` method to derive a tunable model of the closed-loop transfer function $T(s)$ (see Figure 2)

```
% Tunable model of closed-loop transfer (r,n) -> (y,e)
slT0 = ST.getIOTransfer({'r','n'},{'y','e'});
```

```
% Add weighting functions in n and e channels
slT0 = blkdiag(1,LS) * slT0 * blkdiag(1,1/LS);
```

You are now ready to tune the controller gains with `hinfstruct`:

```
rng(0)
opt = hinfstructOptions('Display','final','RandomStart',4);
slT = hinfstruct(slT0,opt);
```

```
Final: Peak gain = 3.89, Iterations = 124
Final: Peak gain = 597, Iterations = 184
      Min decay rate 1.01e-07 is close to lower bound 1e-07
Final: Peak gain = 1.56, Iterations = 125
Final: Peak gain = 1.56, Iterations = 93
Final: Peak gain = 1.56, Iterations = 92
```

Verify that you obtain the same tuned values as with the MATLAB approach:

```
showTunable(s1T)
```

```
C =
```

$$K_p + K_i * \frac{1}{s}$$

```
with Kp = 0.000846, Ki = 0.0103
```

```
Name: C
```

```
Continuous-time PI controller in parallel form.
```

```
-----
```

```
a = 5.49e+03
```

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Use this list to find examples in the documentation.

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“Worst-Case Performance of Uncertain System” on page 1-10

“Loop-Shaping Controller Design” on page 1-13

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